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INTERNAL MIXER INVESTIGATION FOR JT8D ENGINE JET NOISE REDUCTIO--ETC(U)

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# INTERNAL MIXER INVESTIGATION FOR JT8D ENGINE JET NOISE REDUCTION

Volume II—Appendices A, B, C, and D

A.B. Packman and D.C. Eiler



December 1977

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17. Abstract <p>A scale model experimental program was conducted to determine the noise reduction and the impact on propulsive performance that would result from installing a multi-lobed internal mixer on the JT8D engine. Long and short mixer designs were investigated. One-seventh scale mixer models, designed to permit lobe geometry variations, were fabricated and tested along with a model of the JT8D reference exhaust system.</p> <p>The test results indicated that, in general, the long and short mixers produced 3-4 PNdB reduction in Peak Perceived Noise Level relative to the reference exhaust system. Exhaust system performance, in terms of improvement in cruise thrust specific fuel consumption (TSFC), and impact on takeoff thrust, was somewhat better for the long mixer than for the short mixer configurations. However, the short mixers offer significant advantages in terms of weight savings and minimized the hardware changes required for installation in the current JT8D engines. Based on the noise and performance test results in conjunction with the installation considerations, a short mixer design was recommended for evaluation in a full scale engine test.</p>		
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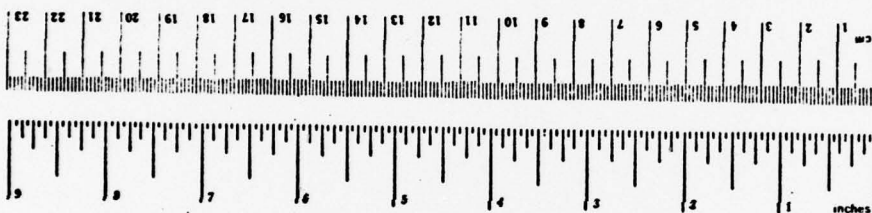
# METRIC CONVERSION FACTORS

## Approximate Conversions to Metric Measures

Symbol	When You Know	Multiply by	To Find	Symbol
<b>LENGTH</b>				
in	inches	2.5	centimeters	cm
ft	feet	30	centimeters	cm
yd	yards	0.9	meters	m
mi	miles	1.6	kilometers	km
<b>AREA</b>				
sq in	square inches	6.5	square centimeters	cm <sup>2</sup>
sq ft	square feet	0.09	square meters	m <sup>2</sup>
sq yd	square yards	0.8	square meters	m <sup>2</sup>
sq mi	square miles	2.6	square kilometers	km <sup>2</sup>
acres	acres	0.4	hectares	ha
<b>MASS (weight)</b>				
oz	ounces	28	grams	g
lb	pounds	0.45	kilograms	kg
	short tons (2000 lb)	0.9	tonnes	t
<b>VOLUME</b>				
teaspoon	teaspoons	5	milliliters	ml
Tablespoon	tablespoons	15	milliliters	ml
fl oz	fluid ounces	30	milliliters	ml
cup	cups	0.24	liters	l
pint	pints	0.47	liters	l
quart	quarts	0.95	liters	l
gallon	gallons	3.8	liters	l
cu ft	cubic feet	0.03	cubic meters	m <sup>3</sup>
cu yd	cubic yards	0.76	cubic meters	m <sup>3</sup>
<b>TEMPERATURE (exact)</b>				
°F	Fahrenheit temperature	5/9 (after subtracting 32)	Celsius temperature	°C

## Approximate Conversions from Metric Measures

Symbol	When You Know	Multiply by	To Find	Symbol
<b>LENGTH</b>				
mm	millimeters	0.04	inches	in
cm	centimeters	0.4	inches	in
m	meters	3.3	feet	ft
km	kilometers	1.1	yards	yd
		0.6	miles	mi
<b>AREA</b>				
cm <sup>2</sup>	square centimeters	0.16	square inches	in <sup>2</sup>
m <sup>2</sup>	square meters	1.2	square yards	yd <sup>2</sup>
ha	square kilometers	0.4	square miles	mi <sup>2</sup>
	hectares (10,000 m <sup>2</sup> )	2.5	acres	acres
<b>MASS (weight)</b>				
g	grams	0.035	ounces	oz
kg	kilograms	2.2	pounds	lb
t	tonnes (1000 kg)	1.1	short tons	short tons
<b>VOLUME</b>				
ml	milliliters	0.03	fluid ounces	fl oz
l	liters	2.1	pints	pt
		1.06	quarts	qt
		0.26	gallons	gal
m <sup>3</sup>	cubic meters	35	cubic feet	ft <sup>3</sup>
		1.3	cubic yards	yd <sup>3</sup>
<b>TEMPERATURE (exact)</b>				
°C	Celsius temperature	9/5 (then add 32)	Fahrenheit temperature	°F



1 inch = 2.54 centimeters. For other exact conversions, use the metric system. See NIST Special Publication 800-43, Units of Length and Mass, for more information.

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APPENDIX C	Nozzle Traverse Results; and
APPENDIX D	Hot/Cold Flow Model Tests to Determine Static Performance of 1/7-Scale JT8D Mixer Exhaust Nozzles.



## APPENDIX A

### TEST CONDITIONS FOR ACOUSTIC AND EXIT PROFILE TESTS

The operating condition for each test point of the configurations tested under the contract are included in this Appendix. The data are listed in the following order:

Configuration Name	Configuration Number
Basic Long Flowpath Mixer	1A
Reference Exhaust System	2Au
Reference Exhaust System (tested at ideally mixed conditions)	2Am
Shallow Scalloped Long Flowpath Mixer	3A
Scalloped Long Flowpath Mixer	4A
Cutback Scalloped Long Flowpath Mixer	5A
Cutback Unscalloped Long Flowpath Mixer	6A
Scalloped Long Flowpath Mixer with Engine Flow Simulation	7A
Cutback Scalloped Long Flowpath Mixer with 7.6 inch Engine Extension	8A
Severe Cutback Short Flowpath Mixer with Engine Flow Simulation	10A
Severe Cutback Short Flowpath Mixer	11A

NOTE: Unless noted otherwise, the long flowpath mixers were tested with a 16 inch engine extension, and the short flowpath mixer was tested without an engine extension.

The following nomenclature is used:

BPR	-	Bypass ratio, fan flowrate divided by primary flowrate
$C_a$	-	Speed of sound in the test chamber, ft/sec.
$F_n$	-	Full scale JT8D engine thrust on an FAA day, lbs
$P_a$	-	Static pressure in the test chamber,
$P_t/P_a$	-	Nozzle pressure ratio, nozzle total pressure divided by test chamber ambient pressure
$RH_a$	-	Relative humidity in the test chamber
$T_a$	-	Temperature in the test chamber, °F
$T_t$	-	Nozzle total temperature, °R
$V$	-	Jet velocity; ideally expanded to test chamber pressure, ft/sec.
$V/C_a$	-	Ratio of jet velocity to test chamber speed of sound
$W$	-	Flowrate, lbm/sec
Primary	-	Primary or inner exhaust stream
Fan	-	Fan or outer exhaust stream
Ideal Mix	-	The mass flow average value, e.g., $V_{mix} = \frac{W_{pri} V_{pri} + W_{fan} V_{fan}}{W_{pri} + W_{fan}}$

The full scale JT8D engine thrust,  $F_n$ , was determined as follows:



- A. Reference System: The  $V_{pri}/C_a$  of each test point was related to the FAA day JT8D engine thrust by the curve of  $V_{pri}/C_a$  vs.  $F_n$ , Figure A-1, developed from full scale engine test data\*. Since for low bypass exhausts the jet noise is essentially independent of small variations in the secondary flow conditions, this procedure allows the model data to be scaled using primary velocity to predict the full size engine noise.
- B. Mixer Nozzles: The  $V_{mix}/C_a$  of each test point was related to the FAA day JT8D engine thrust by the  $V_{mix}/C_a$  vs.  $F_n$  curve of Figure A-1, which was determined by calculating the  $V_{mix}/C_a$  for the engine operating line conditions by the formula defined in the nomenclature listed above\*. For exhausts with internal mixer nozzles, the jet noise is a function of the calculated mixed velocity. This procedure allows small variations in bypass ratio and the resultant effect on mixed velocity to be accounted for correctly, thus, allowing the mixer nozzle data to be scaled properly to predict the effect of a mixer on JT8D engine noise.

The above procedure for relating JT8D engine noise with thrust is based on observations that jet noise is a direct function of the characteristic jet velocity divided by the ambient (test chamber) speed of sound, as shown in ref. 2. Since the model nozzles were tested at pressure, temperature, and density conditions simulating the JT8D engine, density effects on noise could be neglected.

\*For these calculations, it was assumed that the installation of a mixer on the JT8D engine would not change the engine thrust-airflow relationship established for the baseline configuration.



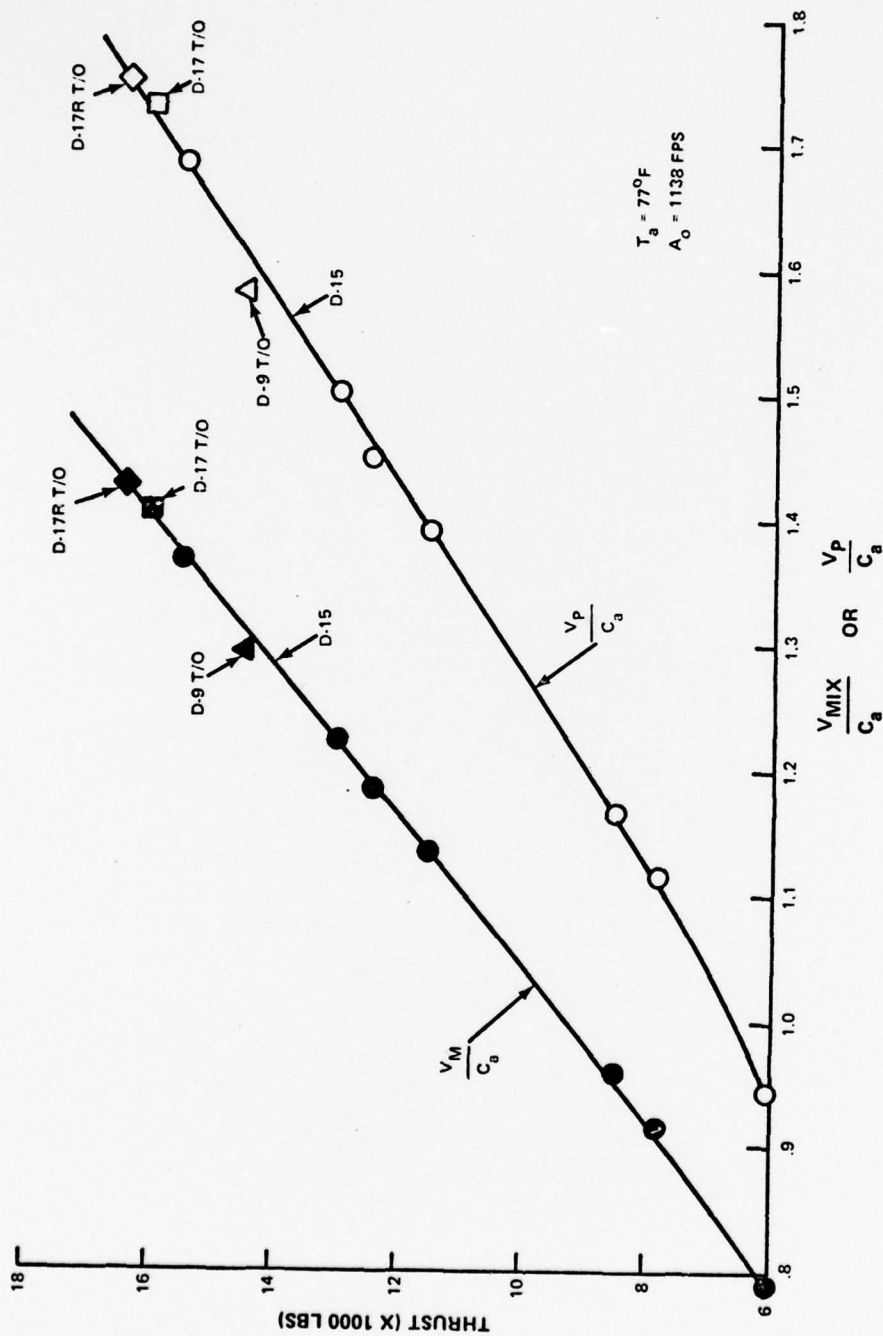


Figure A.1 Thrust vs  $V/C_a$  for Reference and Mixed JT8D-15



TABLE 1A

BASIC LONG FLOWPATH MIXER; WITHOUT ENGINE SECONDARY FLOW SIMULATION;  
CONFIGURATION 1A

Condition Number	Primary				Fan				Ideal Mix				Test Chamber			
	$\frac{P_T}{P_a}$	$T_T$ (°R)	V (FPS)	W (lb/sec)	$\frac{P_T}{P_a}$	$T_T$ (°R)	V (FPS)	W (lb/sec)	BPR	$T_{Tmix}$ (°R)	$\frac{V_{mix}}{C_a}$	$\frac{V_{mix}}{C_a}$ (FPS)	$T_a$ (°F)	$\frac{P_a}{(PSI)}$	$RH_a$	$F_n$ (lb)
5310	2.153	1616.7	1969.	3.109	2.009	710.6	1243.0	3.423	1.101	1,141.9	1.588.5	1.43	52	14.8	30	16500
5309	2.135	1606.7	1947.	3.081	1.995	700.8	1228.0	3.425	1.112	1,129.1	1,567.9	1.41	52	14.8	30	16150
5308*	2.080	1569.7	1808.	3.005	1.963	690.0	1206.0	3.433	1.143	1,100.5	1,528.9	1.38	52	14.8	31	15650
5306	1.858	1462.2	1694.	2.715	1.789	676.5	1119.0	3.225	1.188	1,035.6	1,381.8	1.24	53	14.8	29	13250
5305*	1.798	1434.3	1635.	2.622	1.741	667.4	1087.0	3.153	1.203	1,015.3	1,338.8	1.20	53	14.8	28	12600
5304	1.696	1410.4	1546.	2.345	1.740	653.8	1069.0	3.390	1.445	963.2	1,264.1	1.14	51	14.8	28	11550
5303	1.527	1316.8	1347.	2.203	1.483	625.9	897.0	2.646	1.201	939.8	1,101.5	1.99	52	14.8	27	9000
5302	1.504	1249.9	1288.	2.218	1.446	622.6	863.0	2.557	1.153	914.0	1,060.4	.96	53	14.8	26	8500
5301	1.390	1239.3	1158.	1.905	1.368	616.0	798.0	2.402	1.261	891.7	957.2	.86	51	14.8	26	6900

TABLE 2A<sub>u</sub>REFERENCE EXHAUST SYSTEM WITHOUT ENGINE SECONDARY FLOW SIMULATION;  
CONFIGURATION 2A<sub>u</sub>

Condition Number	Primary				Fan				BPR	Ideal Mix		V pri. C <sub>a</sub>	Test Chamber			F <sub>n</sub> (lb)
	P <sub>T</sub> P <sub>a</sub>	T <sub>T</sub> (°R)	V (FPS)	W (lb/sec)	P <sub>T</sub> P <sub>a</sub>	T <sub>T</sub> (°R)	V (FPS)	W (lb/sec)		T <sub>Tmix</sub> (°R)	Vmix (FPS)		T <sub>a</sub> (°F)	P <sub>a</sub> (PSI)	RH <sub>a</sub>	
5510	2.150	1607.7	1958	3.530	2.012	700.4	1234	3.042	.862	1187.7	1623	1.76	53	14.7	61	16500
5509	2.113	1600.2	1935	3.472	1.985	697.7	1221	3.036	.874	1179.3	1602	1.74	53	14.7	60	16250
5508*	2.057	1573.7	1886	3.339	1.915	687.8	1184	3.086	.924	1148.2	1549	1.70	54	14.7	60	15700
5507	1.993	1524.2	1819	3.290	1.806	681.1	1169	3.027	.920	1120.2	1508	1.64	53	14.7	59	14900
5506	1.862	1464.2	1700	3.058	1.790	673.9	1115	2.922	.956	1077.9	1414	1.53	53	14.7	59	13400
5505*	1.807	1435.8	1644	2.934	1.755	664.5	1089	2.960	1.009	1048.4	1365	1.48	53	14.7	58	12700
5504	1.743	1405.2	1580	2.829	1.694	647.1	1043	2.864	1.012	1023.9	1310	1.42	55	14.7	55	11900
5503	1.486	1314.6	1301	2.157	1.516	628.7	919	2.768	1.283	929.1	1086	1.17	53	14.7	62	8550
5502	1.434	1314.0	1240	2.041	1.464	617.1	872	2.653	1.300	920.1	1032	1.12	50	14.7	68	7850
5501	1.313	1265.4	1068	1.709	1.350	599.2	768	2.393	1.400	876.8	893	.96	50	14.7	68	6200



TABLE 2A<sub>m</sub>

REFERENCE EXHAUST SYSTEM (TESTED AT IDEALLY MIXED CONDITIONS);  
WITHOUT ENGINE SECONDARY FLOW SIMULATION; CONFIGURATION 2A<sub>m</sub>

Condition Number	Primary			Fan			BPR	Ideal Mix		Test Chamber			F <sub>n</sub> (lb)	
	$\frac{P}{P_a}$	$\frac{T}{T_a}$ (°R)	$\frac{V}{V_a}$ (FPS)	$\frac{W}{W_a}$ (lb/sec)	$\frac{P}{P_a}$	$\frac{T}{T_a}$ (°R)		$\frac{V}{V_a}$ (FPS)	$\frac{V_{mix}}{C_a}$	$\frac{T_a}{T_a}$ (°F)	$\frac{P_a}{P_a}$ (PSI)	$\frac{RH_a}{RH_a}$		
7409	2.060	1156.9	1608.7	3.423	2.077	1166.8	1624.1	2.921	0.853	1161.5	1615.8	1.45	60	16800
7408*	2.020	1154.8	1587.0	3.370	2.036	1154.2	1595.0	2.877	0.854	1154.5	1590.7	1.42	60	16300
7407	1.982	1126.9	1547.6	3.367	1.983	1131.4	1551.8	2.802	0.832	1128.9	1549.5	1.39	60	15800
7406	1.858	1091.0	1463.2	3.156	1.881	1090.8	1476.4	2.752	0.874	1090.9	1469.4	1.32	52	14600
7405*	1.792	1038.7	1388.6	3.138	1.796	1035.4	1388.9	2.640	0.841	1037.2	1388.7	1.27	41	13800
7404	1.742	1020.0	1345.7	3.029	1.763	1016.4	1356.5	2.658	0.878	1018.3	1350.7	1.23	41	13100
7403	1.667	982.1	1269.4	2.915	1.689	993.9	1291.8	2.547	0.874	987.6	1279.8	1.17	41	12100
7402	1.487	925.8	1095.2	2.567	1.494	917.6	1096.6	2.220	0.886	922.0	1095.8	1.00	41	9200
7401	1.436	900.2	1034.4	2.447	1.446	901.2	1044.2	2.148	0.878	900.7	1039.0	.95	42	8300

TABLE 3A

SHALLOW SCALLOPED LONG FLOWPATH MIXER; WITHOUT ENGINE SECONDARY  
FLOW SIMULATION; CONFIGURATION 3A

Condition Number	Primary			Fan			BPR	Ideal Mix		Test Chamber				
	$\frac{P}{P_a}$	$\frac{T}{T_a}$ (°R)	$\frac{V}{V_a}$ (FPS)	$\frac{W}{W_a}$ (lb/sec)	$\frac{P}{P_a}$	$\frac{T}{T_a}$ (°R)		$\frac{V}{V_a}$ (FPS)	$\frac{T}{T_a}$ (°F)	$P_a$ (PSI)	$RH_a$	$F_n$ (lb)		
5711	2.133	1617.2	1957	3.144	2.014	705.5	1241	3.291	1.047	1150.9	1591	1.44	45	16650
5710	2.109	1610.2	1940	3.069	1.997	699.4	1229	3.278	1.068	1139.8	1573	1.43	45	16500
5709*	2.037	1575.2	1877	2.966	1.946	686.2	1196	3.232	1.090	1111.6	1523	1.38	46	15600
5708	1.962	1530.2	1804	2.943	1.900	678.5	1170	3.219	1.094	1085.2	1478	1.34	46	15000
5707	1.887	1473.2	1722	2.794	1.818	680.9	1134	3.131	1.121	1054.5	1411	1.28	46	13950
5706*	1.784	1444.2	1634	2.594	1.744	665.4	1085	3.027	1.167	1024.8	1338	1.21	46	12750
5705	1.719	1432.2	1578	2.502	1.693	647.0	1042	2.999	1.199	1004.1	1286	1.17	46	12100
5704	1.477	1321.2	1297.	2.024	1.524	635.2	931	2.790	1.379	923.6	1085	.98	46	8850
5703	1.422	1304.2	1227.	2.136	1.473	622.9	885	2.664	1.247	926.1	1837	.94	47	8200
5712	1.305	1250.7	1052	1.536	1.356	609.7	782	2.346	1.529	863.2	889	.81	43	6200



TABLE 4A

DEEP SCALLOPED LONG FLOWPATH MIXER; WITHOUT ENGINE SECONDARY FLOW SIMULATION; CONFIGURATION 4A

Condition Number	Primary				Fan				Ideal Mix				Test Chamber			
	$\frac{P_T}{P_a}$	$T_T$ (°R)	V (FPS)	W (lb/sec)	$\frac{P_T}{P_a}$	$T_T$ (°R)	V (FPS)	W (lb/sec)	BPR	$T_{Tmix}$ (°R)	$V_{mix}$ (FPS)	$\frac{V_{mix}}{C_a}$	$T_a$ (°F)	$P_a$ (PSI)	$RH_a$	$F_n$ (lb)
6120	2.145	1605.7	1957	3.239	2.001	697.7	1230	3.104	0.958	1161.4	1601.3	1.45	47	14.5	64	16800
6119	2.122	1605.7	1944	3.203	1.982	694.8	1219	3.072	0.959	1159.8	1589.1	1.44	47	14.5	63	16650
6118*	2.085	1561.7	1896	3.158	1.963	696.5	1213	3.095	0.980	1133.5	1557.9	1.41	47	14.5	62	16100
6117	1.986	1516.7	1812	3.031	1.885	681.3	1167	3.044	1.004	1098.2	1488.9	1.35	47	14.5	62	15100
6116	1.863	1466.7	1700	2.826	1.792	668.3	1111	2.952	1.044	1058.9	1399.2	1.27	47	14.5	63	13800
6115*	1.800	1436.7	1640	2.838	1.755	664.8	1089	2.909	1.025	1046.0	1361.1	1.23	47	14.5	63	13100
6114	1.747	1454.6	1612	2.586	1.711	647.7	1053	2.891	1.118	1028.7	1316.9	1.19	48	14.5	64	12400
6113	1.486	1319.8	1303	2.075	1.513	616.2	908	2.658	1.281	924.7	1081.2	0.98	48	14.5	65	8850
6112	1.451	1296.9	1256	2.038	1.472	617.1	880	2.547	1.256	918.4	1046.7	0.95	48	14.5	67	8350
6111	1.314	1258.6	1065	1.649	1.366	613.8	791	2.334	1.415	880.8	904.5	0.82	48	14.5	66	6350

TABLE 5A

CUTBACK SCALLOPED LONG FLOWPATH MIXER; WITHOUT ENGINE SECONDARY FLOW SIMULATION; CONFIGURATION 5A

Condition Number	Primary				Fan				Ideal Mix				Test Chamber			
	$\frac{P_T}{P_a}$	$T_T$ (°R)	V (FPS)	W (lb/sec)	$\frac{P_T}{P_a}$	$T_T$ (°R)	V (FPS)	W (lb/sec)	BPR	$T_{Tmix}$ (°R)	$V_{mix}$ (FPS)	$\frac{V_{mix}}{C_a}$	$T_a$ (°F)	$P_a$ (PSI)	$RH_a$	$F_n$ (lb)
6510	2.140	1616.7	1960	3.282	2.018	694.2	1232	3.231	0.984	1159.2	1598.9	1.44	51	14.9	18	16650
6509	2.136	1608.7	1953	3.266	2.023	701.6	1241	3.270	1.001	1154.9	1596.8	1.44	51	15.0	18	16650
6508*	2.067	1557.7	1884	3.210	1.961	689.6	1206	3.210	1.000	1123.7	1545.0	1.39	51	15.0	17	15800
6507	1.976	1524.7	1812	3.080	1.878	675.9	1159	3.121	1.013	1097.6	1483.4	1.34	51	15.0	17	14950
6506	1.872	1480.7	1716	2.829	1.801	664.0	1112	3.033	1.041	1064.1	1407.9	1.30	29	14.9	18	14750
6505*	1.806	1433.9	1644	2.714	1.756	657.8	1085	2.996	1.059	1034.7	1356.5	1.25	29	14.9	18	13400
6504	1.739	1408.1	1578	3.714	1.695	650.7	1046	2.904	1.070	1016.6	1303.0	1.20	29	14.9	18	12600
6503	1.485	1318.5	1301	2.135	1.517	625.7	918	2.717	1.273	930.5	1086.5	1.00	32	14.9	17	9200
6502	1.440	1291.4	1241	2.043	1.471	619.4	882	2.618	1.282	912.9	1039.3	0.96	32	14.9	16	8500
6501	1.310	1270.6	1067	1.701	1.361	603.0	782	2.396	1.405	880.2	900.4	0.83	34	14.9	16	6500



TABLE 6A

## CUTBACK NON-SCALLOPED LONG FLOWPATH MIXER; WITHOUT SECONDARY FLOW SIMULATION; CONFIGURATION 6A

Condition Number	Primary				Fan				Ideal Mix			Test Chamber				
	$\frac{P_T}{P_a}$	$T_T$ ( $^{\circ}$ R)	V (FPS)	W (lb/sec)	$\frac{P_T}{P_a}$	$T_T$ ( $^{\circ}$ R)	V (FPS)	W (lb/sec)	BPR	$T_{T\text{ mix}}$ ( $^{\circ}$ R)	$V_{\text{mix}}$ (FPS)	$\frac{V_{\text{mix}}}{C_a}$	$T_a$ ( $^{\circ}$ F)	$\frac{P_a}{P_a}$ (PSI)	$RH_a$	$F_n$ (lb)
6310	2.142	1617.7	1964	3.269	1.999	697.2	1230	3.197	0.978	1162.6	1601.1	1.46	39	14.9	22	17000
6309	2.130	1604.7	1944	3.257	1.997	698.0	1229	3.215	0.87	1154.3	1588.8	1.45	38	14.9	22	16800
6308*	2.073	1570.7	1888	3.163	1.965	691.9	1207	3.226	1.020	1126.9	1544.1	1.41	39	14.9	22	16100
6307	1.975	1477.7	1781	3.061	1.870	678.1	1157	3.084	1.007	1076.5	1467.9	1.34	39	14.9	21	15000
6306	1.858	1466.7	1694	2.830	1.791	668.1	1112	3.045	1.076	1052.8	1392.3	1.27	39	14.9	21	13800
6305*	1.797	1426.3	1634	2.762	1.754	662.8	1085	3.042	1.102	1026.0	1346.2	1.23	39	14.9	21	13100
6304	1.743	1419.1	1588	2.658	1.708	649.6	1052	2.979	1.121	1012.4	1304.7	1.19	39	14.9	21	12400
6303	1.483	1333.7	1307	2.084	1.521	624.0	917	2.766	1.327	929.0	1084.6	0.99	39	14.9	21	9000
6302	1.441	1308.7	1251	2.032	1.464	619.4	877	2.592	1.276	922.3	1041.3	0.95	39	14.9	21	8350
6301	1.314	1264.2	1067	1.688	1.360	601.6	778	2.408	1.426	874.7	897.1	0.82	39	14.9	20	6350

TABLE 7A

## DEEP SCALLOPED LONG FLOWPATH MIXER; WITH ENGINE SECONDARY FLOW SIMULATION; CONFIGURATION 7A

Condition Number	Primary				Fan				Ideal Mix			Test Chamber				
	$\frac{P_T}{P_a}$	$T_T$ ( $^{\circ}$ R)	V (FPS)	W (lb/sec)	$\frac{P_T}{P_a}$	$T_T$ ( $^{\circ}$ R)	V (FPS)	W (lb/sec)	$V_{mix}$ (FPS)	$\frac{V_{mix}}{C_a}$	$T_a$ ( $^{\circ}$ F)	$\frac{P_a}{P_a}$ (PSI)	$RH_a$	$F_n$ (lb)		
7510	2.118	1629.4	1950.3	3.172	2.000	697.5	1224.4	3.063	0.966	1171.5	1593.6	1.44	49	14.5	20	16550
7509	2.136	1661.4	1980.3	3.141	1.991	691.0	1215.0	3.068	0.976	1182.1	1602.3	1.45	46	14.5	19	16800
7508*	2.069	1546.7	1872.0	3.052	1.958	682.7	1193.9	3.093	1.013	1111.9	1530.8	1.42	25	14.5	18	16300
7507	1.979	1518.4	1804.5	3.097	1.892	687.7	1172.1	2.932	0.947	1114.4	1496.9	1.39	20	14.5	18	15800
7506	1.859	1460.0	1691.8	2.789	1.773	678.9	1108.3	2.835	1.016	1066.3	1397.7	1.30	19	14.5	18	14250
7505	1.797	1438.5	1635.7	2.660	1.745	664.8	1081.9	2.874	1.080	1036.8	1348.2	1.26	19	14.6	17	13600
7504	1.740	1424.0	1585.3	2.554	1.701	656.9	1052.7	2.806	1.098	1022.5	1306.4	1.22	19	14.6	17	12900
7503	1.471	1324.4	1290.4	1.994	1.542	627.8	935.9	2.725	1.367	922.1	1085.7	1.01	18	14.6	17	9350
7502	1.447	1290.1	1264.9	1.980	1.492	620.5	908.3	2.628	1.327	908.3	1061.5	0.97	35	14.7	25	8700
7501	1.310	1275.1	1086.4	1.609	1.372	599.0	800.1	2.361	1.467	873.1	916.2	0.84	35	14.7	24	6650



**CUTBACK SCALLOPED LONG FLOWPATH MIXER; WITH 7.6" ENGINE EXTENSION;  
WITHOUT ENGINE SECONDARY FLOW SIMULATION: CONFIGURATION 8A**

Condition Number	Primary			W (lb/sec)	Fan		BPR	Ideal Mix		Test Chamber				Fn (lb)	
	$\frac{P_T}{P_a}$	$\frac{T_T}{(^{\circ}R)}$	$\frac{V}{(FPS)}$		$\frac{P_T}{P_a}$	$\frac{T_T}{(^{\circ}R)}$		$\frac{V_{mix}}{(FPS)}$	$\frac{V_{mix}}{C_a}$	$T_a$ ( $^{\circ}F$ )	$\frac{P_a}{(PSI)}$	RH <sub>a</sub>			
7610	2.141	1598.4	1950.5	3.236	2.015	693.0	1230.3	3.054	1158.7	1600.8	1.45	46	14.5	28	16800
7609	2.133	1614.0	1955.4	3.215	1.994	696.5	1225.4	3.002	1170.9	1602.9	1.45	46	14.5	27	16800
7608	2.060	1558.4	1880.7	3.125	1.959	685.2	1201.3	3.012	1129.8	1547.2	1.40	46	14.5	25	16000
7607	1.980	1521.0	1809.8	3.037	1.889	678.4	1165.2	2.916	1108.3	1494.1	1.36	45	14.5	24	15300
7606	1.859	1466.3	1698.1	2.875	1.790	667.7	1108.6	2.782	1073.5	1408.1	1.28	43	14.5	22	13950
7605	1.788	1441.7	1634.9	2.771	1.742	660.3	1079.8	2.735	1053.6	1359.2	1.24	42	14.5	21	13250
7604	1.735	1404.3	1573.4	2.699	1.694	648.8	1044.4	2.673	1028.4	1310.2	1.19	42	14.5	20	12400
7602	1.435	1313.0	1246.8	2.105	1.468	620.3	880.3	2.238	956.1	1058.0	0.96	42	14.5	19	8500
7601	1.315	1273.1	1074.5	1.797	1.356	607.3	779.7	1.722	947.3	930.3	0.85	44	14.5	18	6750

**SEVERE CUTBACK SHORT FLOWPATH MIXER; WITH ENGINE SECONDARY FLOW SIMULATION: CONFIGURATION 10A**

Condition Number	Primary				Fan				BPR	Ideal Mix		Test Chamber				Fn (lb)
	$\frac{P_T}{P_a}$	$T_T$ (°R)	$\dot{V}$ (FPS)	$\dot{W}$ (lb/sec)	$\frac{P_T}{P_a}$	$T_T$ (°R)	$\dot{V}$ (FPS)	$\dot{W}$ (lb/sec)		$T_{T \text{ mix}}$ (°R)	$\dot{V}_{\text{mix}}$ (FPS)	$\frac{\dot{V}_{\text{mix}}}{C_a}$	$T_a$ (°F)	$P_a$ (PSI)	$RH_2$	
7810	2.144	1602.0	1952.3	3.235	2.008	699.0	1231.8	2.922	0.903	1173.5	1610.4	1.44	64	14.5	21	16650
7809	2.123	1600.7	1940.8	3.206	1.982	697.4	1220.5	2.883	0.899	1173.1	1599.8	1.43	64	14.5	21	16500
7808*	2.059	1567.7	1884.5	3.108	1.948	687.1	1197.2	2.916	0.938	1141.5	1551.8	1.38	64	14.5	21	15600
7807	1.995	1519.7	1817.2	3.031	1.880	677.1	1158.9	2.843	0.938	1111.9	1498.6	1.34	63	14.5	21	15000
7806	1.860	1463.3	1697.8	2.837	1.776	667.0	1101.7	2.729	0.962	1072.9	1405.5	1.25	63	14.5	21	13400
7805*	1.798	1429.9	1635.1	2.725	1.748	657.5	1080.4	2.765	1.014	1041.0	1355.8	1.21	63	14.5	21	12750
7804	1.734	1398.7	1569.0	2.608	1.693	647.9	1047.2	2.714	7.041	1015.8	1300.6	1.16	63	14.5	21	11900
7803	1.486	1323.7	1306.0	2.084	1.509	631.1	916.6	2.478	1.187	947.8	1094.7	0.97	65	14.5	20	8700
7802	1.430	1290.1	1230.1	1.982	1.451	616.3	865.2	2.358	1.189	924.1	1031.9	0.92	65	14.5	20	7800
7801	1.309	1244.1	1052.9	1.633	1.357	596.9	773.5	2.221	1.360	871.1	891.9	0.79	65	14.5	20	6900



TABLE 11A  
SEVERE CUTBACK SHORT FLOWPATH MIXER; WITHOUT ENGINE SECONDARY  
FLOW SIMULATION; CONFIGURATION 11A

Condition Number	Primary				Fan			Ideal Mix		Test Chamber				Fn (lb)	
	$\frac{P_T}{P_a}$	$T_T$ (°R)	$V$ (FPS)	$W$ (lb/sec)	$\frac{P_T}{P_a}$	$T_T$ (°R)	$V$ (FPS)	$W$ (lb/sec)	$T_{T\text{ mix}}$ (°R)	$V_{\text{ mix}}$ (FPS)	$\frac{V_{\text{ mix}}}{C_a}$	$T_a$ (°F)	$P_a$ (PSI)		$RH_a$
7910	2.151	1622.0	1968.6	3.424	2.010	704.0	1237.0	2.864	1204.0	1635.5	1.46	62	14.6	18	17000
7909	2.137	1631.0	1966.6	3.356	1.999	704.1	1232.7	2.851	1205.1	1629.4	1.46	61	14.6	18	17000
7908*	2.070	1571.0	1893.3	3.272	1.974	692.3	1213.4	2.929	1156.0	1572.2	1.41	59	14.6	18	16100
7907	1.989	1529.4	1821.8	3.165	1.898	683.3	1174.0	2.841	1089.1	1515.3	1.36	58	14.6	18	15300
7906	1.861	1471.4	1705.8	2.921	1.794	669.3	1114.6	2.779	1052.1	1417.5	1.27	57	14.6	18	13800
7905	1.803	1443.0	1648.3	2.846	1.752	663.3	1088.7	2.728	1061.5	1374.5	1.23	56	14.6	19	13100



## APPENDIX B

### ACOUSTIC DATA, MEASURED MODEL AND SCALED TO JT8D ENGINE

This appendix contains the one-third octave band acoustic data obtained on the contract configurations. Two types of data are presented,

- a.) Measured Model Data: The as-measured data were corrected for microphone and cable frequency response calibrations, and are presented for the microphone distance of 15 ft. used in the test. Since the test chamber is anechoic, the data are free-field and are not contaminated by sound wave reflections.

Atmospheric absorption corrections were used to analytically correct data from test chamber conditions to FAA day (77°F, 70% relative humidity). These corrections were obtained from the formulas of ref. (3). The SPL values from 100 to 80,000 Hz were integrated to provide Overall Sound Pressure Levels (OSPL) for each angle.

- b.) Data Scaled To Predict Full Size JT8D Engine: In order to scale model data to predict full size engine noise, it is necessary first to eliminate the effects of atmospheric absorption on the measured noise spectra. This is done by determining the reduction of SPL levels due to the atmospheric absorption present during the model test, and adding these values to the measured levels, thus producing the true source noise spectra. Since the data were corrected to reflect the atmospheric absorption that would occur on an FAA day to present the model data on a consistent basis in step a.), the additional correction to obtain the true source spectra was added to the FAA day model data. Then the data were scaled to JT8D engine size (7 times linear model dimensions) by reducing the frequency values by a factor of 7.0 and by adding the factor  $20 \log(7)$  to the SPL level of each third-octave band. The scaled data then were extrapolated to 1200 ft. linear distance by subtracting  $20 \log \frac{1200}{15 \sin \theta}$  to account for spherical divergence. Atmospheric attenuation corrections necessary to present the full scale data on an FAA day were applied per ref. 3. The Overall Sound Pressure Level (OASPL) and Perceived Noise Levels (PNL) for each angle are listed at the bottom of the SPL values for each angle. Also listed are the PNL values for 400, 2000, 4000, and 6000 foot linear distances.

The heading information at the top of the data sheets contains pertinent nozzle information in both U.S. customary units as well as the International System of Units (S.I.). However, a more complete listing of conditions is presented in Appendix A. The identification number for each data point is found in the top right hand corner of each data table listed after condition.

To facilitate use of the acoustic data, the sheets have been assembled by increasing condition numbers. Thus, to locate the data for a particular configuration and operating conditions, first find the condition number from the tables of Appendix A.



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled To Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-2**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled To Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-3**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled To Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



# Basic Long Flowpath Mixer Without Engine Secondary Flow Simulation; Configuration 1A; Condition 5304

## a.) Model Data Measured At 15 Ft. Radius

STAND 1206 RIG ID 705367 TEST DATE 11/04/76 SCALE RATIO 7.6/1 RUN NUMBER 20053 CONDITION 5304														
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### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled To Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

[illegible]

**B-6**



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled To Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

[illegible]

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled To Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

**B-8**



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled To Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-9**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled To Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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## B-10



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-12**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-13**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-15**



# Reference Exhaust System Without Engine Secondary Flow Simulation; Configuration 2Au; Condition 5506

## a.) Model Data Measured At 15 Ft. Radius

STAND #206 RIG ID 709507 TEST DATE 11/03/76 SCALE RATIO 7.0/1 RUN NUMBER 76655 CONDITION 5506														
*****														
AREA SPT			PRIMARY FAN			SGR			PRIMARY FAN			MASS FLOW LB/S		
P.R.	1.06	1.79	1.06	1.79		1.06	1.79		1.06	1.79		THRUST, LBL	LB	
TEMP	161.1405.0	676.0	161.1405.0	676.0	161.1405.0	676.0	161.1405.0	676.0	161.1405.0	676.0	161.1405.0	THRUST, MEA	LB	
RND	18/913	0.432	0.070	AC/MS	0.512	1.110			AREA (INDI) SPT			W (INDI) LB/S	3.1	2.9
VCL	SPS 1700.2	1115.0	W/S	510.2	330.9									
*****														
1/3 OCTAVE BAND MODEL JET NOISE DATA 15.0 FT RADII														
*****														
BAND CENTER FREQ (Hz)	50	70	90	100	110	120	130	140	150	160	170	180	190	200
MICROPHONE ANGLES IN DEGREES														
0.050	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.063	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.080	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.100	79.6	84.6	84.7	85.3	85.6	86.4	87.9	90.9	92.6	96.2	100.3			
0.125	81.6	85.5	87.2	87.7	88.5	89.8	91.4	95.4	98.3	104.9				
0.160	86.2	88.6	90.8	91.5	92.7	94.5	97.4	100.9	106.4	109.7				
0.200	84.7	84.9	86.0	86.1	87.5	90.1	92.6	96.9	100.6	104.9				
0.250	86.2	86.9	88.2	89.0	90.6	93.4	95.9	97.9	103.0	107.1				
0.315	86.6	88.5	90.0	91.2	92.0	94.6	96.2	101.3	105.9	109.8				
0.400	90.8	91.7	93.2	94.0	95.8	98.6	100.2	104.6	109.4	113.6				
0.500	93.9	95.3	96.7	97.7	98.3	101.6	102.7	107.4	112.7	116.3				
0.630	96.0	97.7	98.7	99.3	100.4	104.2	107.4	114.0	119.4	123.6				
0.800	97.5	99.6	99.6	99.7	101.3	104.6	111.0	116.7	120.9	125.4				
1.000	96.0	97.4	97.9	99.3	101.1	103.6	107.0	113.0	118.1	121.1				
1.250	97.3	98.4	99.6	100.6	102.5	104.3	107.8	114.0	119.2	123.9				
1.600	98.4	99.6	100.0	100.8	102.7	104.9	108.0	113.5	118.9	123.7				
2.000	98.3	99.3	100.3	100.3	102.3	104.8	108.3	112.9	117.5	121.3				
2.500	96.9	97.7	98.0	100.3	102.3	105.6	108.6	112.3	116.1	119.5				
3.150	96.1	97.0	98.0	99.9	101.6	105.0	108.3	111.7	115.2	118.1				
4.000	95.4	96.0	96.8	98.0	101.0	104.6	108.1	110.7	114.2	117.1				
5.000	95.1	95.5	97.4	99.0	101.2	104.3	107.2	109.8	113.1	115.6				
6.300	94.6	95.5	97.0	98.4	100.6	103.6	106.6	107.5	108.0	109.9				
8.000	94.6	96.0	98.4	98.2	100.7	103.7	106.6	107.0	107.0	107.0				
10.000	91.9	94.0	96.4	97.2	99.2	102.2	104.9	104.1	104.1	99.9				
12.500	91.6	92.9	96.0	96.3	98.3	100.9	102.8	102.1	100.9	97.9				
16.000	91.6	93.5	95.3	97.1	99.0	100.6	100.6	98.9	97.7	94.9				
20.000	87.9	90.7	92.6	92.7	95.3	97.5	98.6	97.9	96.4	93.0				
25.000	86.2	89.0	90.6	92.3	94.1	95.7	94.9	95.9	94.5	90.6				
31.500	84.5	86.4	88.4	90.7	92.7	94.0	94.0	94.1	92.5	89.2				
40.000	82.7	85.1	87.6	89.2	90.6	92.5	91.3	92.6	91.0	89.0				
50.000	80.0	82.4	85.1	86.5	88.9	90.9	91.7	90.4	90.1	87.5				
63.000	78.3	80.3	82.4	84.3	85.7	88.0	88.3	88.0	90.2	88.4				
80.000	75.4	77.4	79.5	81.0	83.4	84.5	85.4	86.2	91.1	89.4				
100.000	70.0	70.0	70.0	70.0	70.0	70.0	70.0	70.0	70.0	70.0				
*****														
LSP: 106.6 106.9 110.6 111.4 115.7 115.7 119.7 122.0 127.0 129.2 129.5														

## b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance

STAND #206 RIG ID 709507 TEST DATE 11/03/76 SCALE RATIO 7.0/1 RUN NUMBER 76655 CONDITION 5506														
*****														
AREA SPT			PRIMARY FAN			SGR			PRIMARY FAN			MASS FLOW LB/S		
P.R.	1.06	1.79				1.06	1.79					THRUST, LBL	LB	
TEMP	161.1405.0	676.0				161.1405.0	676.0					THRUST, MEA	LB	
RND	18/913	0.432	0.070	AC/MS	0.512	1.110				AREA (INDI) SPT		W (INDI) LB/S	3.1	2.9
VCL	SPS 1700.2	1115.0	W/S	510.2	330.9									
*****														
1/3 OCTAVE BAND														
*****														
1260. SIDE LINE														
*****														
ANGLE IN DEGREES														
*****														
50	66.1	66.7	65.6	69.4	70.6	72.4	74.1	77.7	80.7	82.4	81.7			
65	68.3	69.9	71.6	72.7	74.4	76.0	77.7	81.0	84.6	86.0	85.2			
80	71.1	71.5	72.9	72.9	76.0	76.4	79.1	82.7	87.5	88.9	89.1			
95	73.3	73.3	76.7	76.7	76.0	78.5	81.5	85.0	89.4	90.3	83.6			
110	72.8	74.7	75.3	77.1	78.7	79.2	81.9	87.2	91.3	91.3	86.2			
125	74.6	75.3	76.3	77.0	78.5	81.5	84.7	89.1	92.6	91.2	85.0			
140	76.3	76.3	78.6	79.1	80.0	82.2	85.0	90.1	93.7	91.4	85.4			
155	75.5	77.7	78.3	79.7	80.8	82.7	85.1	89.4	93.2	91.0	85.9			
170	75.1	75.9	77.3	78.7	80.2	82.4	85.1	88.5	91.4	89.9	83.0			
185	73.6	76.2	76.7	78.4	80.4	82.5	85.3	87.8	89.9	87.7	80.9			
200	72.5	76.2	76.1	77.8	79.6	81.7	84.7	86.4	87.6	85.0	77.6			
215	71.6	73.0	75.6	77.2	79.3	81.9	84.5	85.1	85.3	81.4	73.1			
230	70.9	72.4	74.9	76.3	78.3	80.9	83.0	83.4	82.6	77.2	68.5			
245	69.7	71.7	73.7	75.7	77.5	79.0	81.9	81.4	79.0	73.0	63.0			
260	67.7	70.5	72.6	74.6	76.4	78.9	80.4	79.5	76.9	70.1	59.0			
275	66.0	68.0	70.0	72.0	73.9	76.7	77.1	77.0	74.0	71.0	60.2			
290	64.4	67.0	69.5	71.2	73.0	75.0	75.0	73.4	69.6	62.5	49.3			
305	61.8	64.0	67.2	69.2	70.8	72.8	72.7	70.3	65.9	59.3	43.4			
320	59.2	61.9	64.4	66.4	67.7	69.2	68.9	66.1	61.3	52.6	36.0			
335	56.5	58.0	61.3	63.1	64.0	65.5	65.2	61.0	56.0	46.4	28.0			
350	52.0	54.1	55.5	61.1	62.6	63.2	62.5	59.4	54.2	43.2	29.4			
365	47.0	52.1	55.7	57.4	58.7	59.5	58.4	54.6	46.2	38.1	17.9			
380	43.0	47.2	50.7	52.3	54.1	56.3	57.1	49.0	43.2	33.2	11.7			
395	38.7	42.4	45.6	48.0	49.7	49.2	45.7	40.5	33.7	20.8	0.0			
*****														
4000 SIDELINE														
PWL	102.5	104.0	104.0	100.5	110.2	122.2	113.5	114.7	116.1	113.0	107.5			
*****														
2000 SIDELINE														
PWL	84.0	86.1	86.0	87.7	91.4	93.4	95.4	97.2	99.0	96.0	90.0			
*****														
4000 SIDELINE														
PWL	74.7	74.9	75.0	80.0	87.1	84.1	86.2	86.4	90.6	80.2	80.0			
*****														
6000 SIDELINE														
PWL	60.0	71.0	72.7	74.4	76.1	78.0	80.2	82.9	85.2	82.5	76.1			



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-17**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-19**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

## B-20



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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B-21



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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B-22



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



# **Shallow Scalloped Long Flowpath Mixer Without Engine Secondary Flow Simulation; Configuration 3A; Condition 5706**

## **a.) Model Data Measured At 15 Ft. Radius**

STAMP 8706 SIG ID 705307 TEST DATE 11/10/76 SCALE RATIO 1.001 RUN NUMBER 7057 CONDITION 5706																			
PRIMARY FAN										SECONDARY FAN									
AREA	SAFT	1.70	1.70	1.70	1.70	1.70	1.70	1.70	1.70	AREA	SAFT	1.70	1.70	1.70	1.70	1.70	1.70	1.70	1.70
TEMP	101.1445.0	600.0	101.002.0	370.0	THROUST.10	10	0.0	N	0.0	TEMP	101.1445.0	600.0	101.002.0	370.0	THROUST.10	10	0.0	N	0.0
WGT	101.1445.0	600.0	101.002.0	370.0	WGT	101.1445.0	600.0	101.002.0	370.0	WGT	101.1445.0	600.0	101.002.0	370.0	WGT	101.1445.0	600.0	101.002.0	370.0

1/3 OCTAVE BAND MODEL AT 15 FT. RADIUS																			
CENTER FREQ	100	125	160	200	250	315	400	500	630	800	1000	1250	1600	2000	2500	3150	4000	5000	6300
100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
125	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
160	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
200	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
315	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
400	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
630	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
800	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1600	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3150	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6300	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

## **b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

STAMP 8706 SIG ID 705307 TEST DATE 11/10/76 SCALE RATIO 1.001 RUN NUMBER 7057 CONDITION 5706																			
PRIMARY FAN										SECONDARY FAN									
AREA	SAFT	1.70	1.70	1.70	1.70	1.70	1.70	1.70	1.70	AREA	SAFT	1.70	1.70	1.70	1.70	1.70	1.70	1.70	1.70
TEMP	101.1445.0	600.0	101.002.0	370.0	THROUST.10	10	0.0	N	0.0	TEMP	101.1445.0	600.0	101.002.0	370.0	THROUST.10	10	0.0	N	0.0
WGT	101.1445.0	600.0	101.002.0	370.0	WGT	101.1445.0	600.0	101.002.0	370.0	WGT	101.1445.0	600.0	101.002.0	370.0	WGT	101.1445.0	600.0	101.002.0	370.0

1/3 OCTAVE BAND MODEL AT 1200 FT. LINEAR DISTANCE																			
CENTER FREQ	100	125	160	200	250	315	400	500	630	800	1000	1250	1600	2000	2500	3150	4000	5000	6300
100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
125	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
160	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
200	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
315	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
400	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
630	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
800	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1600	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3150	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6300	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-26**



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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B-27



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



**Shallow Scalloped Long Flowpath Mixer without Engine Secondary  
Flow Simulation; Configuration 3A; Condition 5711**

**a.) Model Data Measured At 15 Ft. Radius**

STAND 2206 P/F 10 TO 105307 TEST DATE 11/10/76 SCALE RATIO 7.0/1 RUN NUMBER 24057 CONDITION 5711									
PRIMARY FAN									
AREA	SOFT	2.33	2.03	2.33	2.03	THROUST.124	14	14	14
P-2	181	1618.0	706.0	181	1618.0	332.2	THROUST.124	14	14
THRO	181	1618.0	706.0	181	1618.0	332.2	THROUST.124	14	14
WEL	181	1618.0	706.0	181	1618.0	332.2	THROUST.124	14	14
1/3 OCTAVE BAND MODEL JET NOISE DATA 15.0 FT RADIUS									
RECAPTURE ANGLES IN DEGREES									
ANGLE	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
7.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
10.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
11.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
12.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
13.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
14.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
15.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
16.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
17.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
18.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
19.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
20.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
21.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
22.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
23.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
24.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
25.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
26.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
27.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
28.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
29.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
30.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
31.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
32.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
33.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
34.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
35.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
36.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
37.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
38.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
39.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
40.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
41.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
42.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
43.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
44.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
45.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
46.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
47.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
48.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
49.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
50.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
51.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
52.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
53.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
54.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
55.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
56.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
57.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
58.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
59.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
60.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
61.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
62.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
63.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
64.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
65.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
66.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
67.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
68.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
69.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
70.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
71.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
72.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
73.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
74.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
75.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
76.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
77.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
78.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
79.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
80.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
81.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
82.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
83.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
84.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
85.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
86.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
87.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
88.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
89.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
90.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
91.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
92.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
93.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
94.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
95.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
96.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
97.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
98.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
99.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
100.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

STAND 2206 P/F 10 TO 105307 TEST DATE 11/10/76 SCALE RATIO 7.0/1 RUN NUMBER 24057 CONDITION 5711												
PRIMARY FAN												
AREA	SOFT	2.33		2.03	2.33		2.03	THROUST.124	14	14	14	14
P-2	181	1618.0	706.0	181	1618.0	332.2	THROUST.124	14	14	14	14	
THRO	181	1618.0	706.0	181	1618.0	332.2	THROUST.124	14	14	14	14	
WEL	181	1618.0	706.0	181	1618.0	332.2	THROUST.124	14	14	14	14	
PPS	195.175	1241.3	615.3	596.4	370.3	W (H2OILL)	14.1	7.1	3.3	1.5	1.5	
=====												
1/3 OCTAVE BAND												
RECAPTURE ANGLES IN DEGREES												
ANGLE	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
1.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
2.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
3.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
4.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
5.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
6.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
7.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
8.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
9.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
10.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
11.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
12.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
13.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
14.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
15.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
16.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
17.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
18.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
19.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
20.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
21.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
22.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
23.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
24.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
25.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
26.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
27.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
28.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
29.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
30.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
31.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
32.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
33.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
34.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
35.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
36.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
37.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
38.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
39.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
40.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
41.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
42.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
43.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
44.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
45.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
46.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
47.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
48.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
49.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
50.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
51.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
52.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
53.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
54.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
55.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
56.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
57.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
58.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
59.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
60.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
61.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
62.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
63.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
64.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
65.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
66.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
67.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
68.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
69.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
70.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
71.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
72.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
73.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
74.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
75.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
76.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
77.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
78.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
79.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
80.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
81.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
82.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
83.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
84.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
85.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
86.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
87.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
88.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
89.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
90.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
91.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
92.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
93.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
94.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
95.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
96.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
97.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
98.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
99.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
100.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
=====												
4000. SIRELINE												
ANGLE	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
1.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
2.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
3.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
4.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
5.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
6.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
7.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
8.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
9.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
10.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
11.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	
12.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0				



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



# Scalloped Long Flowpath Mixer Without Engine Secondary Flow Simulation; Configuration 4A; Condition 6111

## a.) Model Data Measured At 15 Ft. Radius

STAND 2706 RIG ID 705307 TEST DATE 11/27/76 SCALE RATIO 7.6/1 RUN NUMBER 70601 LOCATION 6111														
PRIMARY FAN														
AREA	100T	1.31	1.36	1.31	1.36	1.31	1.36	1.31	1.36	1.31	1.36	1.31	1.36	1.31
TEMP	181	1279.6	614.0	181	697.4	343.1	181	697.4	343.1	181	697.4	343.1	181	697.4
WET	181	1279.6	614.0	181	697.4	343.1	181	697.4	343.1	181	697.4	343.1	181	697.4
1/3 OCTAVE BAND MODEL														
JET NOISE DATA 15-FT RADIUS														
LINEAR DIST	60	70	80	90	100	110	120	130	140	150	160	170	180	190
ANGLE IN DEGREES	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
FAA DAY	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
4000. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
3000. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
2000. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
1000. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
500. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
250. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
125. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
62.5. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
31.25. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
15.6. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
7.8. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
3.9. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
1.95. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.975. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.4875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.24375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.121875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0609375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.03046875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.015234375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0076171875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00380859375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.001904296875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0009521484375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00047607421875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000238037109375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0001190185546875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00005950927734375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000029754638671875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0000148773193359375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000743865966796875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000003719329833984375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0000018596649169921875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000092983245849609375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000000464916229248046875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0000002324581146240234375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000011622905731201171875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000000058114528656008559375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0000000290572643280042796875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000001452863216400213984375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000000007264316082001069921875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0000000036321580410005349609375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000181607902050026748046875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000090803951025001337240234375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0000000004540197551250006686171875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000022700987756250033430859375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000011350493878125000167167169375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000005675246939062500083583584375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000002837623469531250004179179171875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000000000014188117347656250002089589589375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0000000000070940586738281250001044794479375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000000354702933691406250000522397239375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000000000001773514668457812500002611986198375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000000000000886757334228906250000130599309375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000000044337866711445312500000652996546875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000000000000221689333557221671875000003264982734375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000000011084466677858593750000016324941171875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000000000000055422333389292968750000008162470589375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000000002771116669464648437500000408123527946875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000000000000013855583347323242187500000204061763734375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0000000000000069277916736616109375000001020308818671875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0000000000000034638958368305468750000005101544089375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000000000173194791841527343750000002550772046875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000000000086597395920761937500000012753860234375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000000000000000432986979603809375000000063769301171875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.000000000000000216493489801904687500000031884650589375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0000000000000001082467449009523437500000159423252946875. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.00000000000000005412337245047619375000000797116264734375. SIDELINE	50.0	51.0	52.0	53.0	54.0	55.0	56.0	57.0	58.0	59.0	60.0	61.0	62.0	63.0
0.0000000000000000270616862252380														



# Scalloped Long Flowpath Mixer Without Engine Secondary Flow Simulation; Configuration 4A; Condition 6112

## a.) Model Data Measured At 15 Ft. Radius

STAND 8206 RIG ID 705307 TEST DATE 11/27/76 SCALE RATIO 7.0/1 RUN NUMBER 20003 CONDITION 6112														
*****														
PRIMARY FAN					PRIMARY FAN					PRIMARY FAN				
AREA	10FT	1.45	1.47	1.45	1.47	1.45	1.47	1.45	1.47	1.45	1.47	1.45	1.47	1.45
P.B.	10FT	1297.0	617.0	101	770.0	342.0	101	770.0	342.0	101	770.0	342.0	101	770.0
END	10FT	0.016	0.072	0.072	0.072	0.072	0.072	0.072	0.072	0.072	0.072	0.072	0.072	0.072
VEL	10FT	1259.0	600.0	101	770.0	342.0	101	770.0	342.0	101	770.0	342.0	101	770.0
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1/3 OCTAVE BAND MODEL JET NOISE DATA 15.0FT RADIUS														
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**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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B-33



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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B-34



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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B-35



# Scalloped Long Flowpath Mixer Without Engine Secondary Flow Simulation; Configuration 4A; Condition 6116

## a.) Model Data Measured At 15 Ft. Radius

STAND 2206 RIG ID T05307 TEST DATE 11/29/76 SCALE RATIO 7.0/1 RUN NUMBER 20661 CONDITION 6116														
PRIMARY FAN PRIMARY FAN														
AREA	SOFT	1.00	1.70	1.00	1.70	MASS FLOW	LB/S	130.7	100.5	60.5	62.9	65.0	65.0	65.0
TEMP	181	1467.0	600.0	181	815.0	371.7	THURST,IDE	LB	0.0	0.0	0.0	0.0	0.0	0.0
END	18/113	0.032	0.070	18/113	0.511	1.127	AREA (INCH)	SOFT	2.8	2.9	2.9	2.9	2.9	2.9
VEL	FPS	1700.7	1110.0	FPS	510.7	330.5	W (INCH)	LB/S	2.8	2.9	2.9	2.9	2.9	2.9
1/3 OCTAVE BAND MODEL JET NOISE DATA 15.0 FT RADIUS														
BAND	1000	1250	1500	1750	2000	2250	2500	2750	3000	3250	3500	3750	4000	4250
1000	1250	1500	1750	2000	2250	2500	2750	3000	3250	3500	3750	4000	4250	4500
1000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1750	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2750	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3750	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4750	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5750	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6750	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
7000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
7250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
7500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
7750	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8750	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9750	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
10000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

## b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance

STAND 2206 RIG ID T05307 TEST DATE 11/29/76 SCALE RATIO 7.0/1 RUN NUMBER 20661 CONDITION 6116														
PRIMARY FAN PRIMARY FAN														
AREA	SOFT	1.00	1.70	1.00	1.70	MASS FLOW	LB/S	130.7	100.5	60.5	62.9	65.0	65.0	65.0
TEMP	181	1467.0	600.0	181	815.0	371.7	THURST,IDE	LB	0.0	0.0	0.0	0.0	0.0	0.0
END	18/113	0.032	0.070	18/113	0.511	1.127	AREA (INCH)	SOFT	2.8	2.9	2.9	2.9	2.9	2.9
VEL	FPS	1700.7	1110.0	FPS	510.7	330.5	W (INCH)	LB/S	2.8	2.9	2.9	2.9	2.9	2.9
1/3 OCTAVE BAND														
1200	1500	1800	2100	2400	2700	3000	3300	3600	3900	4200	4500	4800	5100	5400
1200	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1800	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2400	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2700	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3300	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3600	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3900	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4200	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4800	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5400	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5700	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6300	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6600	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6900	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
7200	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
7500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
7800	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8400	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8700	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9300	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9600	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
9900	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
10200	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
10500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
10800	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
11100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
11400	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
11700	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
12000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-37**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-38**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-39**



# Scalloped Long Flowpath Mixer Without Engine Secondary Flow Simulation; Configuration 4A; Condition 6120

## a.) Model Data Measured At 15 Ft. Radius

STAND 2106 RIG ID 709307 TEST DATE 11/29/76 SCALE RATIO 7.0/1 RUN NUMBER 70001 CONDITION 6120									
PRIMARY FAN									
AREA	SOFT	2.94	1.96	SOFT	0.273	0.182	MASS FLOW LB/S	155.8	151.0
P.W.	2.15	2.00	THROUST. DIA. IN	18	18	18	THROUST. DIA. IN	18	18
TEMP	181	1600.0	490.0	181	692.2	387.0	AREA (SQ IN)	0.06	0.06
DIS	1.8713	0.030	0.070	0.030	0.030	0.030	W (INCHES)	1.14	1.14
VIL	12	1000.0	1270.0	1270.0	1270.0	1270.0	W (INCHES)	1.14	1.14
1/3 OCTAVE BAND									
FAR DAY									
1200- 51061302									
MILES IN DEGREES									
50.0	70.0	80.0	90.0	100.0	110.0	120.0	130.0	140.0	150.0
50	68.1	68.7	70.7	72.0	73.5	74.4	76.1	79.2	82.6
60	70.0	72.2	74.5	76.9	79.4	81.9	84.4	87.9	91.4
70	73.0	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0
80	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
90	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
100	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
110	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
120	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
130	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
140	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
150	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
160	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
170	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
180	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
190	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
200	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
210	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
220	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
230	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
240	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
250	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
260	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
270	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
280	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
290	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
300	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
310	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
320	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
330	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
340	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
350	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
360	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
370	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
380	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
390	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
400	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
410	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
420	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
430	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
440	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
450	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
460	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
470	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
480	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
490	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
500	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
510	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
520	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
530	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
540	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
550	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
560	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
570	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
580	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
590	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
600	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
610	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
620	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
630	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
640	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
650	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
660	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
670	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
680	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
690	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
700	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
710	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
720	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
730	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
740	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
750	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
760	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
770	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
780	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
790	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
800	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
810	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
820	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
830	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
840	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
850	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
860	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
870	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
880	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
890	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
900	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
910	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
920	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
930	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
940	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
950	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
960	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
970	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
980	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
990	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
1000	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0

## b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance

STAND 2106 RIG ID 709307 TEST DATE 11/29/76 SCALE RATIO 7.0/1 RUN NUMBER 70001 CONDITION 6120									
PRIMARY FAN									
AREA	SOFT	2.94	1.96	SOFT	0.273	0.182	MASS FLOW LB/S	155.8	151.0
P.W.	2.15	2.00	THROUST. DIA.	18	18	18	THROUST. DIA.	18	18
TEMP	181	1600.0	490.0	181	692.2	387.0	AREA (SQ IN)	0.06	0.06
DIS	1.8713	0.030	0.070	0.030	0.030	0.030	W (INCHES)	1.14	1.14
VIL	12	1000.0	1270.0	1270.0	1270.0	1270.0	W (INCHES)	1.14	1.14
1/3 OCTAVE BAND									
FAR DAY									
1200- 51061302									
MILES IN DEGREES									
50.0	70.0	80.0	90.0	100.0	110.0	120.0	130.0	140.0	150.0
50	68.1	68.7	70.7	72.0	73.5	74.4	76.1	79.2	82.6
60	70.0	72.2	74.5	76.9	79.4	81.9	84.4	87.9	91.4
70	73.0	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0
80	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
90	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
100	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
110	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
120	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
130	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
140	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
150	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
160	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
170	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
180	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
190	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
200	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
210	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
220	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
230	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
240	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
250	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
260	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
270	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
280	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
290	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
300	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
310	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
320	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
330	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
340	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
350	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
360	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
370	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
380	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
390	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
400	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
410	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
420	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
430	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
440	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
450	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
460	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
470	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
480	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
490	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
500	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
510	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
520	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
530	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
540	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
550	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
560	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
570	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
580	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
590	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
600	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
610	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
620	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
630	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
640	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
650	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
660	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
670	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
680	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
690	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
700	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
710	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
720	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
730	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
740	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
750	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
760	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
770	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
780	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
790	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
800	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
810	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
820	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
830	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
840	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
850	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
860	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
870	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
880	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
890	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
900	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
910	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
920	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
930	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
940	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
950	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
960	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
970	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
980	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
990	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
1000	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
1/3 OCTAVE BAND									
FAR DAY									
1200- 51061302									
MILES IN DEGREES									
50.0	70.0	80.0	90.0	100.0	110.0	120.0	130.0	140.0	150.0
50	68.1	68.7	70.7	72.0	73.5	74.4	76.1	79.2	82.6
60	70.0	72.2	74.5	76.9	79.4	81.9	84.4	87.9	91.4
70	73.0	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0
80	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
90	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
100	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
110	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
120	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
130	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
140	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
150	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
160	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
170	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
180	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
190	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
200	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
210	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
220	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
230	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
240	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
250	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
260	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
270	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
280	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
290	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
300	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
310	75.0	77.0	79.0	81.0	83.0	85.0	87.0	89.0	91.0
320	75.0	77.0	79.0	81					



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-41**



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-42**



# Severe Cutback Non-Scalloped Long Flowpath Mixer Without Engine Secondary Flow Simulation; Configuration 6A; Condition 6303

## a.) Model Data Measured At 15 Ft. Radius

STAND 8200 P/N 11 670530 TEST DATE 12/01/76 SCALE RATIO 1.000 T-071 RUN NUMBER 10000 CONDITION 6303									
PRIMARY FAN					SECONDARY FAN				
AREA	SAFT	1.40	1.52	SON	1.40	1.52	MASS FLOW LB/S	101.4	135.7
TEMP	181	1336.0	624.0	181	743.1	346.7	THURST/INCH LB	6.6	8
END	18/113	0.033	0.072	0.033	0.032	1.155	AREA (INCH) SAFT	7.1	7.0
WEL	195	1304.9	619.3	195	730.2	340.2	W (INCH) LB/S	7.1	7.0
1/3 OBTAIN DATA ALL NOISE DATA 15.0 FT RADIUS									
RECIPIENT ANGLES IN DEGREES									
ANGLE	0.0	30.0	60.0	90.0	120.0	150.0	180.0	210.0	240.0
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
30.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
60.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
90.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
120.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
150.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
180.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
210.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
240.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
270.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
300.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
330.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
360.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
390.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
420.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
450.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
480.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
510.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
540.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
570.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
600.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
630.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
660.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
690.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
720.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
750.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
780.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
810.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
840.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
870.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
900.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
930.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
960.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
990.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1020.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1050.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1080.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1110.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1140.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1170.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1200.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1230.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1260.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1290.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1320.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1350.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1380.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1410.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1440.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1470.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1500.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

## b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance

STAND 8200 P/N 11 670530 TEST DATE 12/01/76 SCALE RATIO 1.000 T-071 RUN NUMBER 10000 CONDITION 6303									
PRIMARY FAN					SECONDARY FAN				
AREA	SAFT	1.40	1.52	SON	1.40	1.52	MASS FLOW LB/S	101.4	135.7
TEMP	181	1336.0	624.0	181	743.1	346.7	THURST/INCH LB	6.6	8
END	18/113	0.033	0.072	0.033	0.032	1.155	AREA (INCH) SAFT	7.1	7.0
WEL	195	1304.9	619.3	195	730.2	340.2	W (INCH) LB/S	7.1	7.0
1/3 OBTAIN DATA ALL NOISE DATA 15.0 FT RADIUS									
RECIPIENT ANGLES IN DEGREES									
ANGLE	0.0	30.0	60.0	90.0	120.0	150.0	180.0	210.0	240.0
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
30.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
60.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
90.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
120.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
150.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
180.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
210.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
240.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
270.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
300.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
330.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
360.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
390.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
420.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
450.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
480.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
510.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
540.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
570.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
600.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
630.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
660.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
690.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
720.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
750.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
780.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
810.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
840.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
870.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
900.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
930.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
960.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
990.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1020.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1050.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1080.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1110.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1140.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1170.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1200.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1230.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1260.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1290.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1320.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1350.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1380.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1410.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1440.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1470.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1500.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

B-44



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-46**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-47**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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B-48



# Severe Cutback Non-Scalloped Long Flowpath Mixer Without Engine Secondary Flow Simulation; Configuration 6A; Condition 6309

## a.) Model Data Measured At 15 Ft. Radius

STAND 2706 PIC 10 670530 TEST DATE 12/01/76 SCALE RATIO 7.0/1 RUN NUMBER 20063 LUNATION 0309														
PRIMARY FAN														
AREA	SQFT	P.W.	1.00	SQR	2.13	1.00	MASS FLOW	LB/S	159.7	137.3	KG/S	72.5	78.3	
TEMP	(F)	1005.0	400.0	(K)	541.7	387.0	THROUST. IDL	4.0						
WPM	LB/FT2	0.030	0.009	KG/MS	0.402	1.113	AREA (FWD)	56.1						
VEL	FPS	1700.7	1226.3	M/S	593.3	375.0	W (MODEL)	1.0/5	3.3	3.2	0.6/5	1.5	1.5	
L/3 0.154V BAND MODEL JET NOISE DATA 15.0 FT. RADIUS														
BAND CENTER FREQ. (Hz)	60	70	80	90	100	110	120	130	140	150	160	170	180	190
150	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
160	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
170	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
180	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
190	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
200	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
210	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
220	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
230	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
240	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
260	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
270	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
280	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
290	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
300	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
310	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
320	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
330	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
340	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
350	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
360	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
370	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
380	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
390	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
400	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
410	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
420	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
430	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
440	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
450	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
460	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
470	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
480	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
490	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
510	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
520	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
530	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
540	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
550	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
560	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
570	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
580	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
590	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
600	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
610	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
620	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
630	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
640	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
650	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
660	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
670	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
680	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
690	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
700	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
710	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
720	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
730	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
740	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
750	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
760	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
770	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
780	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
790	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
800	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
810	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
820	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
830	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
840	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
850	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
860	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
870	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
880	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
890	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
900	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
910	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
920	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
930	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
940	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
950	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
960	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
970	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
980	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
990	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1010	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1020	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1030	0.0	0.0	0.0	0										



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-50**



# **Cutback Scalloped Long Flowpath Mixer Without Engine Secondary** **Flow Simulation; Configuration 5A; Condition 6501**

## **a.) Model Data Measured At 15 Ft. Radius**

STAND 2206 RLE ID 703307 TEST DATE 12/03/76 SCALE RATIO 1.0/1 RUN NUMBER 20000 CONDITION 6501														
PRIMARY FAN PRIMARY FAN														
AREA	SUP	SON	MASS FLOW LB/S	THRUST LBS	AREA (INCH) SUP	AREA (INCH) SON	W (INCH) LB/S	W (INCH) LB/S	W (INCH) LB/S	W (INCH) LB/S	W (INCH) LB/S	W (INCH) LB/S	W (INCH) LB/S	W (INCH) LB/S
P.R.	1.31	1.36	1.36	1.36	1.36	1.36	1.36	1.36	1.36	1.36	1.36	1.36	1.36	1.36
TIME	183	127.0	0.034	0.072	AL/MS	0.344	1.159	AREA (INCH) SUP	W (INCH) LB/S	1.7	2.4	AL/MS	0.0	1.0
RWD	18/PT3	0.034	0.072	AL/MS	0.344	1.159	AREA (INCH) SUP	W (INCH) LB/S	1.7	2.4	AL/MS	0.0	1.0	1.0
VEL	875	1000.0	701.5	875	325.2	230.2	W (INCH) LB/S	1.7	2.4	AL/MS	0.0	1.0	1.0	1.0

1/3 OCTAVE BAND MODEL JET NOISE DATA 15.0 FT RADIUS														
CENTER FREQ (Hz)	60	70	80	90	100	110	120	130	140	150	160	170	180	190
0.050	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.063	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.080	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.125	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.160	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.200	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.315	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.400	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.630	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.800	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1.250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1.600	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2.500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3.150	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6.300	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
10.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
12.500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
16.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
20.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
25.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
31.500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
40.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
50.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
63.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
80.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
100.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

DATA = 124.7

## **b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

STAND 2206 RLE ID 703307 TEST DATE 12/03/76 SCALE RATIO 1.0/1 RUN NUMBER 20000 CONDITION 6501														
PRIMARY FAN PRIMARY FAN														
AREA	SUP	SON	MASS FLOW LB/S	THRUST LBS	AREA (INCH) SUP	AREA (INCH) SON	W (INCH) LB/S	W (INCH) LB/S	W (INCH) LB/S	W (INCH) LB/S	W (INCH) LB/S	W (INCH) LB/S	W (INCH) LB/S	W (INCH) LB/S
P.R.	1.31	1.36	1.36	1.36	1.36	1.36	1.36	1.36	1.36	1.36	1.36	1.36	1.36	1.36
TIME	183	127.0	0.034	0.072	AL/MS	0.344	1.159	AREA (INCH) SUP	W (INCH) LB/S	1.7	2.4	AL/MS	0.0	1.0
RWD	18/PT3	0.034	0.072	AL/MS	0.344	1.159	AREA (INCH) SUP	W (INCH) LB/S	1.7	2.4	AL/MS	0.0	1.0	1.0
VEL	875	1000.0	701.5	875	325.2	230.2	W (INCH) LB/S	1.7	2.4	AL/MS	0.0	1.0	1.0	1.0

1/3 OCTAVE BAND														
CENTER FREQ (Hz)	60	70	80	90	100	110	120	130	140	150	160	170	180	190
0.050	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.063	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.080	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.125	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.160	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.200	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.315	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.400	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.630	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.800	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1.250	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
1.600	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
2.500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
3.150	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
4.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
5.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
6.300	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
8.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
10.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
12.500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
16.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
20.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
25.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
31.500	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
40.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
50.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
63.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
80.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
100.000	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

DATA = 124.7

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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B-53



**a.) Model Data Measured At 15 Ft. Radius**

[illegible]

**b.) Model Data Scaled to Predict JTCD Engine Jet Noise At 1200 Ft. Linear Distance**

[illegible]

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



**Flow Simulation; Configuration 5A; Condition 6505**

a.) Model Data Measured At 15 Ft. Radius

[illegible]

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

[illegible]

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-57**



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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B-58



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



# Reference Exhaust System Tested at Ideally Mixed Conditions Without Engine Secondary Flow Simulation; Configuration 2Am; Condition 7401

## a.) Model Data Measured At 15 Ft. Radius

STAND 2206 RIG ID 709307 TEST DATE 07/10/77 SCALE RATIO 7.0/1 RUN NUMBER 20074 CONDITION 7401														
*****														
PRIMARY FAN														
AREA	SQFT	1.44	1.45	1.44	1.45	1.44	1.45	1.44	1.45	1.44	1.45	1.44	1.45	1.44
P.A.	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
TEMP	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
WIND	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
VEL	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
*****														
1/3 OCTAVE BAND MODEL JET NOISE DATA 15.0 FT. RADIUS														
*****														
MICROPHONE ANGLES IN DEGREES														
*****														
RAW CORRELATED SPL - (INCHES)														
*****														
*****														
CENTER FREQ	60	70	80	90	100	110	120	130	140	150	160	170	180	190
(HRTZ)	60	70	80	90	100	110	120	130	140	150	160	170	180	190
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
-100	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.						



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



**a.) Model Data Measured At 15 Ft. Radius**

1391 103.2 104.1 105.1 106.4 108.1 116.2 117.7 114.6 117.5 126.5 121.2

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-63**



### a.) Model Data Measured At 15 Ft. Radius

1291 105.0 105.0 106.0 106.2 109.9 112.7 116.3 116.6 119.7 123.0 123.0

GAPOC - 136-

[illegible][illegible]

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**B-64**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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B-65



**a.) Model Data Measured At 15 Ft. Radius**

109-1 109-2 109-3 109-4 110-1 110-2 110-3 110-4 111-1 111-2 111-3 111-4

[illegible]

**B-66**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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**B-67**



### a.) Model Data Measured At 15 Ft. Radius

#### b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance

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B-68



a.) Model Data Measured At 15 Ft. Radius

b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



#### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (i.e., 70% RH)



AD-A057 310

PRATT AND WHITNEY AIRCRAFT GROUP EAST HARTFORD CONN  
INTERNAL MIXER INVESTIGATION FOR JT8D ENGINE JET NOISE REDUCTIO--ETC(U)  
DEC 77 A B PACKMAN, D C EILER  
PWA-5582-VOL-2

F/G 20/1

DOT-FA76WA-3809

UNCLASSIFIED

FAA-RD-77-132-2

NL

2 OF 2

AD  
A057310





**a.) Model Data Measured At 15 Ft. Radius**

b.) Model Data Scaled to Predict IT3D Engine Jet Noise At 1200 Ft. Linear Distance

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 76% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model D, ta Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



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**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

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a.) Model Data Measured At 15 Ft. Radius

[illegible]

CPA 107.4 107.5 107.6 107.7 110.0 111.3 113.8 116.0 119.1 121.7 121.8

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

[illegible]

B-93



### a.) Model Data Measured At 15 Ft. Radius

[illegible]

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

[illegible]

B-94



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

[illegible]

**B-95**



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

[illegible]

**B-97**



### a.) Model Data Measured At 15 Ft. Radius

b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



**a.) Model Data Measured At 15 Ft. Radius**

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

**B-102**



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

#### b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



### a.) Model Data Measured At 15 Ft. Radius

**b.) Model Data Scaled to Predict JT8D Engine Jet Noise At 1200 Ft. Linear Distance**

Measured model and scaled data are freefield and have been adjusted to reflect the atmospheric absorption present on an FAA day (77°F, 70% RH)



## **APPENDIX C**

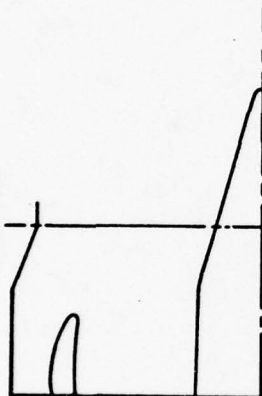
### **NOZZLE TRAVERSE RESULTS**

The figures contained in this section present the results of nozzle traverse testing conducted under the FAA contract as described in Section 3.

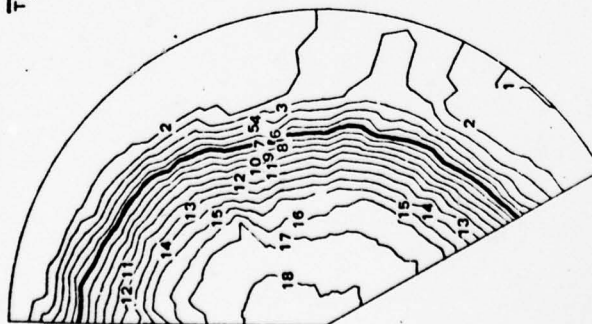
Contour plots of total temperature, total pressure and calculated fully expanded velocity are presented with contour line values normalized to average temperature or pressure and the calculated ideally mixed velocity. Test results are based on testing at takeoff nozzle pressure ratio.



"M" FLANGE



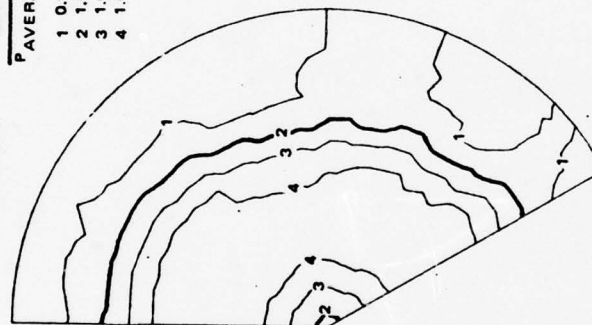
TEMPERATURE



$\frac{T_{LOCAL}}{T_{AVERAGE}}$

- 1 0.70
- 2 0.75
- 3 0.80
- 4 0.85
- 5 0.90
- 6 0.95
- 7 1.00
- 8 1.05
- 9 1.10
- 10 1.15
- 11 1.20
- 12 1.25
- 13 1.30
- 14 1.35
- 15 1.40
- 16 1.45
- 17 1.50
- 18 1.55

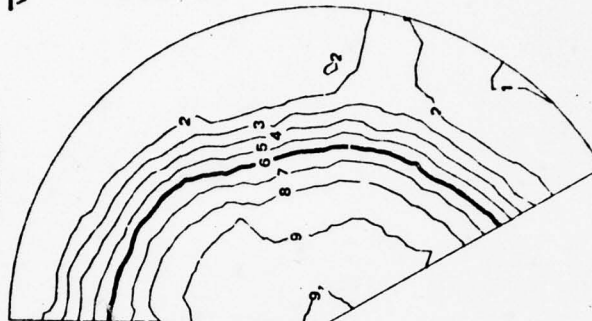
PRESSURE



$\frac{P_{LOCAL}}{P_{AVERAGE}}$

- 1 0.98
- 2 1.00
- 3 1.02
- 4 1.04

VELOCITY



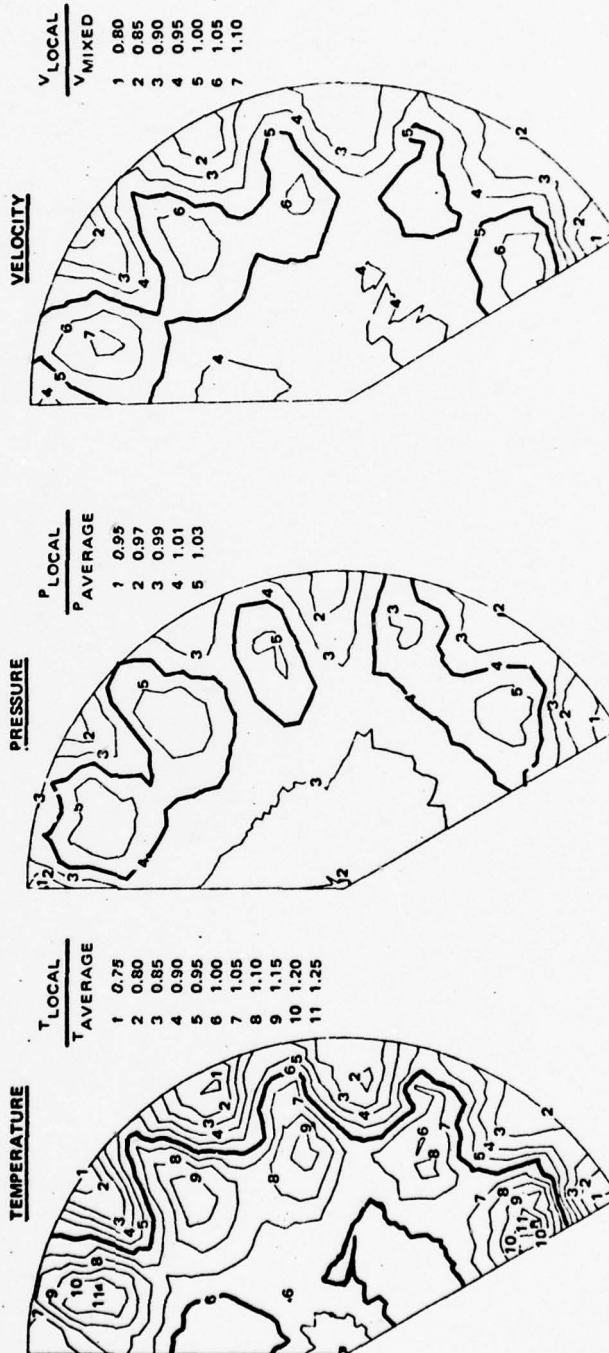
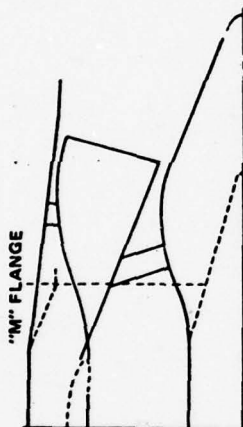
$\frac{V_{LOCAL}}{V_{MIXED}}$

- 1 0.75
- 2 0.80
- 3 0.85
- 4 0.90
- 5 0.95
- 6 1.00
- 7 1.05
- 8 1.10
- 9 1.15

EXHAUST NOZZLE TRAVERSE RESULTS  
REFERENCE EXHAUST SYSTEM  
CONFIGURATION 2A

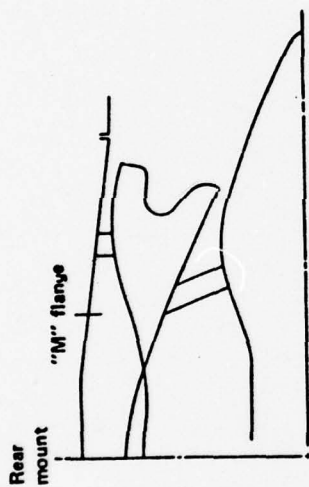


— BASIC LONG FLOWPATH MIXER  
 - - - REFERENCE EXHAUST SYSTEM



EXHAUST NOZZLE TRAVERSE RESULTS  
 BASIC LONG FLOWPATH MIXER  
 CONFIGURATION 1A

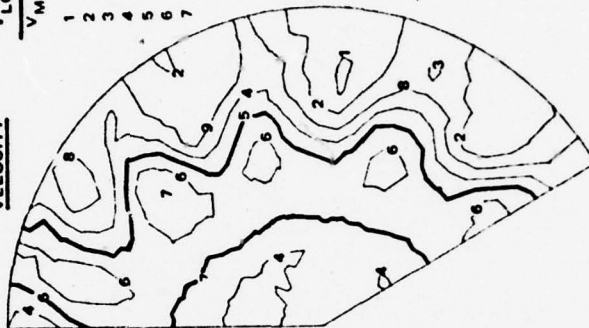




$V_{LOCAL}$   
 $V_{MIXED}$

1	0.80
2	0.85
3	0.90
4	0.95
5	1.00
6	1.05
7	1.10

VELOCITY



$P_{LOCAL}$   
 $P_{AVERAGE}$

1	0.96
2	0.98
3	1.00
4	1.02
5	1.04

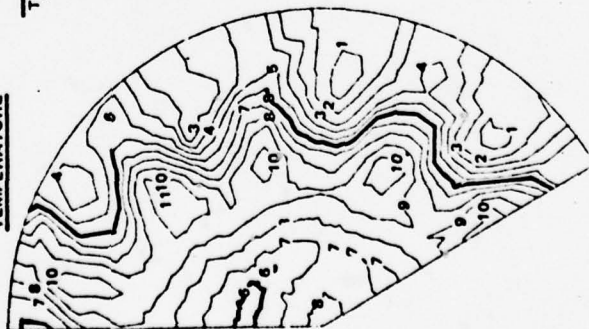
PRESSURE



$T_{LOCAL}$   
 $T_{AVERAGE}$

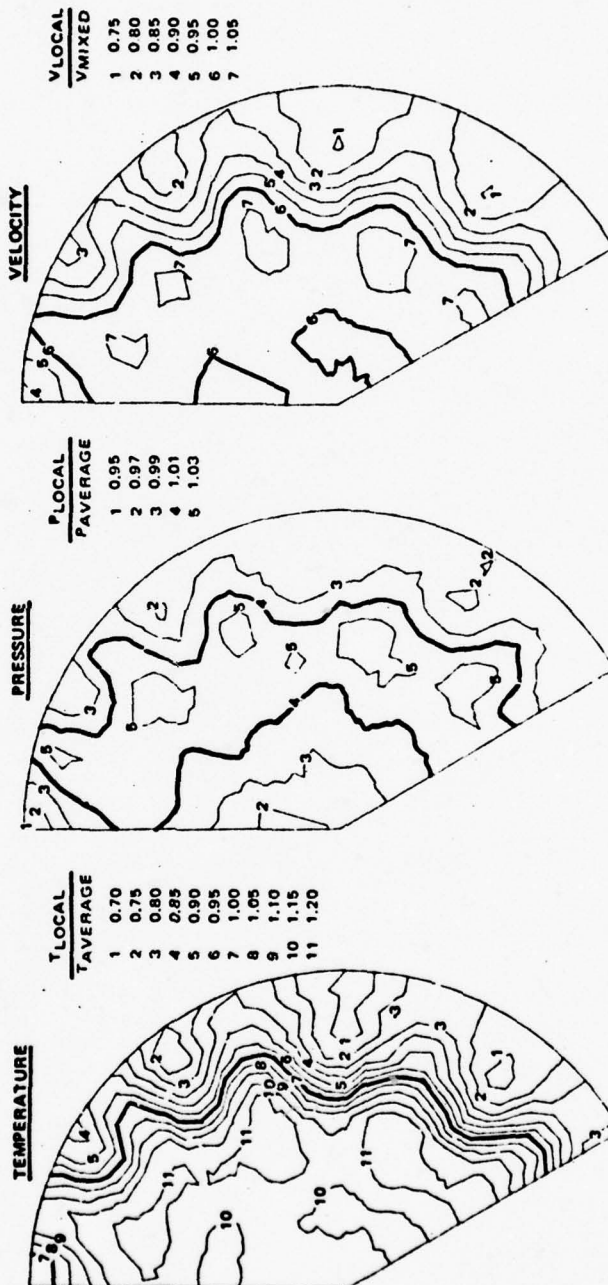
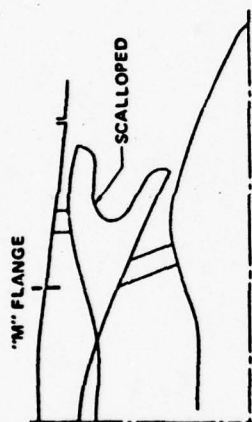
1	0.75
2	0.80
3	0.85
4	0.90
5	0.95
6	1.00
7	1.05
8	1.10
9	1.15
10	1.20
11	1.25

TEMPERATURE



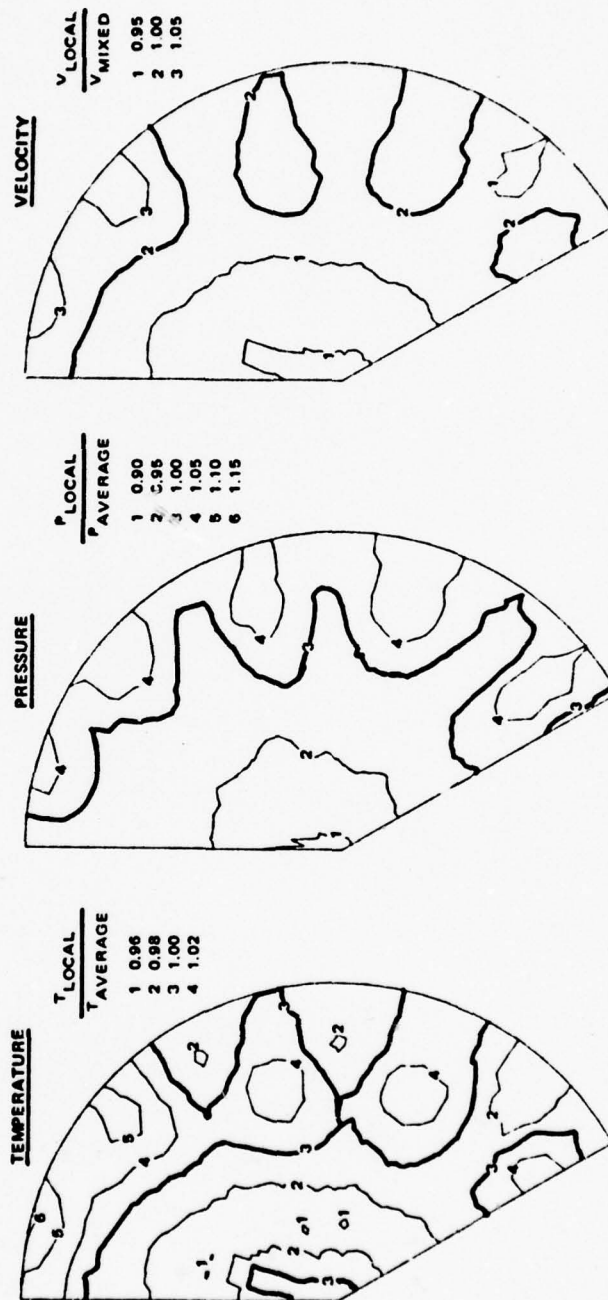
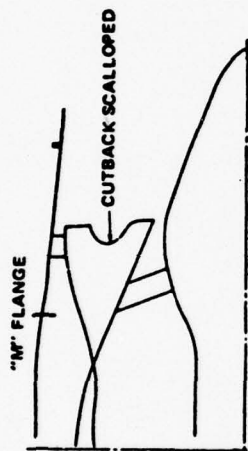
EXHAUST NOZZLE TRAVERSE RESULTS  
SHALLOWED SCALLOPED LONG FLOWPATH MIXER  
CONFIGURATION 3A





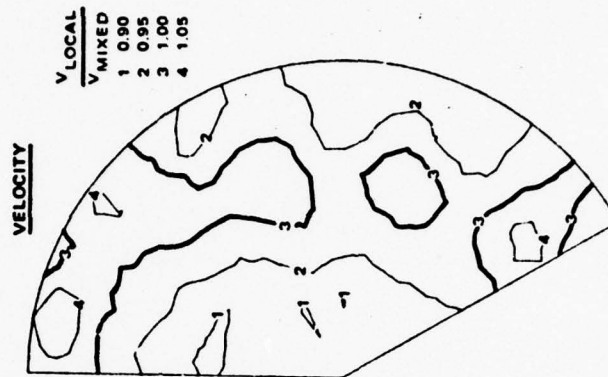
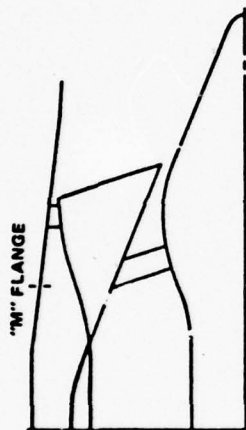
EXHAUST NOZZLE TRAVERSE RESULTS  
SCALLOPED LONG FLOWPATH MIXER  
CONFIGURATION 4A





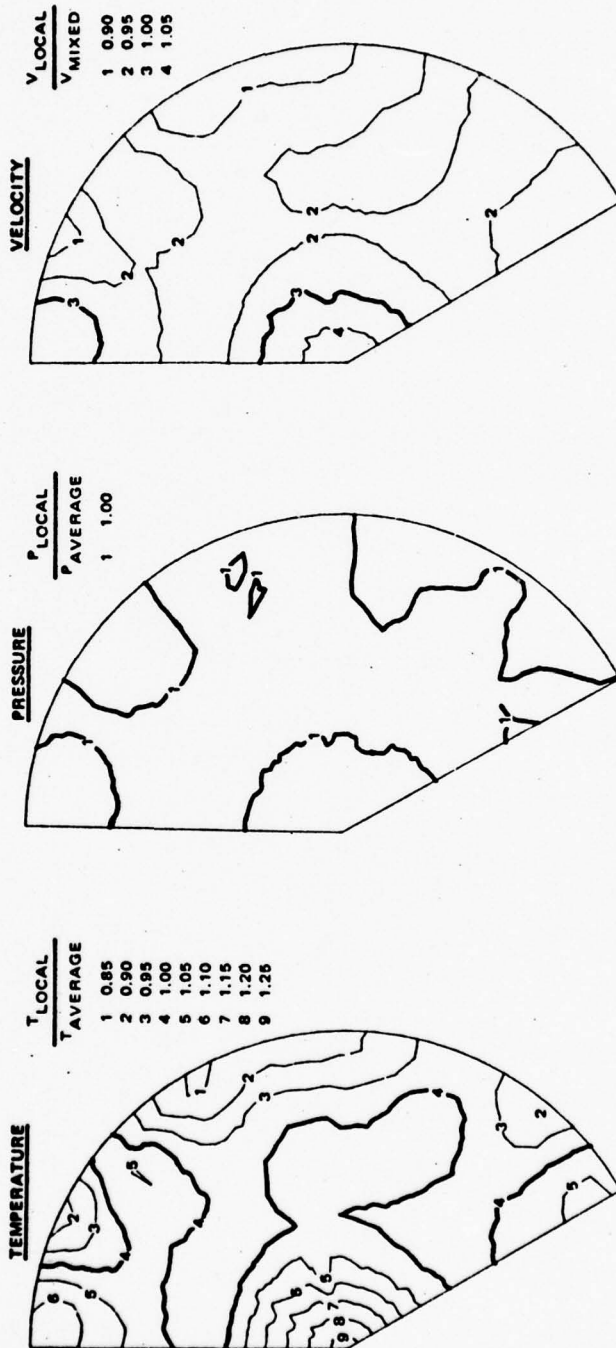
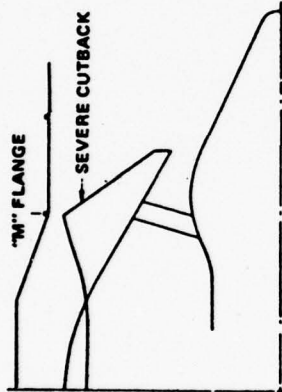
EXHAUST NOZZLE TRAVERSE RESULTS  
CUTBACK SCALLOPED LONG FLOWPATH MIXER  
CONFIGURATION 5A





EXHAUST NOZZLE TRAVERSE RESULTS  
SEVERE CUTBACK LONG FLOWPATH MIXER  
CONFIGURATION 6A

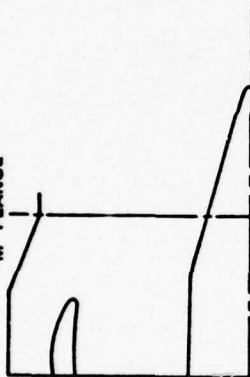




EXHAUST NOZZLE TRAVERSE RESULTS  
SEVERE CUTBACK SHORT FLOWPATH MIXER WITH SECONDARY FLOW SIMULATION  
CONFIGURATION 10A

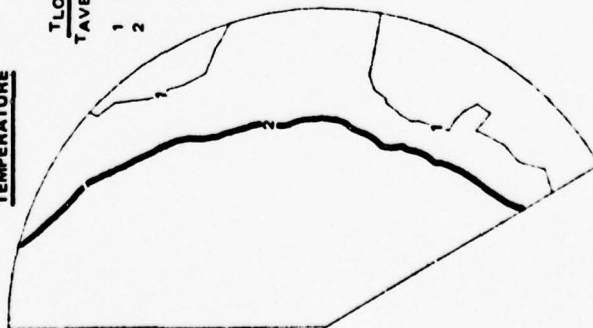


"M" FLANGE



TEMPERATURE

$\frac{T_{LOCAL}}{T_{AVERAGE}}$   
1 0.95  
2 1.00



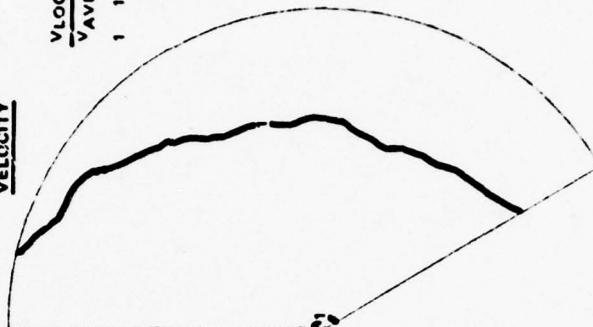
PRESSURE

$\frac{P_{LOCAL}}{P_{AVERAGE}}$   
1 0.98  
2 1.00



VELOCITY

$\frac{V_{LOCAL}}{V_{AVERAGE}}$   
1 1.00



EXHAUST NOZZLE TRAVERSE RESULTS REFERENCE EXHAUST  
NOZZLE SYSTEM TESTED AT IDEALLY MIXED CONDITIONS  
CONFIGURATION 2A<sub>m</sub>



**APPENDIX D**

**HOT/COLD FLOW MODEL TESTS  
TO DETERMINE STATIC PERFORMANCE  
OF 1/7-SCALE JT8D MIXER EXHAUST NOZZLES**

**Prepared by**

**FLUIDYNE ENGINEERING CORPORATION**



# **FLUIDDYNE ENGINEERING CORPORATION**

## **SUMMARY**

This report presents the results of hot/cold flow, 1/7-scale model tests conducted to determine static performance of two mixer exhaust nozzles for a JT8D engine. The test program was conducted by FluidDyne Engineering Corporation for Pratt & Whitney Division of United Technologies Corporation. The model tests were performed in the Channel 11 static thrust stand at the FluidDyne Medicine Lake Aerodynamic Laboratory.

Three model configurations were tested. Static performance was determined for a free mixer (Reference) and two 12-lobe mixers.

Test conditions included core nozzle pressure ratios from  $\lambda_g = 1.6$  to 3.2, and core-to-fan total temperature ratios from 1.0 to 2.41. Fan-to-core total pressure ratios were nominally .88, .93, and 1.00. The test program included 69 performance tests.

Facility checkout tests were made using two standard ASME long-radius metering nozzles. These tests demonstrated facility data accuracy at hot/cold flow conditions similar to the mixer tests.

Test results include static thrust coefficients, nozzle discharge coefficients, and effective throat areas.



# **FLUIDDYNE ENGINEERING CORPORATION**

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# **FLUIDYNE ENGINEERING CORPORATION**

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10a-c	Nozzle Discharge Coefficients



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### DEFINITION OF SYMBOLS

A	Cross-section area, in. <sup>2</sup>
c	Real-gas A/A* correction factor, dimensionless
C	Balance readout, counts
C <sub>D</sub>	Discharge coefficient, dimensionless
C <sub>T</sub>	Static thrust coefficient, dimensionless
D	Diameter
F	Steam thrust, lb
g	Acceleration of gravity, 32.174 ft/sec <sup>2</sup>
G	Real-fas stream thrust correction factor, dimensionless
H	Thrust component, lb
H <sub>2</sub>	Axial balance force, lb
K	Critical weight flow parameter, °R <sup>1/2</sup> /sec
K	Balance force calibration factor, lb/count
L	Calibration load, lb
M	Mach number, dimensionless
m	Mass flow rate, slugs/sec
P	Pressure, static unless otherwise specified by subscript, psia
ΔP	Static pressure difference across seal, psi
r	Radial distance, in
R <sub>N</sub>	Reynolds number, dimensionless
T	Temperature, °R
v	Velocity, ft/sec
V	Vertical balance force, lb
W	Weight flow rate, lb/sec
y	Distance from wall
δ	Boundary layer thickness
γ	Ratio of specific heats, dimensionless
Δ	Incremental quantity
λ	Pressure ratio, P <sub>t</sub> /P <sub>a</sub> , dimensionless
θ	Meridian angle measured clockwise looking upstream, degrees
ρ	Density, slugs/ft <sup>3</sup>



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### Subscripts

a	Ambient
e	Exit
i	Ideal
r	Resultant
t	Total conditions
w	Wall
x	Axial
y	Vertical
$\infty$	Freestream
1,2,4,5,7,8,9	See Figure 6

### Superscripts

*	Sonic condition
---	-----------------



## **FLUIDYNE ENGINEERING CORPORATION**

### **1.0 INTRODUCTION**

The present test program was conducted to provide mixer nozzle performance data for a JT8D turbofan engine. The test model was a 1/7-scale simulation of the exhaust system flowpath, including two mixer nozzles.

The mixer nozzles were furnished by Pratt & Whitney. Additional model components were designed and fabricated by FluidDyne using contours and instrumentation locations specified by P&W. The models were attached to existing model-to-facility adapters. The tests were conducted in FluidDyne's two-temperature-flow static thrust stand (Channel 11). Technical liaison for Pratt and Whitney was performed by Mr. Jerrold Blatt and Mr. Maurice Bridge.

This report describes the test facility, test models, data acquisition and analysis procedures, and presents the test results. Test conditions and major test results are tabulated in Figure 7 and are plotted in Figures 8-10.



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### **2.0 FACILITY DESCRIPTION**

The tests were performed in Channel 11 at Fluidyne's Medicine Lake Aerodynamic Laboratory. Channel 11 is a two-temperature-flow static thrust stand used to determine performance of exhaust nozzles in which the two exhaust flows are at different temperatures. Nozzle thrust is determined from force measurement with a strain gage force balance. The general arrangement of Channel 11 is shown in Figure 1. Photographs of test model installations are presented in Figure 5a.

The airflows for both the cold and hot passages of a test nozzle are obtained from the facility high-pressure dry air storage system. Air for the cold passage is throttled, metered through a long-radius ASME nozzle, ducted to the cold passage of the test nozzle, and finally exhausted to atmosphere. Air for the hot passage is throttled, passed through a regenerative storage heater, mixed with unheated bypass flow to achieve a desired temperature, metered through a long-radius ASME nozzle, ducted to the hot passage of the test nozzle, and finally exhausted to atmosphere.

The air heater used for the hot flow contains alumina pebbles which are preheated to approximately 1250°F with a combustion heater. The heater capacity is nominally 40 lbs/sec at 1200°F.

The model assembly is supported by a strain gage force balance and is isolated from the facility piping by two elastic seals; see schematic in Figure 6. Calibration of the balance and seals is described in Section 4.7.

The ASME meter at Station 1 is water-cooled to protect the elastic seal from thermal effects. Since the cooling water is confined to the upstream (i.e., non-metric) hardware only, no tare forces are introduced by the water supply lines.



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Facility instrumentation is provided to calculate mass flow rates at Stations 1 and 4, and to calculate the exit thrust produced by the test nozzle; details are described in Section 4.0. The data were recorded with Polaroid cameras and digital printers.



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### **3.0 MODEL DESCRIPTION**

The present tests used existing model adapters from the test programs reported in References 5, 6, 7, and 8, model components provided by P & W, and model components fabricated by FluidDyne for the present tests. Assembly drawings of the models are shown in Figure 2. Photographs of test hardware are shown in Figure 5.

#### **3.1 Model Adapters**

The test models attached to common adapting hardware which supplied separately-metered flows to the fan and core nozzles. The fan air flow was nominally at ambient temperature for all tests. The core air flow was nominally at ambient temperature for the "cold" tests, but was heated to approximately 600-800°F for the "hot" tests.

The main support member for the adapters (Spider, 0937-902) had been rebuilt for the Reference 7 tests. Adapters for the core passage consisted of an insulated duct, a jet breaker, a choke plate, two screens, and a common core shroud adapter which supported a common splitter/mixer adapter. Charging station instrumentation in the core passage included three 10-tube area-weighted total pressure ( $P_{t8}$ ) rakes, five area-weighted total temperature ( $T_{t8}$ ) probes, one thermocouple for controlling the flow temperature, three static pressure taps on the inner wall, and four static pressure taps on the outer wall.

The adapters for the annular fan passage included a choke plate, two screens, a common bellmouth contraction, and a common fan cowl adapter. Charging station instrumentation in the fan passage included four 12-tube, area-weighted,  $P_{t7}$  rakes; one 4-probe  $T_{t7}$  thermocouple rake, one thermocouple for control purposes; and four static taps on the inner and outer walls.



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The single control thermocouples at the two charging stations were used to set the desired temperature ratio,  $T_{t8}/T_{t7}$ . Outputs from these thermocouples were amplified, divided, and displayed on a digital panel meter to provide the facility valve operators with a visual indication of the actual temperature ratio.

### **3.2 Nozzle Components**

Nozzle components included a reference splitter (free mixer), two 12-lobed mixers, three plugs, two fan duct cases, and one tailpipe.

The mixers were of the same basic design; one had deep scallops (Configuration 4A) and another had scallops and shortened mixing lobes (Configuration 5A).

Three test configurations were assembled, as summarized in Figure 3. Photographs of the test hardware, including partial assemblies, are shown in Figures 5a-d.

The only static pressure taps on the nozzle components were added for the thrust reverser tests; details are shown in the drawings (Figure 4).

### **3.3 Standard ASME Nozzles**

Facility demonstration and checkout tests were performed using two standard ASME long-radius flow metering nozzles (Figures 2a and 5a). The upper nozzle (cold flow) simulated the fan passage of the mixer nozzle, while the lower nozzle (hot flow) simulated the core passage. The two ASME nozzles were tested separately and simultaneously (only simultaneous test results are presented in this report).



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### 4.0 DATA ANALYSIS PROCEDURES

The following subsections describe the data analysis procedures used in the present test program. Station notations are defined in Figure 6. Computer programs written in BASIC language are included in the Appendix.

#### 4.1 $P_{t7}$ and $P_{t8}$ Definitions

$P_{t7}$  was defined as the area-weighted average of four 12-probe rakes. The area-weighting method was adequate for the present tests since only non-distorted pressure profiles were involved.

For all configurations except the reference model (Configuration 2A),  $P_{t8}$  was defined as the area-weighted average of three 10-probe rakes. For the reference model tests,  $P_{t8}$  was defined as 1.0075 times the mass-flow-derived  $P_{t8}$ . This correlation was used because of disagreement between the three  $P_{t8}$  rakes. The correlation was obtained by comparing the mass-flow-derived  $P_{t8}$  to the area-weighted  $P_{t8}$  on the configuration 2A tests. After the configuration 2A tests were completed, a second screen was added to the core passage upstream of the core charging station, resulting in good agreement between the three  $P_{t8}$  rakes; the correlation was not used for subsequent tests.

The above paragraphs define  $P_{t7}$  and  $P_{t8}$  at the charging station rakes. These total pressures were then adjusted to a hypothetical charging station located 1.2 inches further downstream. The calculation procedure, specified by P&W, is detailed in the Appendix. The  $P_t$  corrections were on the order of 0.1%.



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### 4.2 Flow Rates

The mass flow rates through the test nozzles were determined using choked ASME long-radius metering nozzles. The core nozzle flow rate was calculated at Station 1, (see Figure 6) and the fan nozzle flow rate was calculated at Station 4, using the following equations.

$$W_1 = W_8 = \frac{K_1 C_{D1} A_1 P_{t1}}{\sqrt{T_{t1}}}$$

$$W_4 = W_7 = \frac{K_4 C_{D4} A_4 P_{t4}}{\sqrt{T_{t4}}}$$

The critical flow factor, K, was calculated as a function of total pressure and total temperature.

$$K = 0.52820 + aT_t + bT_t^2 + cT_t^3 + 0.186 \times 10^{-4} \times P_t \times e^{-.0067(T_t - 5000)}$$

where:

$$\begin{aligned} a &= 0.1654 \times 10^{-4} \\ b &= -0.2119 \times 10^{-7} \\ c &= 0.6008 \times 10^{-11} \end{aligned}$$

$T_t$  is in °R and  $P_t$  is in psia.



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This equation was obtained by curve-fitting tabulated values in Reference 1; the curve-fit is accurate to within  $\pm 0.03\%$  for  $0 < P_t < 30$  atmospheres and  $460 < T_t < 700^\circ\text{R}$ , and is accurate to within  $\pm 0.1\%$  for  $0 < P_t < 40$  atmospheres and  $460 < T_t < 1800^\circ\text{R}$ .

$C_{D_4}$  was calculated using a semi-empirical equation

$$C_{D_4} = 1 - 0.184 R_{N_4}^{-0.2}$$

and varied from 0.991 to 0.993 for the present tests.

$C_{D_1}$  was calculated from a similar equation, modified to account for a thermal boundary layer. This thermal boundary layer results from water-cooling of the Station 1 meter.

$$C_{D_1} = 1 - (0.184 R_{N_1}^{-0.2}) (1.574 - 0.574 T_{t_1}/T_w)$$

The above equation was derived assuming constant static pressure in the boundary layer, a  $1/7$  power velocity profile, thermal boundary layer thickness equal to velocity boundary layer thickness, and a density distribution in the boundary layer defined by

$$\frac{\rho}{\rho_\infty} = \frac{T_\infty}{T_w} - \left( \frac{T_\infty}{T_w} - 1 \right) \left( \frac{y}{\delta} \right)^{1/7}$$

$T_w$ , the wall temperature at the nozzle throat, was estimated from heat-balance calculations of heat transfer from the air stream to the cooling water.  $T_w$  values calculated for the present tests varied from  $105^\circ$  to  $165^\circ\text{F}$ .  $R_{N_1}$  was calculated using a mean temperature  $(T_w + T_{t_1})/2$ .  $C_{D_1}$ , calculated using the above equation, varied from 0.991 to 0.997 for the present tests. Given sufficient wall-cooling,  $C_{D_1}$  may exceed unity (Reference 2).

The above equation for  $C_{D_1}$  is believed to be correct within  $\pm 0.002$ , on the basis of results from facility demonstration



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tests. These demonstration tests included test series with either a 2.5-inch or a 4-inch diameter ASME nozzle located downstream of the water-cooled Station 1 meter. The downstream nozzle was essentially at adiabatic conditions (thin-wall construction, backside insulated). Flow rates calculated at Station 1 (using the above  $C_{D1}$  equation) agreed within  $\pm 0.25\%$  with flow rates calculated at the downstream nozzle (using adiabatic wall  $C_D$ ), thereby indicating the adequacy of the  $C_{D1}$  equation.

$A_4$ , the geometric throat area of the Station 4 meter, was 2.5475 in<sup>2</sup>.  $A_1$ , the geometric throat area of the Station 1 meter, was calculated assuming thermal expansion from 70°F to  $T_w$ . The largest value of  $A_1$  calculated for the present tests was 1.3490 in<sup>2</sup>, representing a thermal expansion (area change) of 0.16% from the nominal area of 1.3468 in<sup>2</sup>.

$P_{t1}$  and  $P_{t4}$  were measured on Heise bourdon-tube gages.  $T_{t1}$  and  $T_{t4}$  were measured with shielded chromel/alumel thermocouples and recorded on the facility Vidar system (analog to digital converter, printer).

Calculated flow rates (lbm/sec) for the present tests were in the ranges

$$2.6 < W_1 < 10.2$$

$$2.2 < W_4 < 9.1$$

### 4.3 Discharge Coefficients and Effective Throat Areas

Discharge coefficient is defined as the ratio of actual flow rate through a nozzle to the ideal isentropic flow rate at the overall nozzle pressure ratio. Overall nozzle pressure ratios are defined as  $\lambda_7 = P_{t7}/P_a$  and  $\lambda_8 = P_{t8}/P_a$ .

$$C_{D7} = \frac{W_4 \sqrt{T_{t7}}}{K_7 A_7 P_{t7} (A^*/A)_7} \quad \text{and} \quad C_{D8} = \frac{W_1 \sqrt{T_{t8}}}{K_8 A_8 P_{t8} (A^*/A)_8}$$



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$K_7$  and  $K_8$  were evaluated using a previous equation, as functions of  $P_{t7}$ ,  $T_{t7}$  and  $P_{t8}$ ,  $T_{t8}$ .

The throat area of the 3.75-inch ASME nozzle was  $A_7 = 11.0447 \text{ in}^2$ . The throat area of the 4.00-inch ASME nozzle, when measured at room temperature, was  $A_8 = 12.5664 \text{ in}^2$ . The actual throat area of the 4.00-inch nozzle, when tested with hot flow, was calculated assuming thermal expansion from  $70^\circ\text{F}$  to a recovery temperature,  $T_w$ .  $T_w$  was calculated assuming isentropic expansion and a recovery factor of 0.89, i.e.,

$$T_w/T_{t8} = (T_8/T_{t8}) + 0.89 (1 - T_8/T_{t8})$$

$A_8$  values calculated for the ASME tests ranged from 12.710 to 12.714  $\text{in}^2$ . For the mixer nozzle tests, reference areas  $A_7$  and  $A_8$  were not defined, and therefore, fan and core nozzle discharge coefficients could not be calculated. However, the effective throat areas ( $C_{D7}A_7 + C_{D8}A_8$ ) were calculated. In addition, an overall nozzle discharge coefficient was calculated as  $C_{D9} = (C_{D7}A_7 + C_{D8}A_8)/A_9$  where  $A_9$  is the exit area of the tailpipe (see Figure 6a) and equals 14.21  $\text{in}^2$ .

$P_{t7}$  and  $P_{t8}$  were measured on multiple-tube mercury manometers, and were defined as described in Section 4.1.

$T_{t7}$  and  $T_{t8}$  were measured with shielded chromel/alumel thermocouples. However, since the thermocouple rakes were further upstream than the pressure rakes, the measured temperatures were adjusted to account for heat transfer (between the air and the model walls) occurring between the temperature and pressure rake stations. This adjustment was based on analytic estimates which indicated that  $T_{t8}$  should decrease  $5.7^\circ$  when the core and fan temperatures differed by  $760^\circ\text{F}$ . An energy balance required that the fan flow temperature would increase by approximately the same amount. The temperature



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adjustments calculated for the present tests ranged up to  $-6^\circ$  for  $T_{t_8}$  and  $+6^\circ$  for  $T_{t_7}$ .

$A^*/A$ , the isentropic area ratio, is used to correct the ideal flow rate when the nozzle is unchoked.  $A^*/A$  for the fan nozzle was calculated using equations valid for  $\gamma = 1.4$ , obtained from Reference 3.

$$A^*/A = 3.86393\lambda^{-0.71429} \sqrt{1 - \lambda^{-0.28571}} \text{ for } \lambda \leq 1.8929$$

and

$$A^*/A = 1 \text{ for } \lambda \geq 1.8929$$

$A^*/A$  for the core nozzle was obtained by correcting the  $\gamma = 1.4$  value for "real gas effects," to account for  $\gamma_8$  being significantly less than 1.4. The correction was derived by curve-fitting tabulated values from Reference 4; no corrections were indicated for  $T_t < 900^\circ\text{R}$ . First, the critical pressure ratio was expressed as a function of total temperature:

$$1/\lambda^* = 9.667 \times 10^{-6} \times T_t (\text{°R}) + 0.5196$$

If  $\lambda \geq \lambda^*$ , then  $A^*/A = 1$ . If  $\lambda < \lambda^*$  and  $900 < T_t < 1260^\circ\text{R}$ ,

$$c = 1 + \left(\frac{1}{\lambda} - \frac{1}{\lambda^*}\right) 5.728 \times 10^{-5} (T_t - 900)$$

If  $\lambda < \lambda^*$  and  $1260 < T_t < 1800^\circ\text{R}$ ,

$$c = 1 + \left(\frac{1}{\lambda} - \frac{1}{\lambda^*}\right) [2.615 \times 10^{-5} (T_t - 1260) + 0.020621]$$

Finally,

$$A^*/A = c \times (A^*/A)_{\gamma = 1.4}$$

For the present tests,  $c$  (denoted  $c^*$  on computer output sheets) varied from 1.000 to 1.002.



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### 4.4 Thrust Measurement

The net static axial thrust of an exhaust nozzle is defined as the axial exit momentum of the exhaust flow, plus the excess of exit pressure over ambient pressure times the exit area.

$$H_x = mv_{e_x} + (P_e - P_a) A_{e_x}.$$

The net static thrust of an exhaust nozzle model was determined in the present test program by applying the momentum equation to the control volume shown in Figure 6. The analysis of axial forces applied to the control volume includes entering stream thrusts ( $F_1$  and  $F_4$ ), a balance force ( $H_2$ ), various pressure-area terms and the axial exit stream thrust, ( $H_x + p_a A_{e_x}$ ). The axial balance force,  $H_2$ , as used here, included seal tare forces. Summing axial forces,

$$H_x = F_1 + F_4 + p_2 (A_2 - A_1) + p_5 (A_5 - A_4) - p_a (A_2 + A_5) - H_2$$

The stream thrust at Station 4 is the exit stream thrust of a choked long-radius ASME nozzle, and was calculated as:

$$F_4 = G_4 (1 + 1.4 C_{D_4} C_{T_4}) .52828 P_{t_4} A_4$$

Use of  $\gamma = 1.4$  and  $P^*/P_t = .52828$  in the above equation imply an ideal gas. The factor  $G$ , derived from tabulated values in References 1 and 4, corrects the stream thrust from that of an ideal gas to that of a real gas.

$$\text{If } T_t < 560^\circ\text{R}, G = 1.00012 + 6.8338 \times 10^{-6} \times P_t \text{ (psia)}$$

$$\text{If } T_t > 560^\circ\text{R}, G = 1.0044 - (4.196 - .0059 P_t) (T_t + 460) \times 10^{-6}$$

$C_{D_4}$  has already been discussed;  $C_{T_4}$  was calculated in an analogous manner,



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$$C_{T_4} = 1 - 0.109 R_{N_4}^{-0.2}$$

This equation is a semi-empirical expression of the thrust coefficient of an ASME nozzle at a pressure ratio of  $\lambda = 1.8929$  (corresponding to  $P^*/P_t = .52828$ ). For the present tests,  $G_4$  varied from 1.000 to 1.0013, and  $C_{T_4}$  varied from 0.995 to 0.997.

The stream thrust at Station 1 was calculated as:

$$F_1 = G_1 (1 + 1.4 C_{D_1} C_{T_1}) .52828 P_{t_1} A_1$$

Each variable in this equation has been previously described, except  $C_{T_1}$ .  $C_{T_1}$  was calculated in a similar manner as  $C_{T_4}$ , but was modified to account for the thermal boundary layer described in the discussion of  $C_{D_1}$  in Section 4.1:

$$C_{T_1} = 1 - (0.109 R_{N_1}^{-0.2}) (0.828 + 0.172 T_{t_1} T_w)$$

The above equation was derived using the same assumptions as in the derivation of  $C_{D_1}$ .  $C_{T_1}$  for the present tests varied from 0.992 to 0.996.

Static pressures  $p_2$  and  $p_5$  were measured on mercury manometer. Ambient pressure ( $p_a$ ) was measured on a Haas mercury manometer (barometer).  $A_5$  and  $A_2$ , the geometric reference areas for the seals, were both  $3.5 \text{ in}^2$ .

The vertical thrust,  $H_y$ , was calculated from the two vertical balance forces:

$$H_y = V_1 + V_3$$

The resultant thrust,  $H_r$  was calculated as the vector sum of the axial thrust  $H_x$ , and vertical thrust  $H_y$ . Resultant thrust vector angle,  $\alpha$ , was determined as:



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$$\alpha = \tan^{-1} \frac{H_y}{H_x}$$

The sign convention for positive values of thrust components and vector angle is defined in Figure 6a. Note that for the present tests, the nozzle is mounted horizontally and there is no vertical thrust.

### 4.5 Thrust Coefficient

The static thrust coefficient of an exhaust nozzle is defined as the ratio of the measured nozzle net thrust to the ideal thrust of the actual mass flow when expanded isentropically from  $P_t$  to  $P_a$ .

$$C_T = \frac{H}{m v_i}$$

For the dual-flow tests, the ideal thrust was calculated as the sum of the fan nozzle ideal thrust and the core nozzle ideal thrust:

$$C_T = \frac{H}{m_7 v_{i7} + m_8 v_{i8}}$$

For the present tests, thrust coefficients were calculated for the axial and vertical thrust components and for the resultant vector.

$$C_{T_x} = \frac{H_x}{m_7 v_{i7} + m_8 v_{i8}}$$

$$C_{T_y} = \frac{H_y}{m_7 v_{i7} + m_8 v_{i8}}$$

$$C_{T_r} = \sqrt{C_{T_x}^2 + C_{T_y}^2}$$



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Ideal thrust was calculated using a dimensionless ideal thrust function,  $m_i v_i / P_t A^*$ , which is a function of both  $\lambda$  and  $\gamma$ .

$$m_i v_{i7} = (A^*/A)_7 C_{D7} A_7^2 P_{t7} (m_i v_i / P_t A^*)_7$$

$$m_i v_{i8} = (A^*/A)_8 C_{D8} A_8^2 P_{t8} (m_i v_i / P_t A^*)_8$$

where

$$\begin{aligned} (m_i v_i / P_t A^*) &= \gamma \left[ \frac{2}{\gamma + 1} \right]^{\frac{\gamma}{\gamma - 1}} \sqrt{\frac{\gamma + 1}{\gamma - 1}} \sqrt{1 - \gamma^{\frac{1 - \gamma}{\gamma}}} \\ &= 1.81162 \sqrt{1 - \lambda^{-0.28571}}, \text{ for } \gamma = 1.4. \end{aligned}$$

For the present tests,  $\gamma_7$  was taken to be 1.400. However,  $\gamma_8 \neq 1.4$  and, therefore,  $(m_i v_i / P_t A^*)_8$  obtained from the above equation was corrected to account for "real gas effects" by multiplying by the ratio

$$\frac{(m_i v_i / P_t A^*) \text{ for real gas}}{(m_i v_i / P_t A^*) \text{ for } \gamma = 1.4}$$

This ratio was calculated from tabulated values in Reference 4; for the present range of test conditions this factor was obtained from a curve-fit expression

$$.9957 - 5.81 \times 10^{-5} \times (T_{t8}, ^\circ R - 1000) + 1.25 \times 10^{-3} \times (\lambda_8 - 1)$$

and varied between .9950 and .9972.



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### **4.6 Pressure and Temperature Data**

Pressure instrumentation for facility pressures and charging station pressures were described previously. All other pressures in the model were measured using multiple-tube mercury manometers. Model pressure data were reduced to absolute pressures (psia) and dimensionless ratios ( $P/P_{t7}$ ,  $P/P_{t8}$ ,  $P/P_a$ ). The results are tabulated on computer output sheets, contained in the Appendix.

Facility and charging station temperature data were obtained using shielded chromel/alumel thermocouples, and were recorded on the facility Vidar system. Temperatures were expressed in °F or °R, or both.

### **4.7 Force Balance Calibration**

The force balance calibration determines the output characteristics of both the force balance flexures and the elastic seals which provide pressure-tight expansion joints between the metric model assembly and the non-metric facility structure. The outputs of the strain-gage flexures are very linear with applied load, but the seals provide an additional axial force which is a function of both axial load and seal pressure. Most of this force carryover results from radial seal deflections required to support the static pressure differentials across the seals when the ducts are pressurized. Consequently, the seal and balance assembly is calibrated under simulated operating conditions of loads and seal differential pressures. The calibration for this mixed flow facility is further complicated by the fact that the location of the applied load during a test is a function of the hot/cold flow split and nozzle pressure ratios; the calibration must, therefore, duplicate both the magnitude and location of the net force which was experienced during a test. As a result of these requirements, it has been found expedient to calibrate "on-point," that is, to determine the balance



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output characteristics while simultaneously reproducing the forces, force location, and seal pressures experienced at a specific test point.

The forces and force location for each test point are not known exactly until the on-point calibration is completed. The initial test data are, therefore, reduced (by computer) using a preliminary calibration. The computer is programmed so that, as it reduces the initial test data, it also prints out the required calibration information (calibration load and load location), such that an accurate on-point calibration can then be made.

The on-point calibration is made with the seals pressurized to the pressures measured during the test, and the loads applied at the locations obtained from the preliminary data. The loads are then varied slightly to obtain approximately the balance output measured during the test. The on-point calibration factors are then calculated as:

$$K_2 = \left[ (L_x + \Delta P_2 A_2 + \Delta P_5 A_5) / C_2 \right]$$

$$K_1 = V_1 / C_1$$

$$K_3 = V_3 / C_3$$

The final balance forces for reducing test data were then calculated as

$$H_2 = K_2 C_2, \quad V_1 = K_1 C_1, \quad V_3 = K_3 C_3,$$

using the actual balance outputs measured during the test.



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### 5.0 PRESENTATION OF RESULTS

Test conditions and major test results are tabulated in Figure 7 and are plotted in Figures 8 through 10. Detailed data and calculations are contained in a separate Data Appendix.

The tabulation in Figure 7 includes: configuration, run number, actual values of the independent test variables ( $\lambda_7, \lambda_8, T_{t8}/T_{t7}$ ), the pressure ratio split ( $\lambda_7/\lambda_8$ ), and the major test results ( $C_T, C_{D7}, C_{D8}, C_{D9}, W_7/W_8$ ). For the ASME nozzle tests,  $C_{D7}$  and  $C_{D8}$  are tabulated in place of effective throat areas.

#### 5.1 ASME Checkout Tests

Standard ASME nozzles (Figures 2a and 5a) were tested to demonstrate facility data accuracy in determining  $C_D$  and  $C_T$  of test nozzles. Results of these tests are tabulated in Figure 7 (sheet 1) and are plotted in Figure 8.

Target performance curves for the ASME nozzles are shown in the figures. These predictions are based on semi-empirical equations, and were obtained by analysis of ASME nozzle exit surveys conducted by Fluidyne.

The test results were statistically analyzed in terms of bias (average difference between actual and predicted values) and scatter (standard deviation of the individual differences from their average). This analysis is summarized in the following table:

Runs	Bias:			Standard Deviation:		
	$C_{D7}$	$C_{D8}$	$C_T$	$C_{D7}$	$C_{D8}$	$C_T$
1 - 5	.0002	-.0010	-.0006	.0009	.0012	.0006
6 - 10	.0003	-.0008	.0004	.0014	.0018	.0017



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### **5.2 Mixer Nozzle Tests**

Major results from the mixer nozzle performance tests are tabulated in Figure 7, Sheets 2 and 3. Thrust Coefficients are plotted versus core nozzle pressure ratio in Figures 9a - 9c. Justification of high  $C_T$  values ( $C_T > 1.00$ ) for mixer nozzles was discussed in Reference 5; similar reasoning applies for the present test results.

Overall nozzle discharge coefficients,  $C_{Dg}$ , are plotted in Figures 10a - 10c. The abscissa is the mass-flow-weighted pressure ratio,  $\lambda = (W_7\lambda_7 + W_8\lambda_8)/(W_7 + W_8)$ . The ordering of the figures is the same as for the thrust coefficients in Figure 9.

In addition to the specified test program, Configuration 2A was tested at three other times (twice near middle and near the end of program). These additional tests demonstrated facility data repeatability.



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8. Kirschbaum, R. A., and Brasket, R. G., "Hot/Cold Flow Model Tests to Determine Static Performance and Jet-Wing Surface Temperatures for 1/7-Scale JT8D-217 Mixer Exhaust Nozzles." FluidDyne Report 1067, Pratt & Whitney Agreement of 30 December 1975. March 1976.



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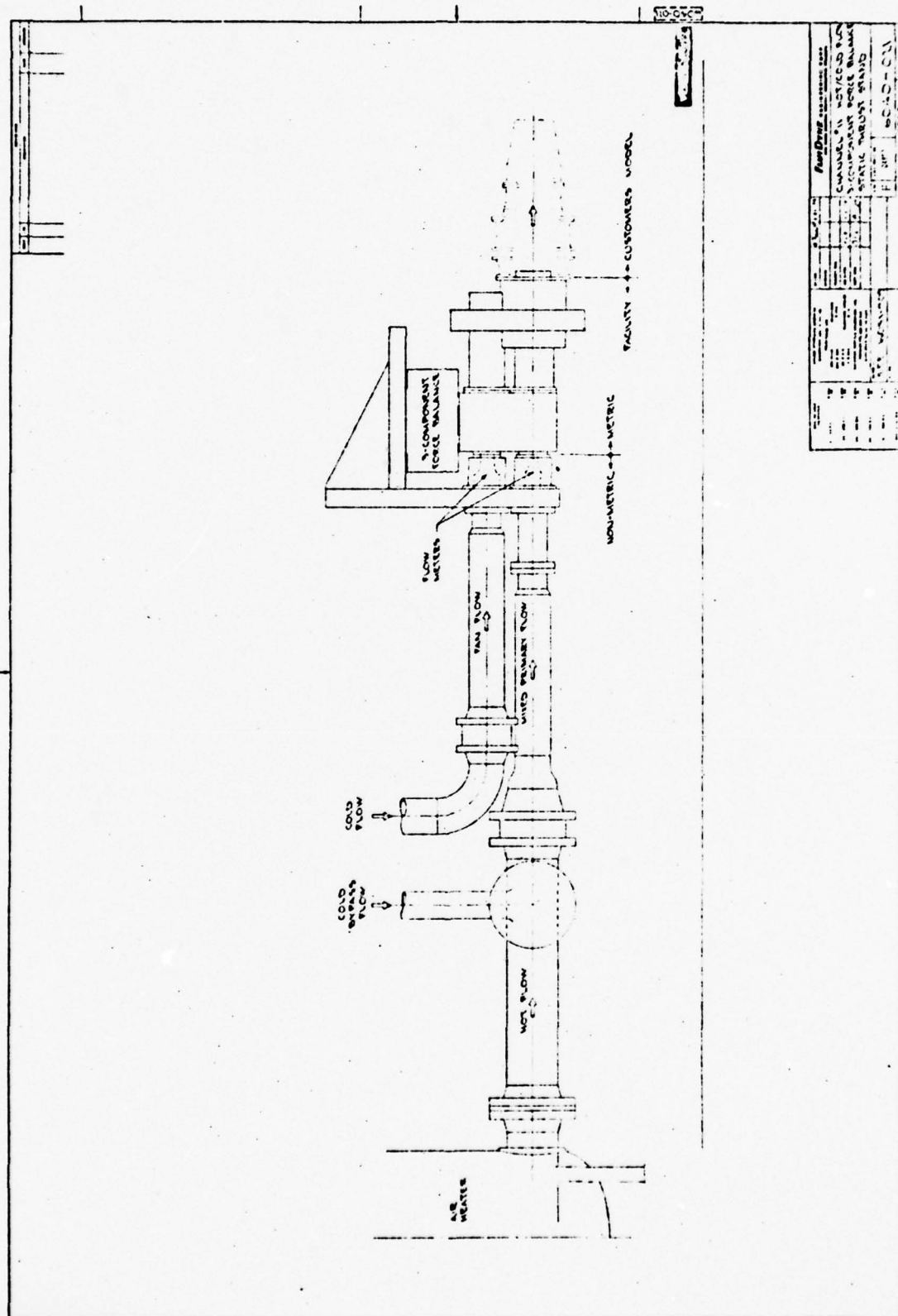


FIGURE 1. CHANNEL 11 FACILITY LAYOUT



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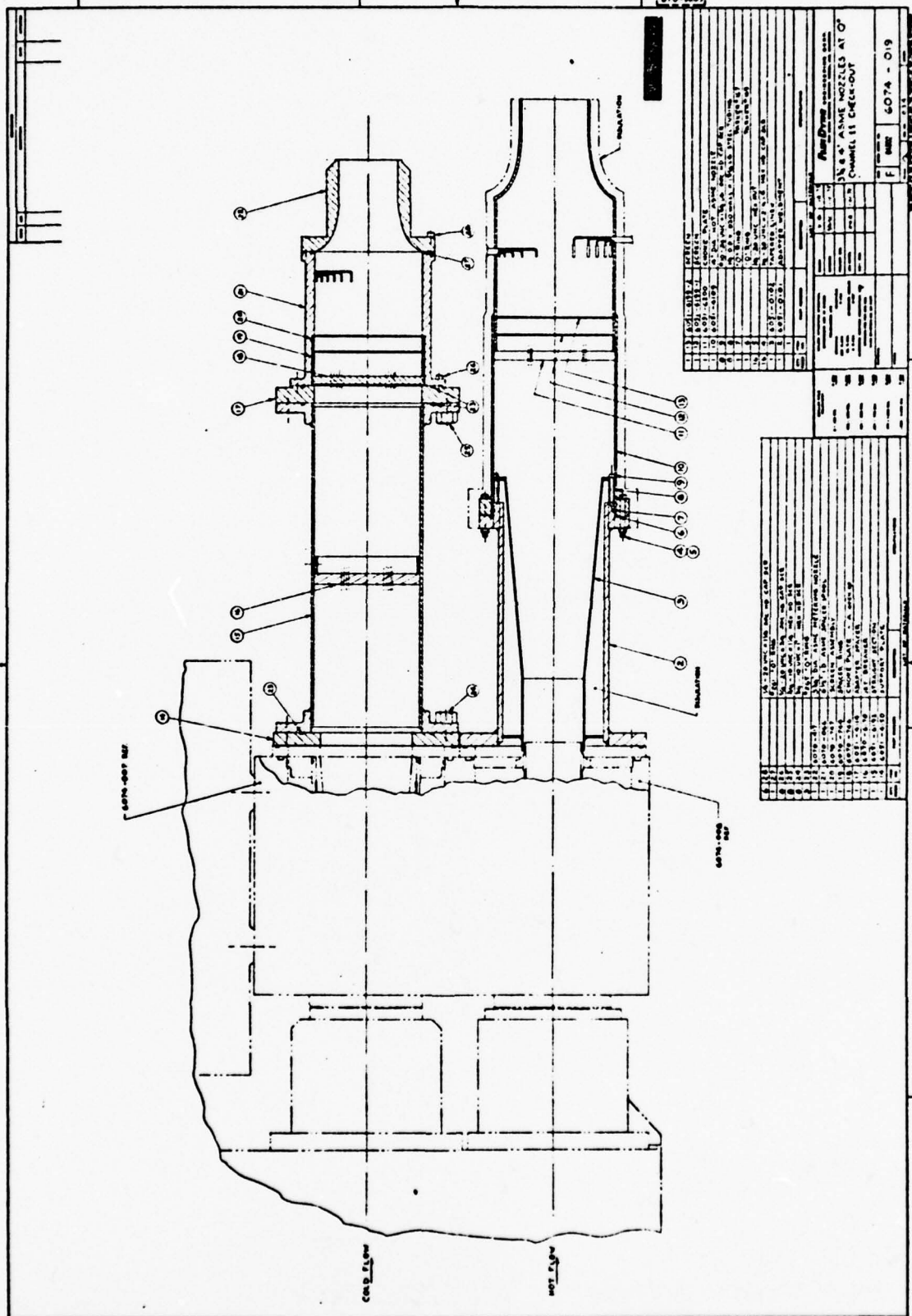
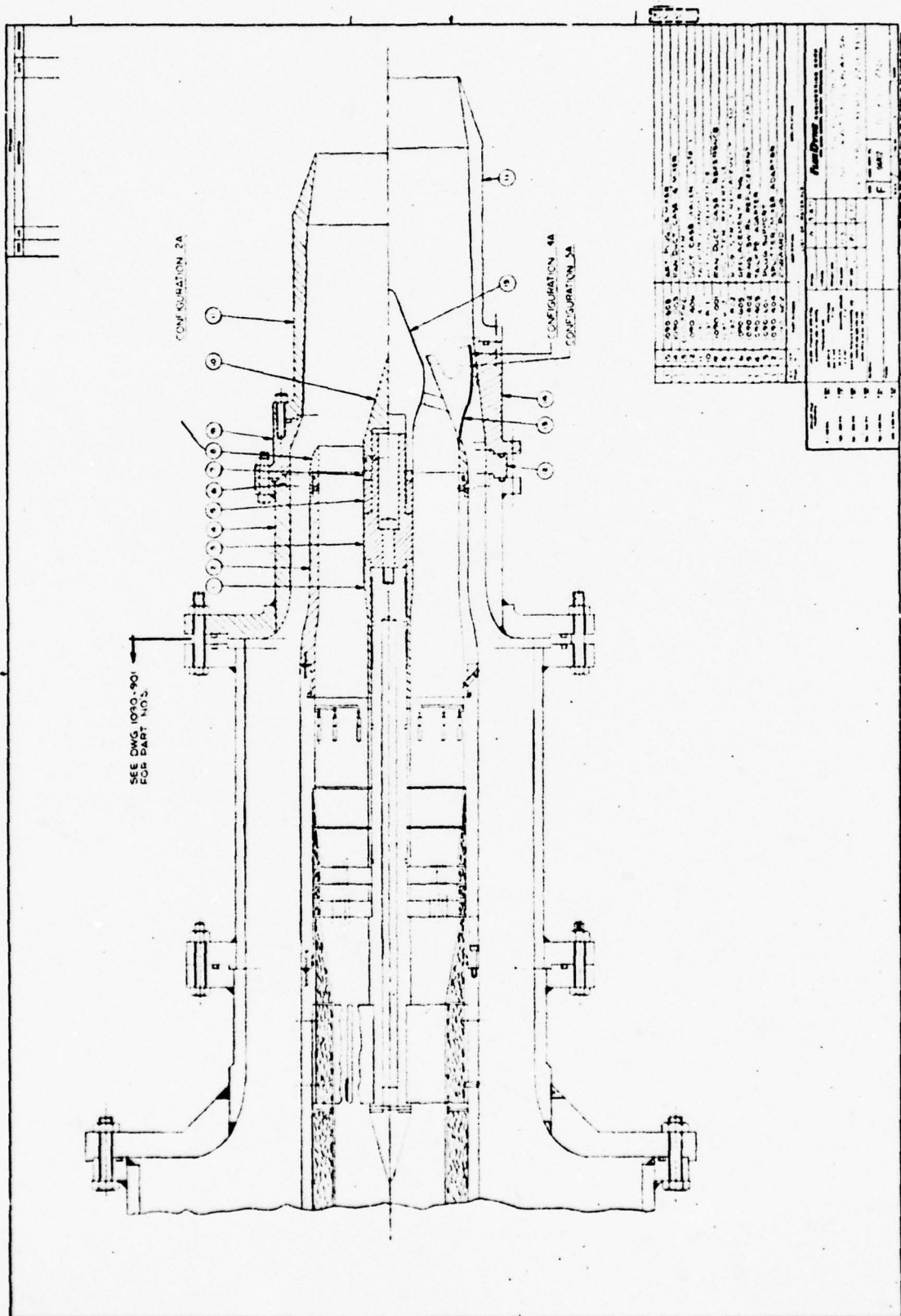


FIGURE 2a. ASME NOZZLE TEST ASSEMBLY DRAWING



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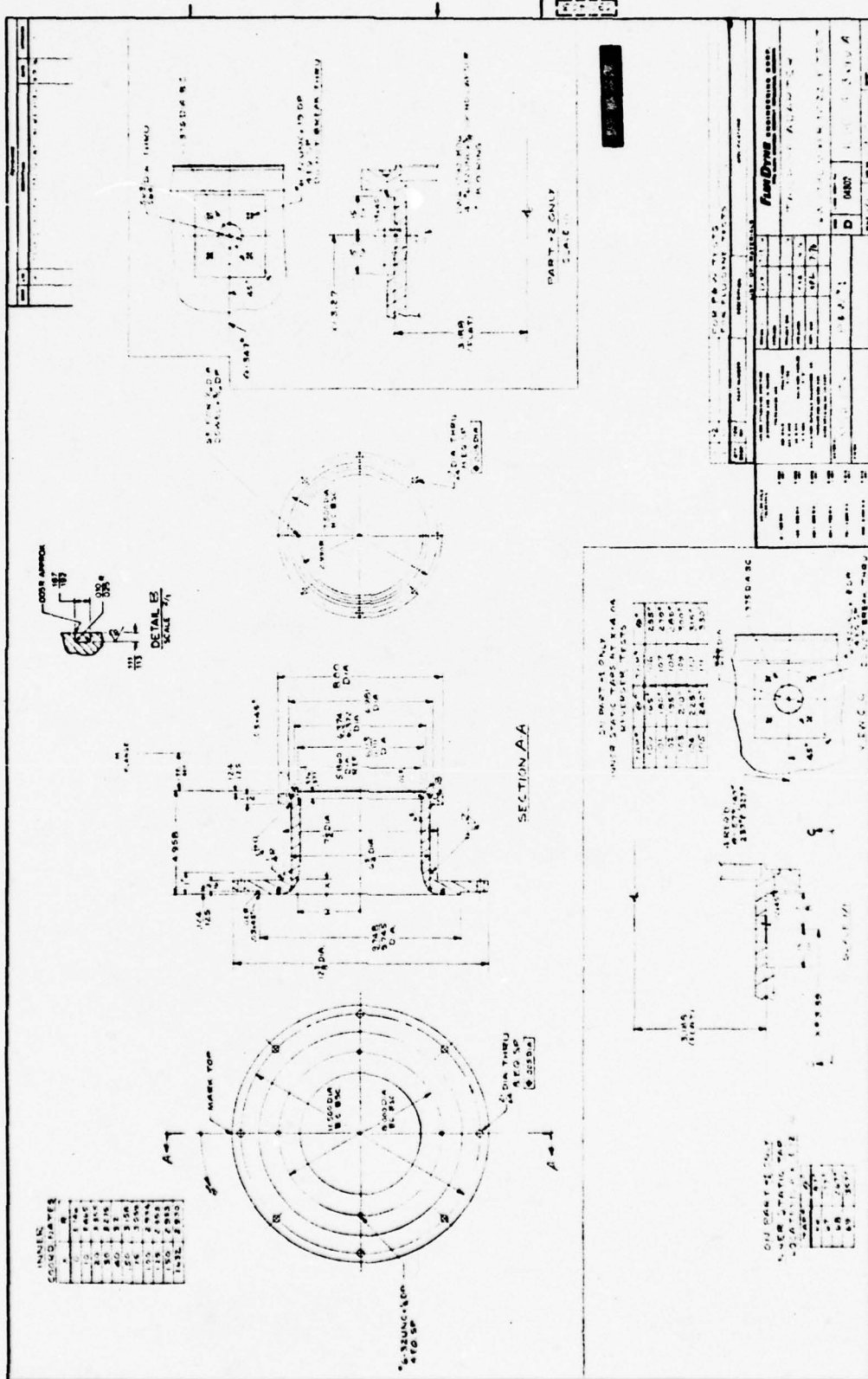


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Run Numbers	Config.	Splitter or Mixer	Tailpipe	Fan Duct Case	Swirl, TEGV, Tangential Struts, Fan P <sub>t</sub> Distortion	Other
1-10	Dual ASME					
11-38, 214	2A	Reference Splitter	Common (401)	Reference (001)	No	
114-123	5A	16" Mixer, Cutback, Scalloped	Common (401)	16" Mixer (003)	No	1090-406 duct case spacer
124-153	4A	16" Mixer, Scalloped	Common (401)	16" Mixer (003)	No	1090-406 duct case spacer

FIGURE 3. SUMMARY OF TEST CONFIGURATIONS



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**FIGURE 4a. TAILPIPE ADAPTER**



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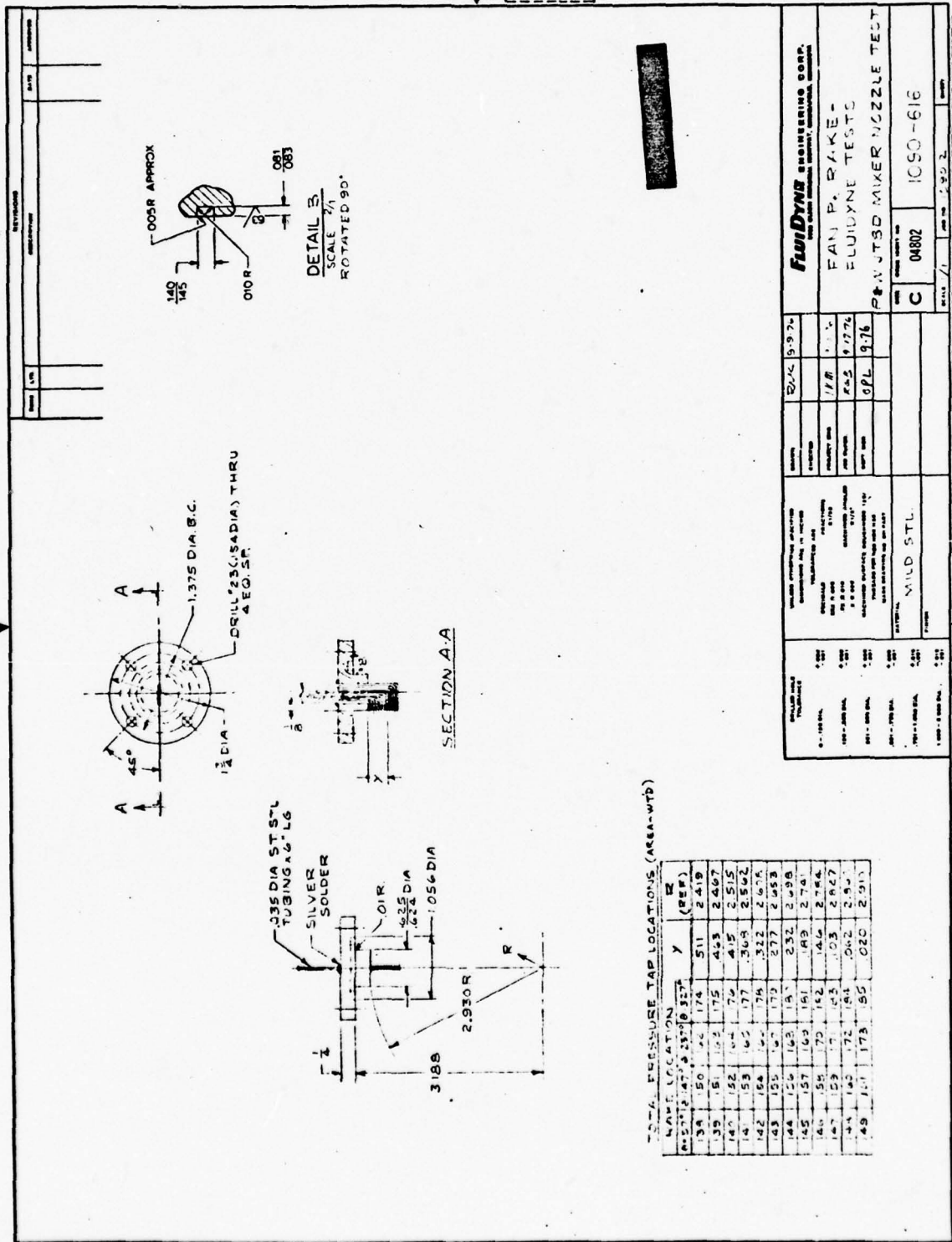


FIGURE 4b. FAN Pt RAKE



[illegible]

**FIGURE 4c. FAN DUCT CASE-REFERENCE**



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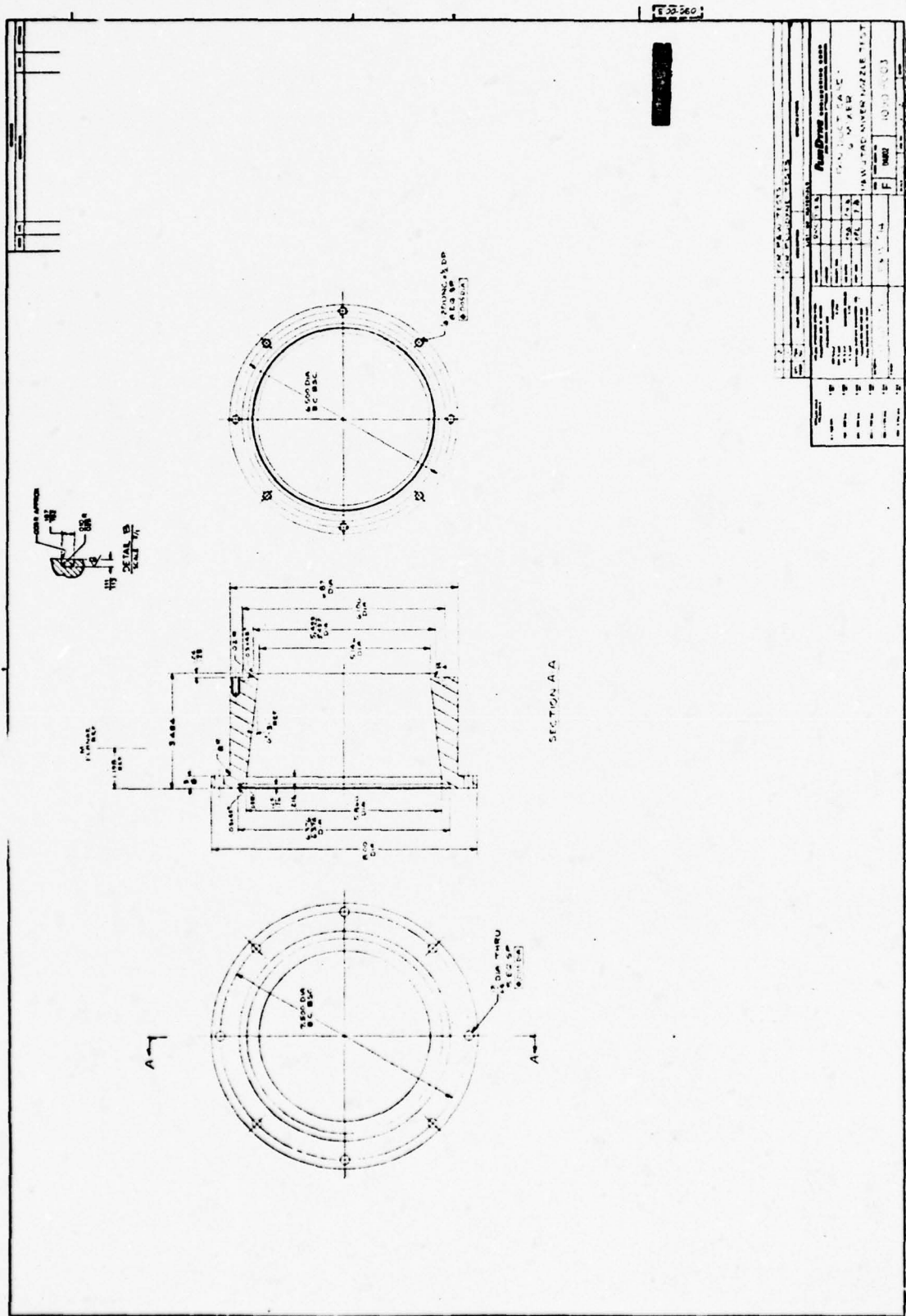


FIGURE 4d. FAN DUCT CASE - 16" MIXER



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FIGURE 4e. TAILPIPE - COMMON



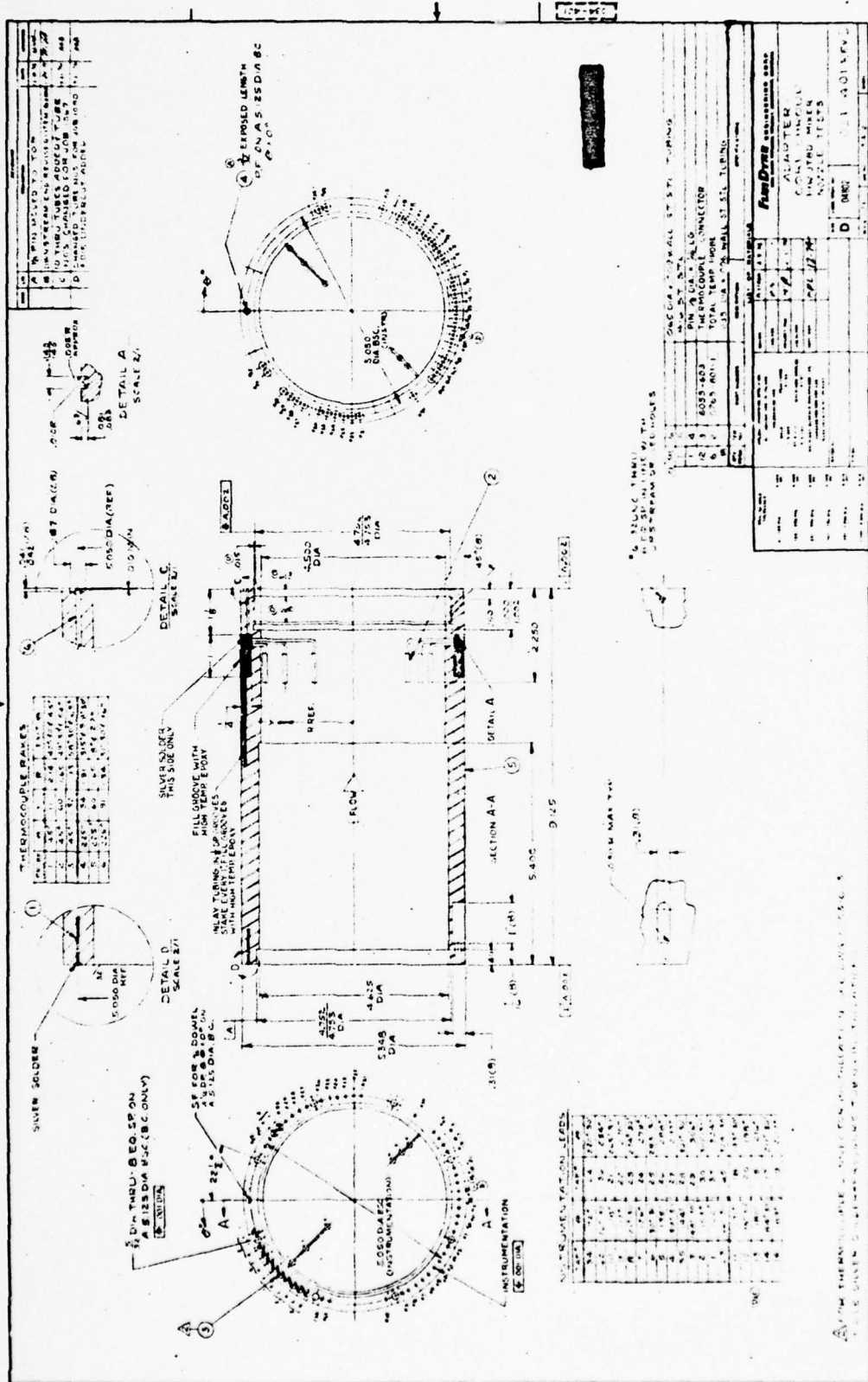
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FIGURE 4f. CORE SHROUD ADAPTER



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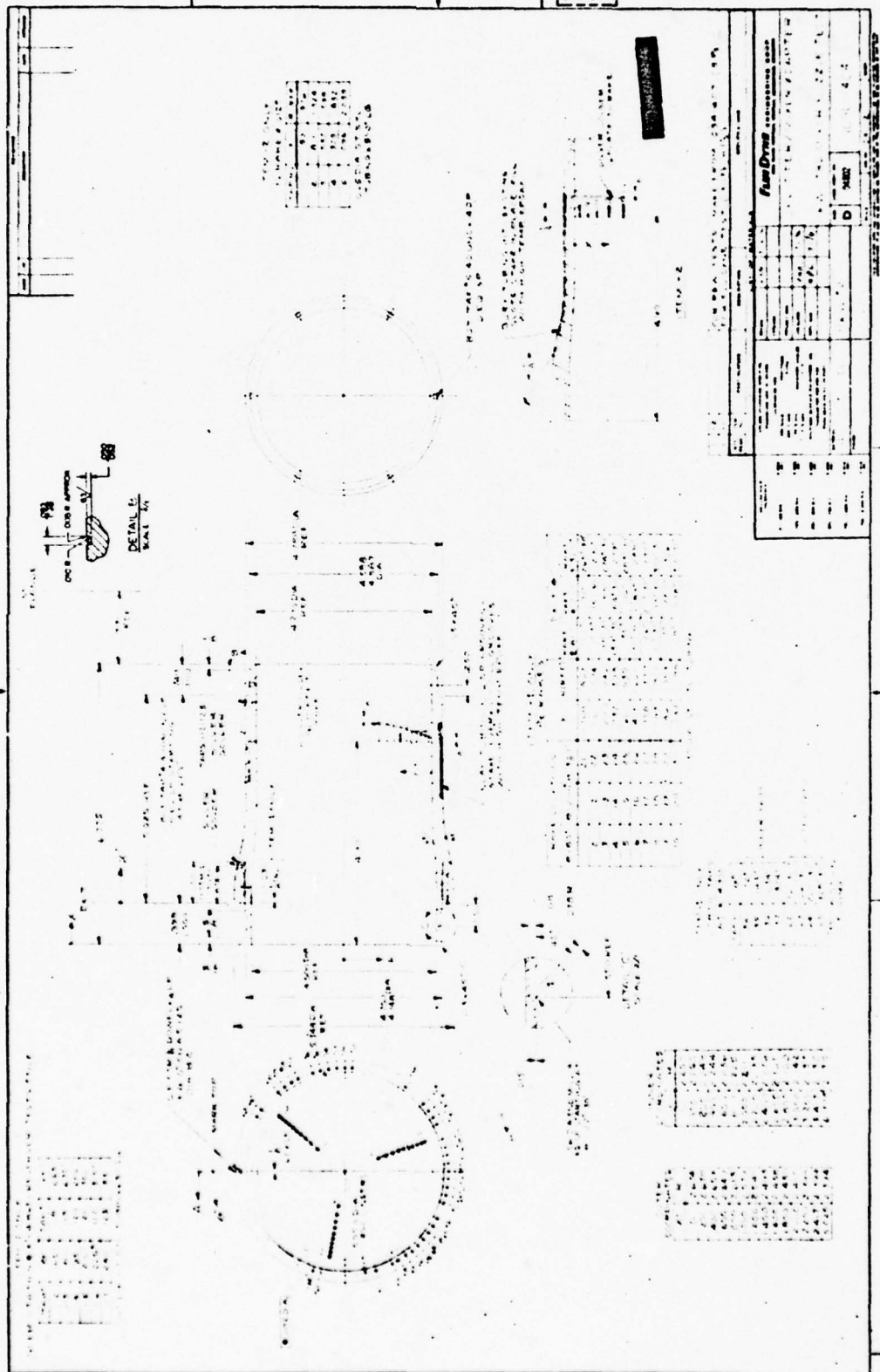


FIGURE 49. SPLITTER/MIXER ADAPTER

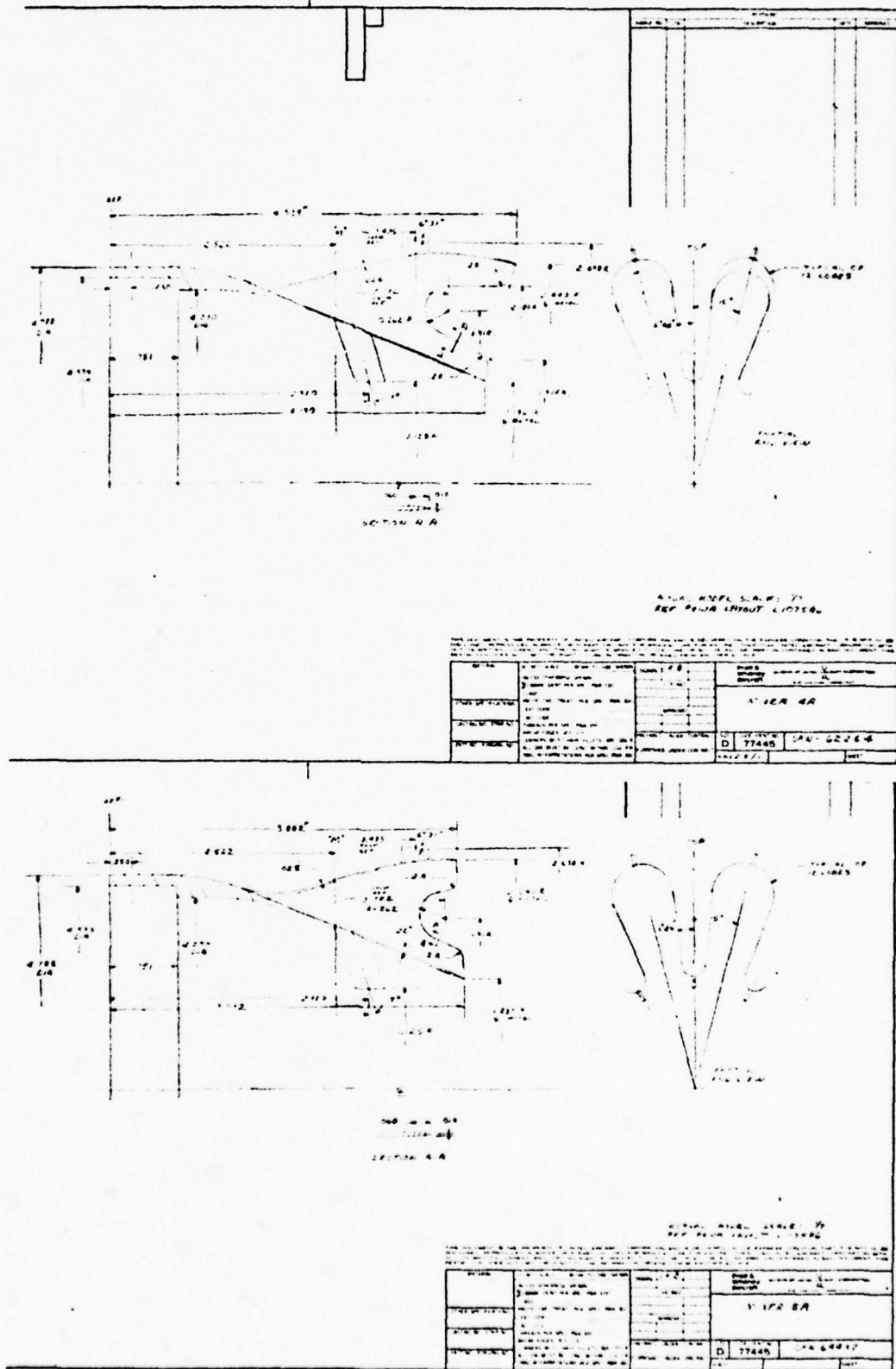


[illegible]

#### FIGURE 4h. SPLITTER - REFERENCE



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**FIGURE 41. MIXER - CONFIGURATION 4A and 5A**



Technical drawing of a P&W JT8D mixer nozzle test plug. The drawing includes a side view and a top view. The side view shows a conical nozzle with a 1/4 inch diameter at the base, a 1.001 inch diameter at the top, and a 1.002 inch diameter at the base of the cone. The cone has a 1.078 inch diameter at the top and a 1.078 inch diameter at the base. The side view also shows a 1.598 inch reference dimension, a 1.078 inch reference dimension, and a 1.078 inch reference dimension. The top view shows a circular plug with a 1.001 inch diameter, a 1.002 inch diameter, and a 1.078 inch diameter. The top view also shows a 1.078 inch reference dimension, a 1.078 inch reference dimension, and a 1.078 inch reference dimension. The drawing includes a 'FLOW' arrow pointing to the right. The drawing is labeled 'P&W JT8D MIXER NOZZLE TEST' and 'AFT PLUG - REFERENCE'.

FIGURE 4j. AFT PLUG - REFERENCE



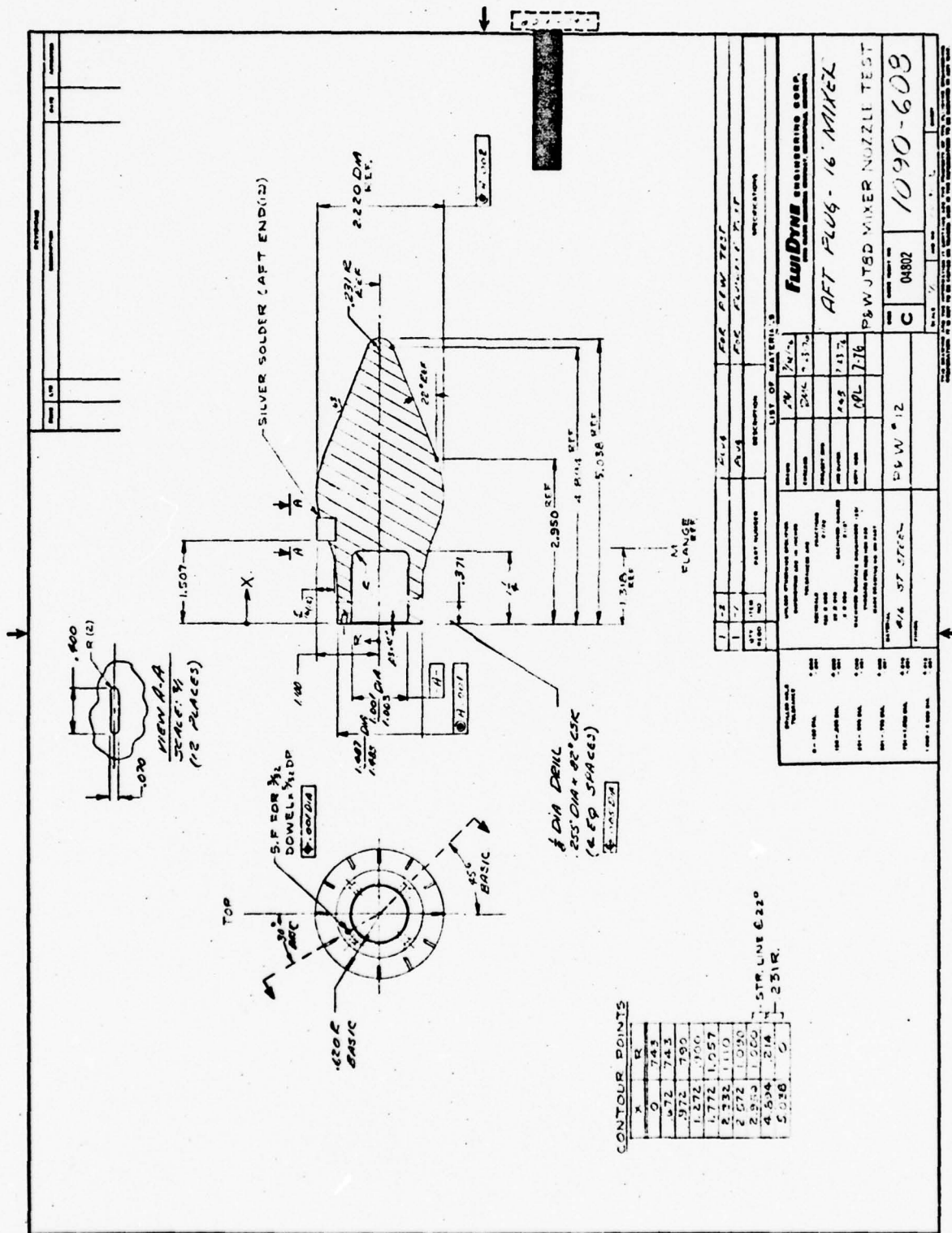
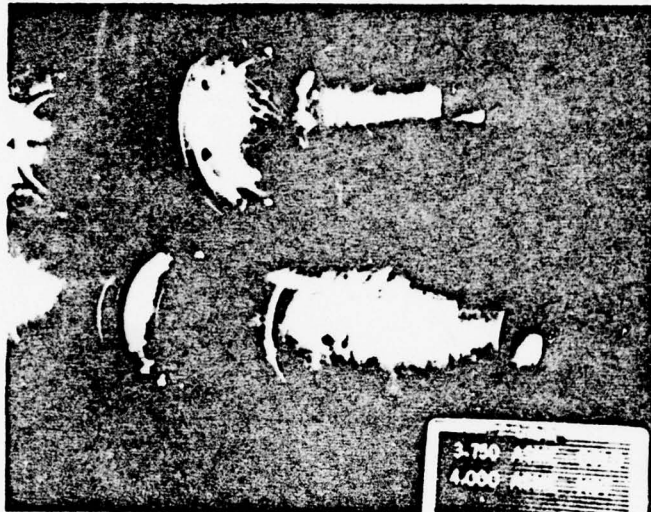


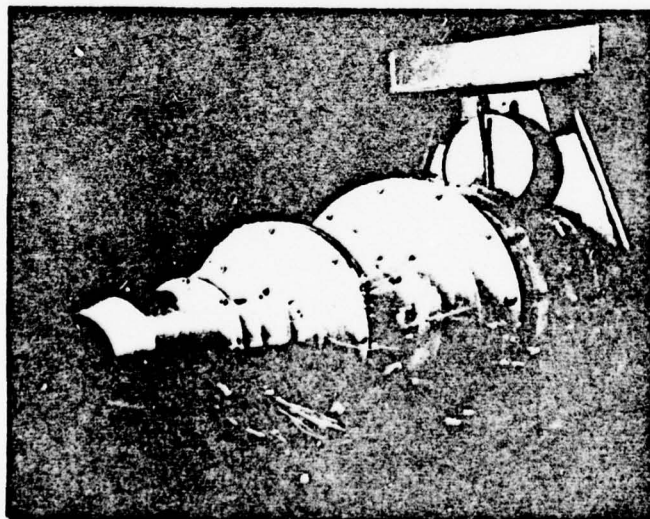
FIGURE 4k. AFT PLUG - 16" MIXER



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ASME nozzle  
Installation



typical mixer  
nozzle installation

FIGURE 5a. MODEL PHOTOGRAPHS



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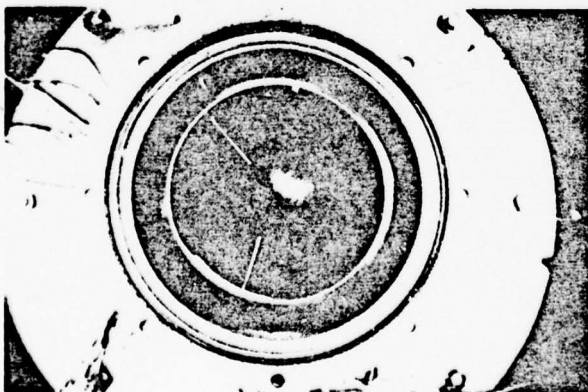
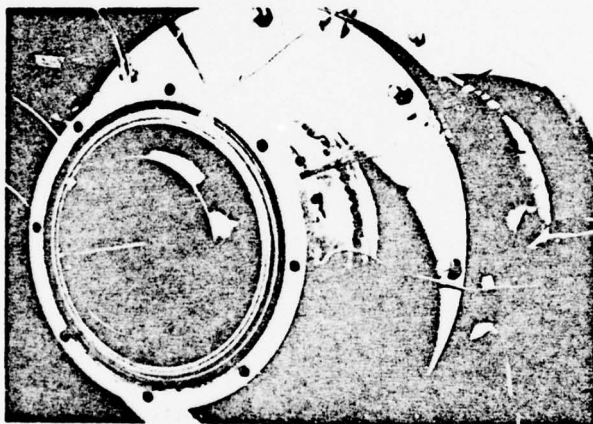
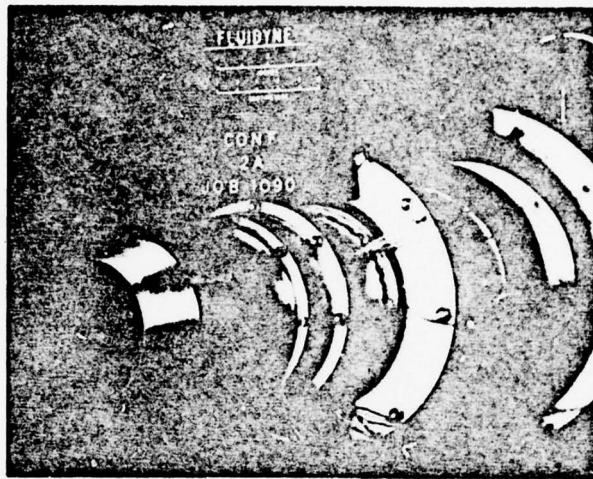


FIGURE 5B. MODEL PHOTOGRAPHS



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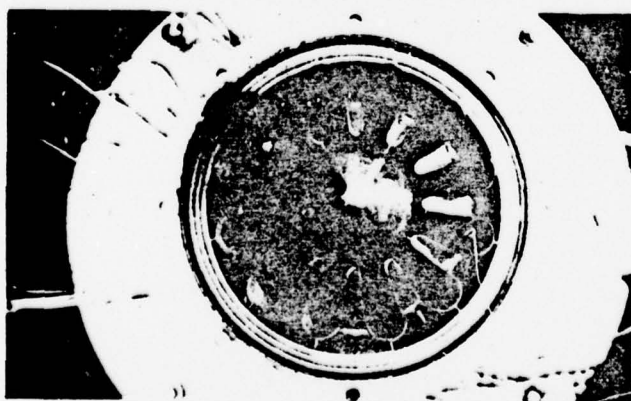
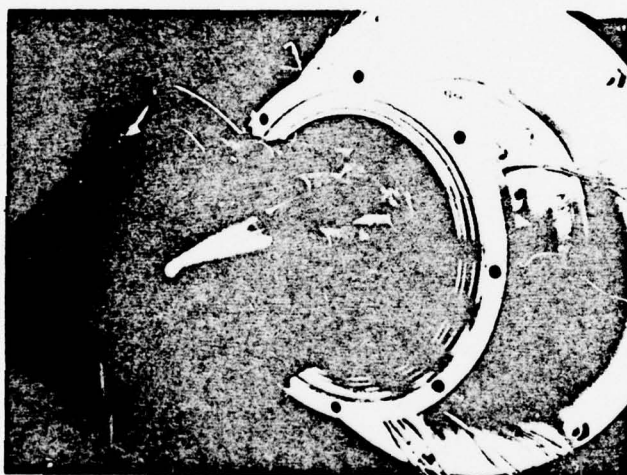


FIGURE 5c. MODEL PHOTOGRAPHS



**FLUIDYNE** ENGINEERING CORPORATION

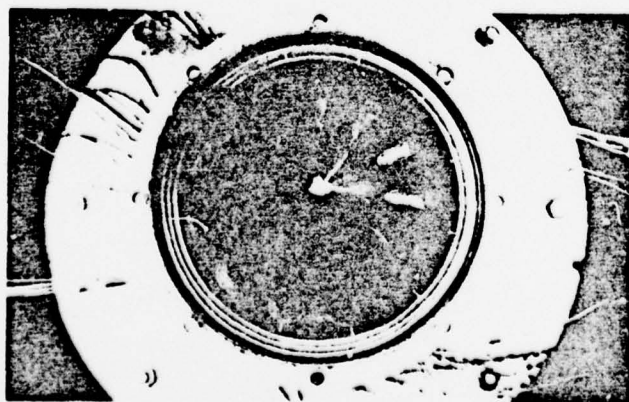
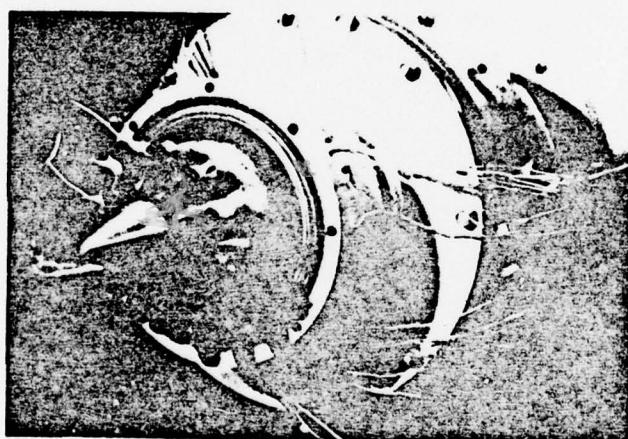
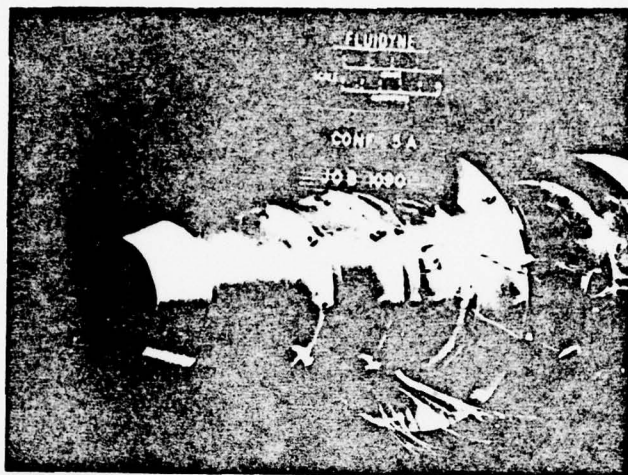
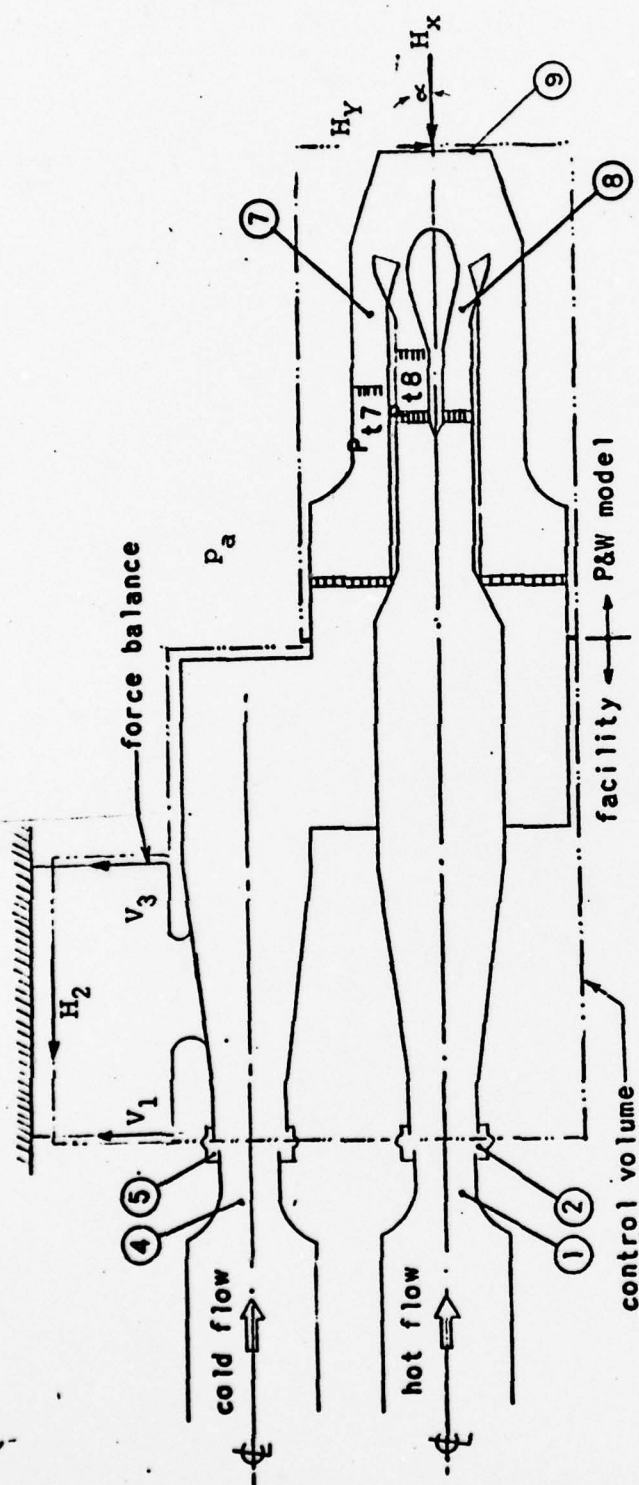


FIGURE 5d. MODEL PHOTOGRAPHS



# FLUIDDYNE ENGINEERING CORPORATION

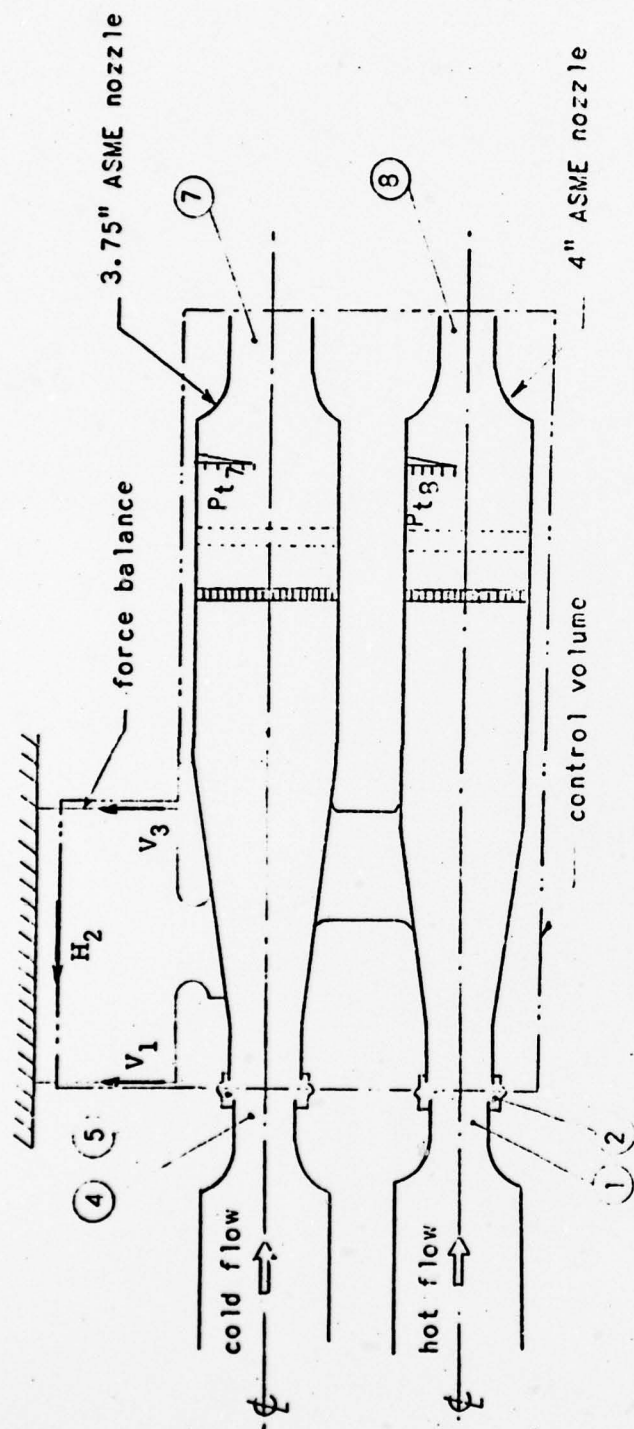


Station	Description
1	ASME Meter Throat (core flow)
2	Flexible Seal (core flow)
4	ASME Meter Throat (fan flow)
5	Flexible Seal (fan flow)
7	Fan Nozzle
8	Core Nozzle
9	Nozzle Exit

FIGURE 6a. STATION NOTATION, MIXER MODEL TESTS



# FLUIDYNE ENGINEERING CORPORATION



Station	Description
1	ASME Meter Throat (core flow)
2	Flexible Seal (core flow)
4	ASME Meter Throat (fan flow)
5	Flexible Seal (fan flow)
7	3.75" Cold ASME Nozzle Throat
8	4" Hot ASME Nozzle Throat

FIGURE 6b. STATION NOTATION, ASME CHECKOUT TESTS



# FLUIDYNE ENGINEERING CORPORATION

Config.	Run No.	$\lambda_7$	$\lambda_8$	$T_{t8}/T_{t7}$	$C_T$	$C_{D7}$	$C_{D8}$			
3.75" ASME cold, upper & 4" ASME hot, lower	1.0	1.506	1.489	1.008	.9919	.9894	.9882			
	2.0	1.761	1.745	1.007	.9932	.9907	.9885			
	3.1	2.010	1.985	1.006	.9944	.9904	.9893			
	4.1	2.258	2.242	1.004	.9950	.9904	.9893			
	5.0	2.510	2.490	1.003	.9954	.9927	.9921			
	6.2	1.504	1.529	2.489	.9916	.9883	.9908			
	7.2	1.762	1.752	2.483	.9914	.9895	.9890			
	8.2	2.011	1.993	2.472	.9959	.9926	.9902			
	9.2	2.264	2.257	2.455	.9965	.9917	.9876			
	10.2	2.518	2.485	2.446	.9966	.9919	.9879			
Config.	Run No.	$\lambda_7$	$\lambda_8$	$\lambda_7/\lambda_8$	$T_{t8}/T_{t7}$	$C_{T_r}$	$C_{D7A7}$	$C_{D8A8}$	$C_{D9}$	$W_7/W_8$
2A	11.4	1.495	1.609	.929	.998	.9884	4.993	8.525	.951	.530
	12.3	2.065	2.214	.932	.997	.9929	5.279	8.636	.979	.569
	13.2	2.533	2.717	.932	.997	.9929	5.286	8.665	.982	.568
	14.2	3.011	3.236	.930	.997	.9886	5.245	8.695	.981	.560
	15.0	1.601	1.609	.995	1.000	.9888	5.889	7.622	.951	.767
	16.0	2.210	2.221	.995	.999	.9933	6.047	7.825	.976	.768
	17.1	2.403	2.714	.885	.998	.9932	4.681	9.304	.984	.445
	18.1	2.833	3.223	.879	.996	.9894	4.563	9.376	.981	.427
	19.1	1.490	1.617	.921	2.117	.9923	4.759	8.646	.943	.721
	20.1	2.057	2.195	.937	2.125	.9973	5.149	8.670	.973	.817
	21.0	2.481	2.698	.919	2.119	.9974	4.952	8.887	.974	.751
	22.0	2.934	3.204	.916	2.117	.9931	4.927	8.950	.977	.739
	23.0	1.600	1.594	1.004	2.128	.9935	5.774	7.575	.940	1.124
	24.0	2.204	2.206	.999	2.130	.9967	5.885	7.891	.970	1.095
	25.1	2.402	2.712	.886	2.113	.9959	4.503	9.363	.976	.624
	26.1	2.832	3.209	.882	2.111	.9936	4.481	9.372	.975	.618
	27.1	1.493	1.620	.921	2.271	.9932	4.804	8.540	.939	.764
	28.2	2.051	2.209	.928	2.265	.9977	5.024	8.752	.969	.809
	29.0	2.529	2.713	.932	2.257	.9966	5.066	8.749	.972	.818
	30.3	2.983	3.215	.928	2.264	.9949	5.016	8.827	.974	.800
	31.0	1.604	1.605	1.000	2.275	.9939	5.748	7.614	.940	1.146
	32.0	2.212	2.208	1.001	2.272	.9976	5.924	7.815	.967	1.154
	33.0	2.404	2.717	.885	2.256	.9986	4.467	9.395	.976	.638
	34.0	2.843	3.224	.882	2.269	.9932	4.453	9.428	.977	.633
	35.1	1.489	1.612	.924	2.387	.9949	4.747	8.652	.943	.766
	36.0	2.062	2.208	.934	2.404	.9998	5.089	8.729	.973	.852
	37.0	2.533	2.719	.932	2.402	.9976	5.053	8.777	.973	.839
	38.0	3.002	3.210	.935	2.392	.9960	5.106	8.741	.975	.853
	214.0	1.668	1.823	.915	2.269	.9966	4.772	8.819	.954	.744

FIGURE 7. RUN SCHEDULE AND MAJOR TEST RESULTS  
(Sheet 1 of 3)



# FLUIDYNE ENGINEERING CORPORATION

Config.	Run No.	$\lambda_7$	$\lambda_8$	$\lambda_7/\lambda_8$	$T_{t8}/T_{t7}$	$C_{T_r}$	$C_{D_{7A_7}}$	$C_{D_{8A_8}}$	$C_{D_9}$	$W_7/W_8$
5A	114.0	1.492	1.606	.929	.992	.9767	5.535	7.713	.932	.646
	115.0	1.676	1.790	.932	.992	.9789	5.633	7.756	.942	.668
	116.0	2.056	2.199	.935	.993	.9851	5.788	7.851	.960	.687
	117.0	2.512	2.685	.936	.993	.9875	5.829	7.894	.966	.688
	118.0	2.982	3.185	.936	.989	.9871	5.824	7.890	.965	.687
	119.0	1.488	1.599	.931	2.286	.9819	5.252	7.625	.906	.950
	120.0	1.675	1.802	.930	2.280	.9881	5.339	7.693	.917	.973
	121.0	2.052	2.197	.934	2.272	.9964	5.508	7.825	.938	.999
	122.0	2.516	2.696	.933	2.275	1.0009	5.529	7.856	.942	.999
	123.0	2.988	3.192	.936	2.282	1.0016	5.551	7.832	.942	1.011
4A	124.0	1.496	1.607	.931	1.000	.9764	5.473	8.059	.947	.616
	125.0	2.043	2.194	.931	1.000	.9871	5.602	8.251	.969	.632
	126.0	2.517	2.687	.937	1.000	.9881	5.641	8.220	.970	.643
	127.0	2.990	3.181	.940	1.000	.9862	5.669	8.218	.971	.649
	128.0	1.603	1.605	.999	.998	.9753	5.869	7.650	.946	.765
	129.0	2.200	2.190	1.004	.997	.9864	6.010	7.793	.966	.773
	130.0	2.383	2.691	.885	.993	.9868	5.355	8.513	.970	.555
	131.0	2.818	3.181	.886	.993	.9846	5.385	8.537	.974	.557
	132.0	1.499	1.608	.933	2.159	.9798	5.290	7.900	.923	.900
	133.0	2.056	2.198	.935	2.160	.9926	5.481	8.087	.949	.938
	134.0	2.512	2.698	.931	2.134	.9972	5.458	8.137	.951	.919
	135.0	2.992	3.190	.938	2.135	.9968	5.482	8.053	.947	.940
	136.0	1.617	1.614	1.002	2.139	.9784	5.724	7.567	.930	1.115
	137.0	2.201	2.189	1.006	2.146	.9919	5.902	7.696	.951	1.138
	138.0	1.499	1.602	.935	2.325	.9792	5.300	7.838	.919	.948
	139.0	2.051	2.186	.938	2.285	.9940	5.494	8.039	.947	.977
	140.0	2.513	2.689	.935	2.288	1.0008	5.475	8.071	.948	.968
	141.0	2.988	3.177	.941	2.294	1.0004	5.492	8.097	.951	.975
	142.0	1.605	1.605	1.000	2.296	.9808	5.690	7.486	.922	1.159
	143.0	2.190	2.190	1.000	2.285	.9945	5.845	7.681	.946	1.160
	144.0	2.375	2.697	.881	2.281	1.0005	5.107	8.367	.943	.819
	145.0	2.820	3.178	.837	2.277	1.0010	5.154	8.398	.948	.829

FIGURE 7. RUN SCHEDULE AND MAJOR TEST RESULTS (Cont.)  
(Sheet 2 of 3)



**FLUIDYNE ENGINEERING CORPORATION**

Config.	Run No.	$\lambda_7$	$\lambda_8$	$\lambda_7/\lambda_8$	$T_{t8}/T_{t7}$	$C_{Tr}$	$C_{D7}A_7$	$C_{D8}A_8$	$C_{D9}$	$w_7/w_8$
4A(cont.)	146.0	1.485	1.597	.930	2.411	.9828	5.248	7.867	.917	.943
	147.1	2.052	2.211	.928	2.386	.9988	5.315	8.067	.940	.954
	148.1	2.509	2.714	.924	2.382	1.0024	5.332	8.117	.945	.946
	149.0	2.983	3.187	.936	2.400	1.0056	5.467	8.078	.948	.991
	150.0	2.380	2.692	.884	2.420	1.0003	5.105	8.351	.941	.849
	151.0	2.816	3.185	.884	2.422	1.0015	5.105	8.390	.944	.845
	152.0	2.695	2.698	.999	2.283	1.0006	5.847	7.705	.948	1.155
	153.0	2.695	2.686	1.003	2.145	.9963	5.880	7.700	.950	1.130

FIGURE 7. RUN SCHEDULE AND MAJOR TEST RESULTS (Cont.)  
(Sheet 3 of 3)



# **FLUIDYNE ENGINEERING CORPORATION**

Symbol	$T_{t8}/T_{t7}$	Run Nos.
○	1.000	1-5
◇	2.470	6-10

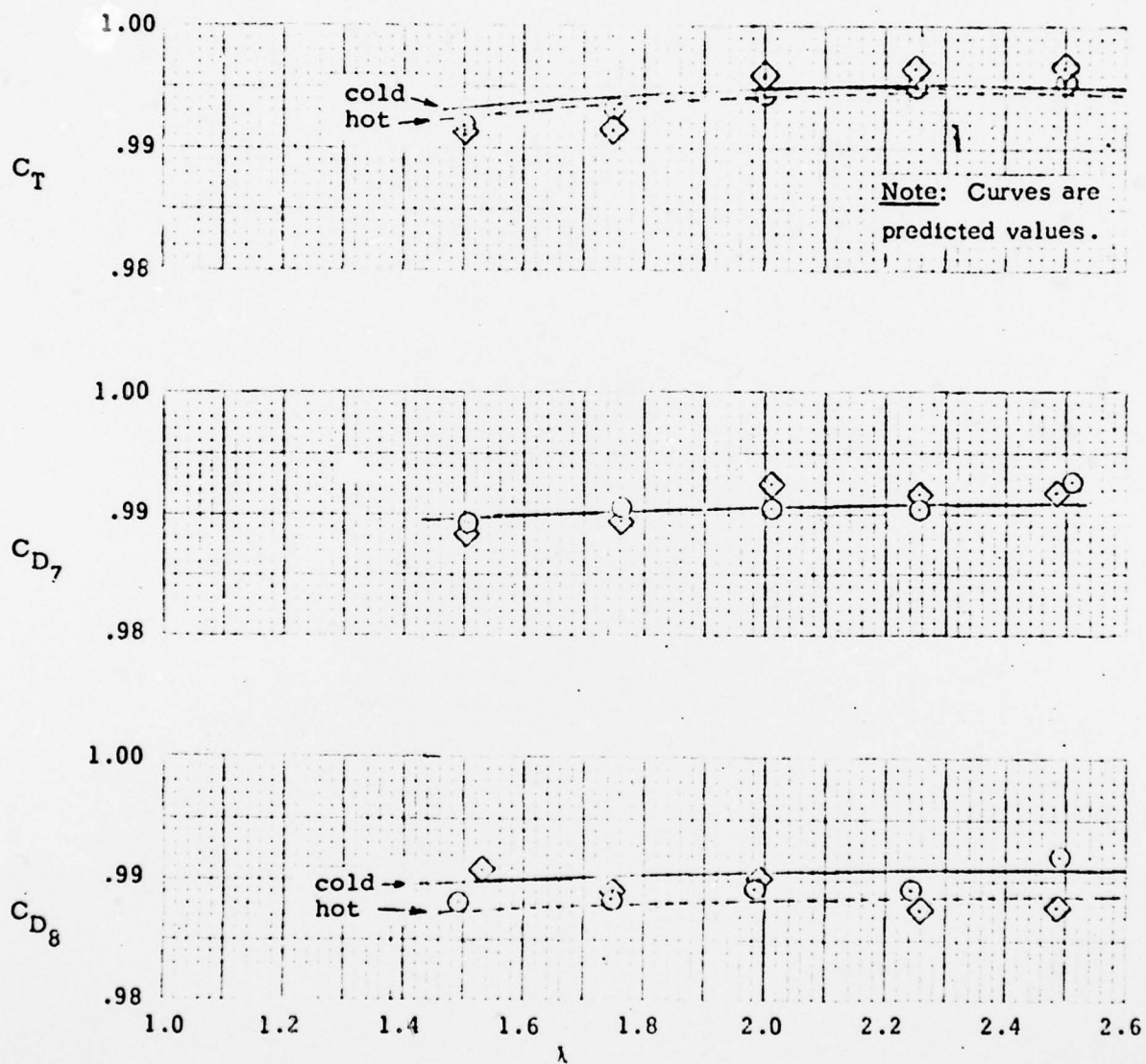


FIGURE 8. THRUST AND DISCHARGE COEFFICIENTS, ASME CHECKOUT TESTS



# FLUIDYNE ENGINEERING CORPORATION

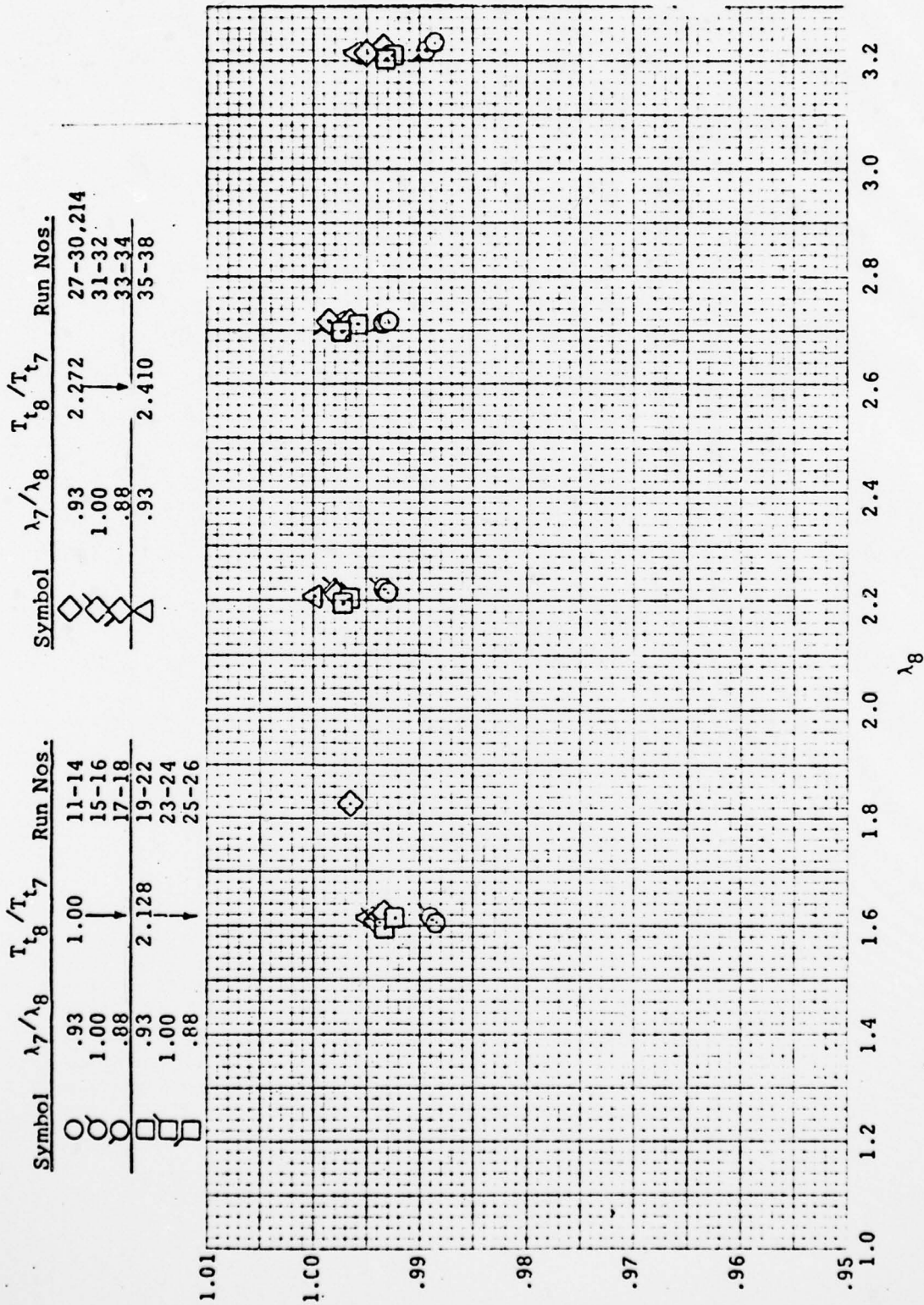


FIGURE 9a. THRUST COEFFICIENTS, CONFIGURATION 2A



# FLUIDDYNE ENGINEERING CORPORATION

Symbol	$\lambda_7/\lambda_8$	$T_{t8}/T_{t7}$	Run Nos.
○	.03	1.000	114-118
◇	↓	2.272	119-123

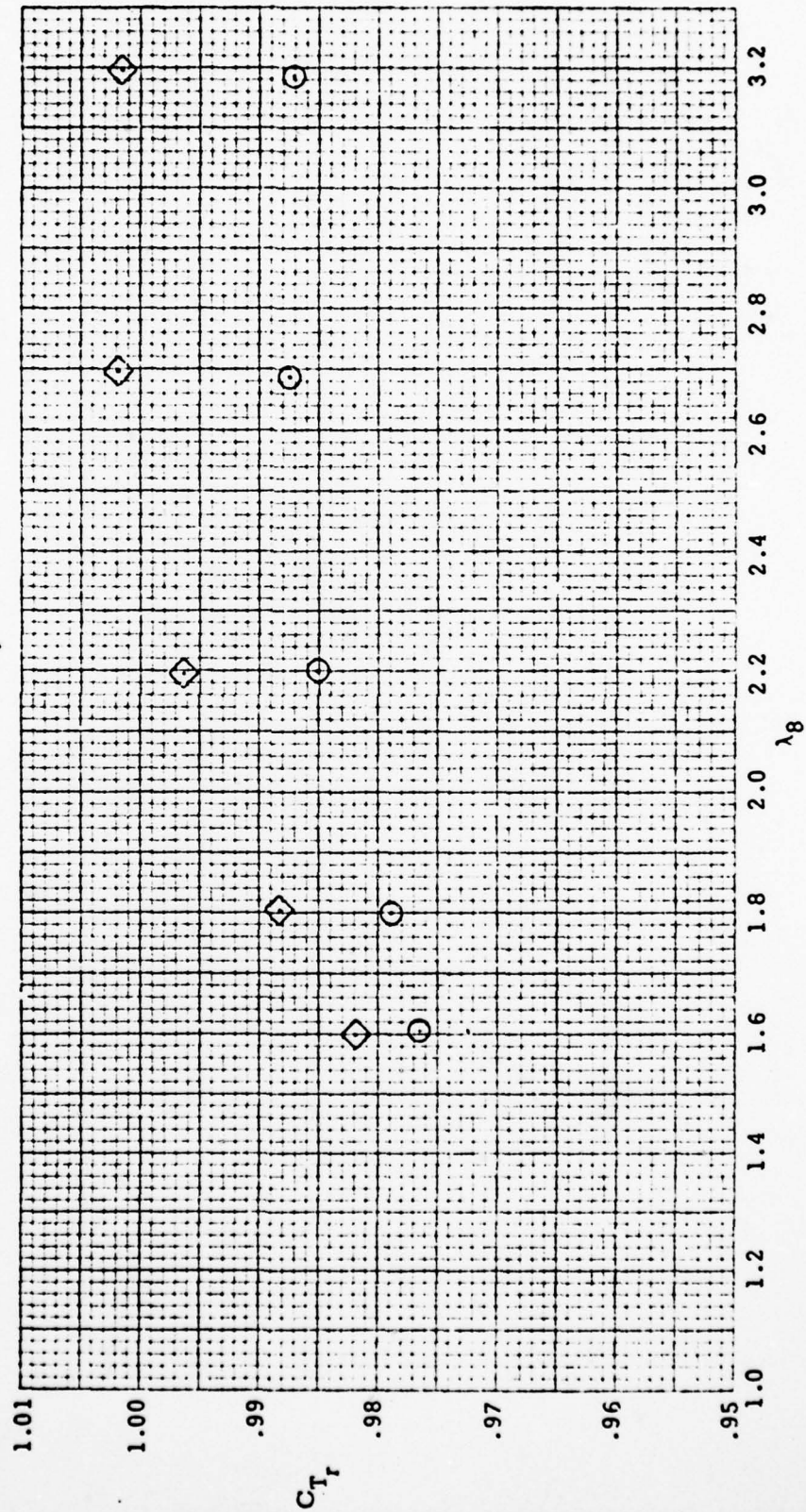


FIGURE 9b. THRUST COEFFICIENTS, CONFIGURATION 5A



# FLUIDDYNE ENGINEERING CORPORATION

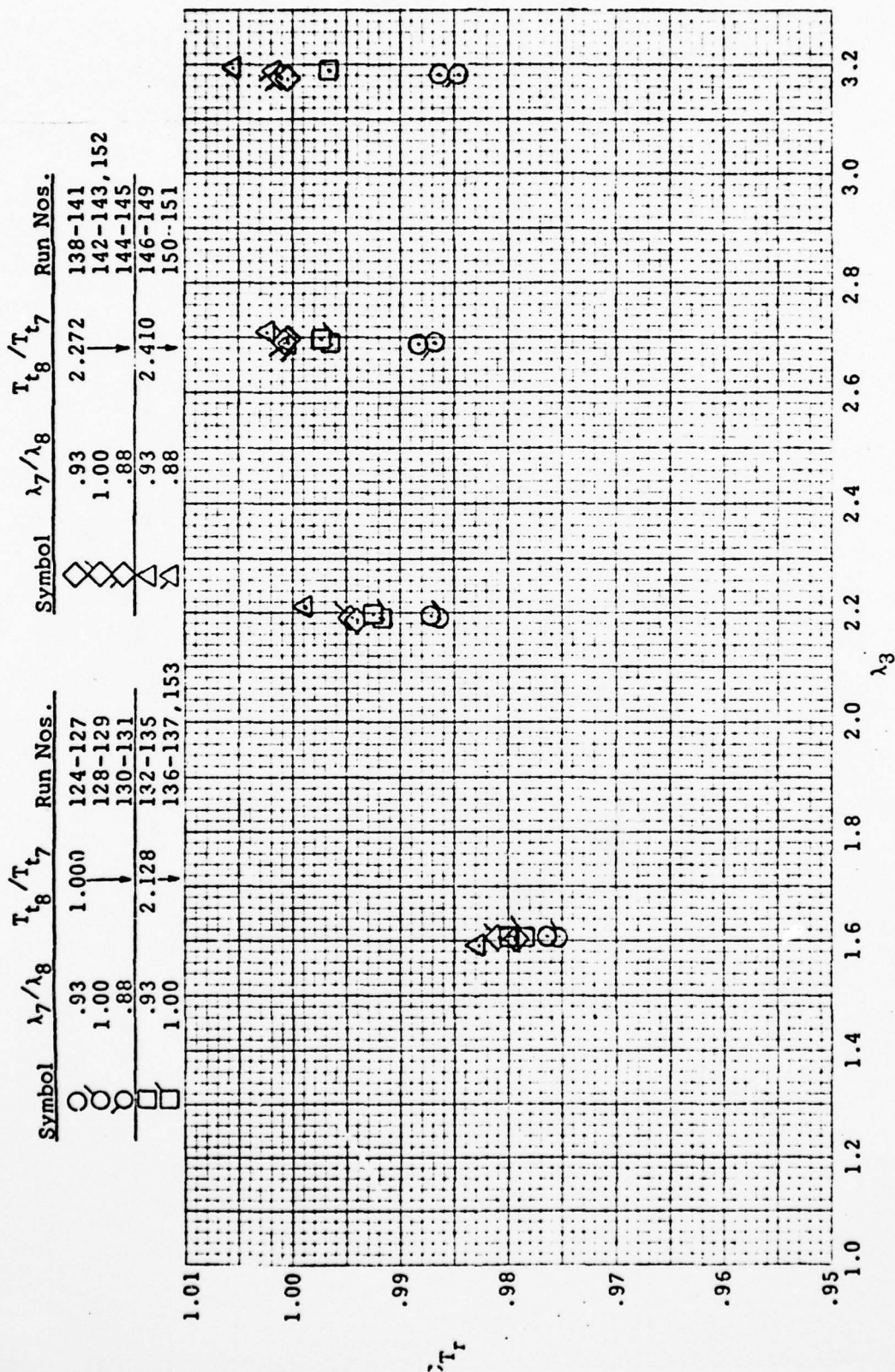


FIGURE 9c. THRUST COEFFICIENTS, CONFIGURATION 4A



# FLUIDYNE ENGINEERING CORPORATION

Symbol	$\lambda_7/\lambda_8$	$T_{t8}/T_{t7}$	Run Nos.
○	.93	1.00	11-14
○	1.00	↓	15-16
○	.88	↓	17-18
□	.93	2.128	19-22
□	1.00	↓	23-24
□	.88	↓	25-26

Symbol	$\lambda_7/\lambda_8$	$T_{t8}/T_{t7}$	Run Nos.
◇	.93	2.272	27-30, 214
◇	1.00	↓	31-32
◇	.88	↓	33-34
△	.93	2.410	35-38

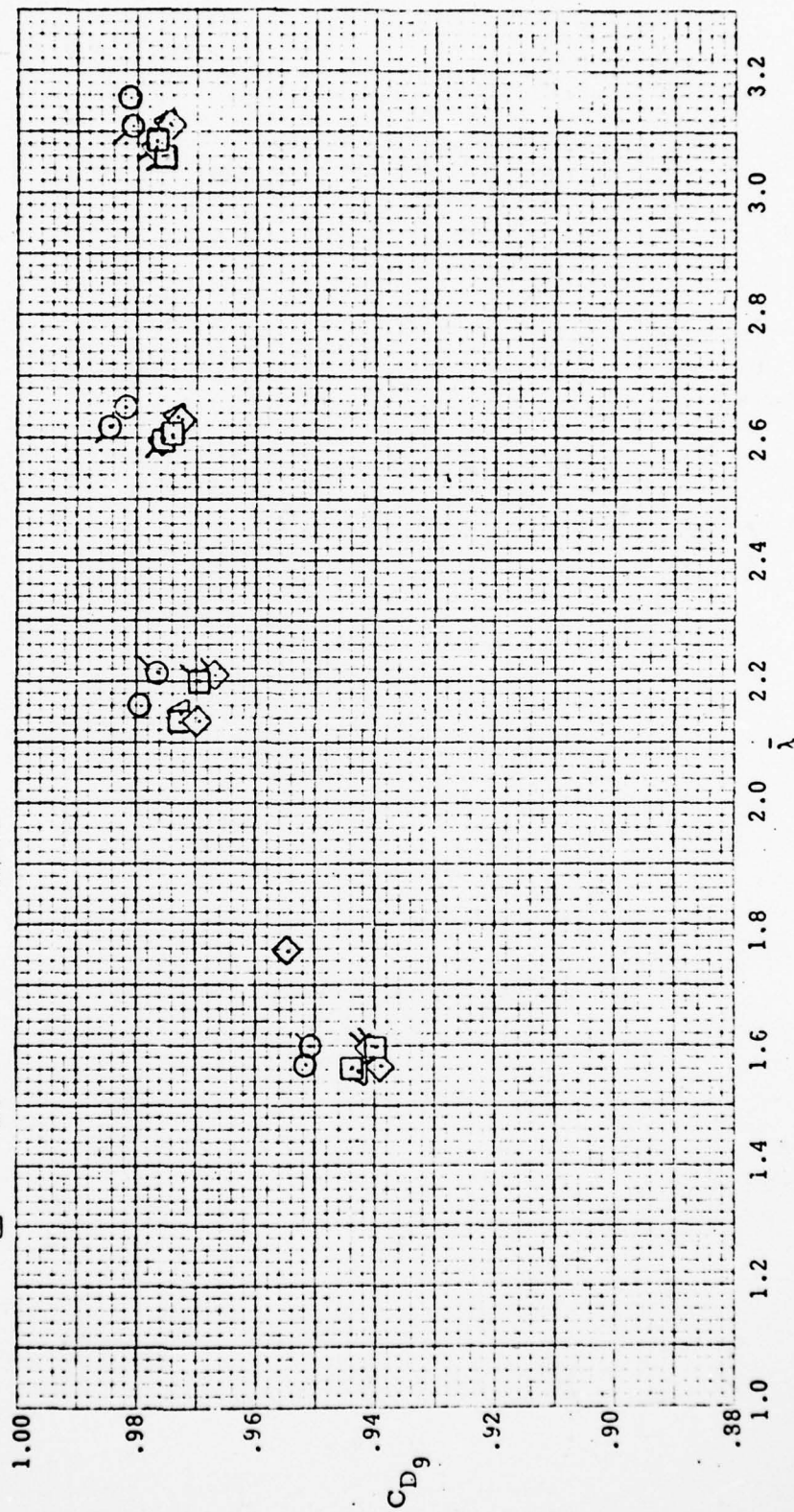


FIGURE 10a. NOZZLE DISCHARGE COEFFICIENTS, CONFIGURATION 2A



# FLUIDDYNE ENGINEERING CORPORATION

Symbol	$\lambda_7/\lambda_8$	$T_{t8}/T_{t7}$	Run Nos.
○	.93	1.000	114-118
◇	↓	2.272	119-123

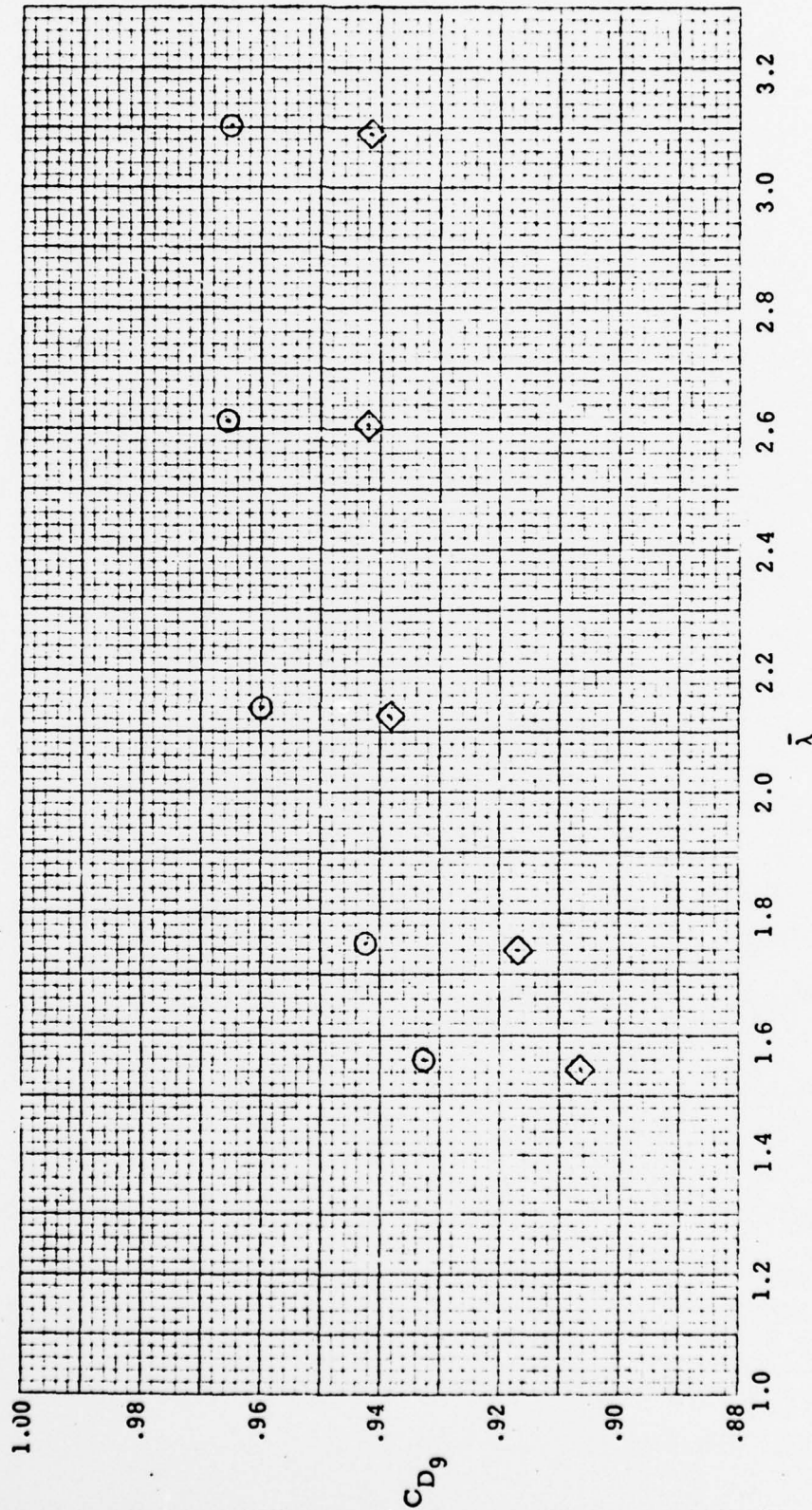


FIGURE 10b. NOZZLE DISCHARGE COEFFICIENTS, CONFIGURATION 5A



# FLUIDDYNE ENGINEERING CORPORATION

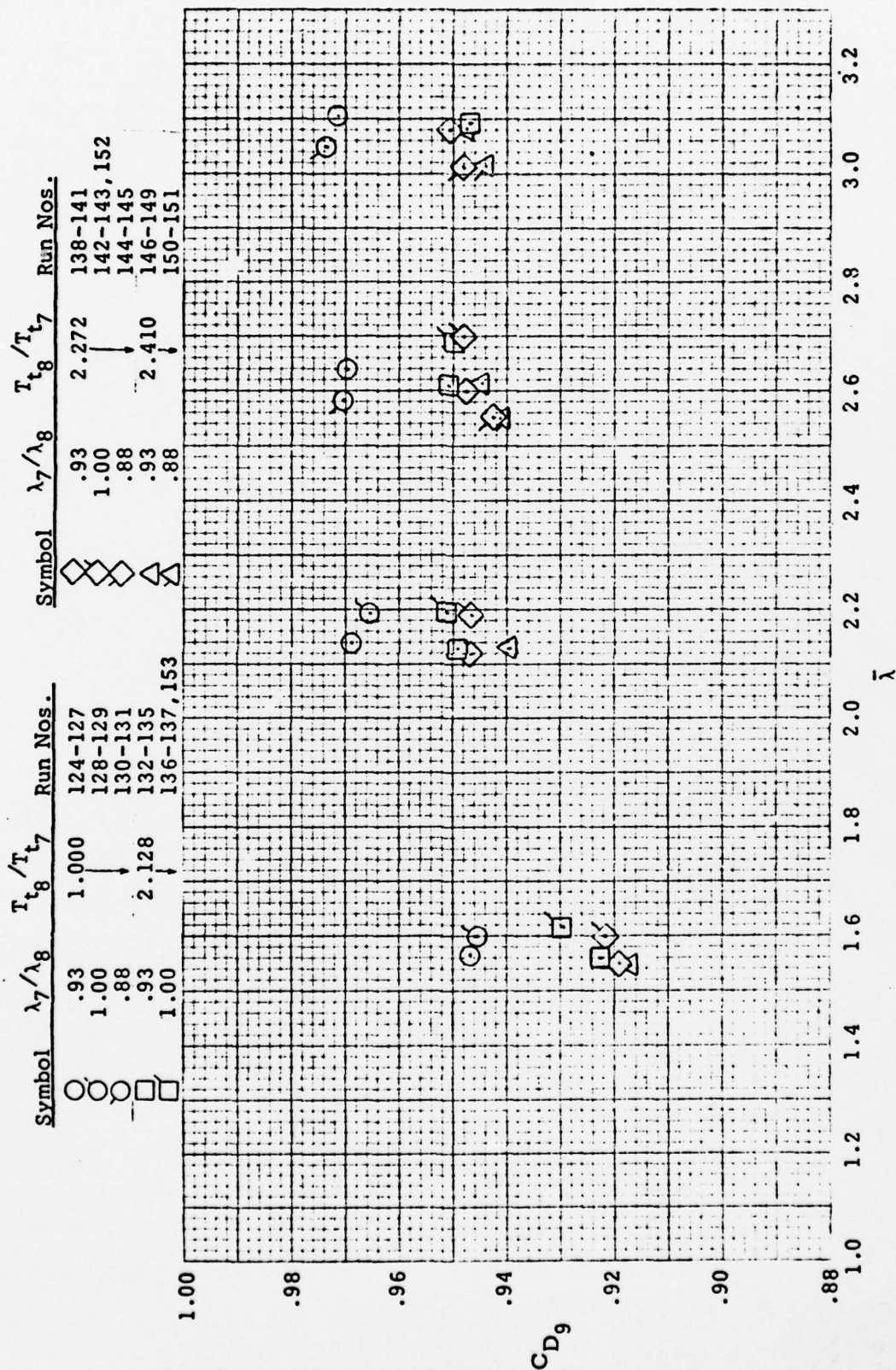


FIGURE 10c. NOZZLE DISCHARGE COEFFICIENTS, CONFIGURATION 4A